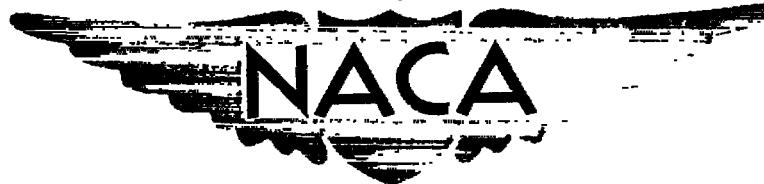


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RESEARCH MEMORANDUM

TABULATED PRESSURE DATA FOR A 60° DELTA-WING-BODY-TAIL
MODEL WITH A HOT JET EXHAUSTING FROM A
PYLON-MOUNTED NACELLE

By Edwin E. Lee, Jr., and John M. Swihart

Langley Aeronautical Laboratory
Langley Field, Va.

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TABULATED PRESSURE DATA FOR A 60° DELTA-WING--BODY--TAIL
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SUMMARY

An investigation has been conducted in the Langley 16-foot transonic tunnel at Mach numbers from 0.80 to 1.05 and an average Reynolds number of 10×10^6 to determine the interference effects of a hot jet exhaust on the pressure distribution of a 60° delta wing. The jet issued from the base of a pylon-mounted nacelle and was operated at total-pressure ratios from 1 to 7. The nacelle was located at 2 spanwise, 2 chordwise, and 3 vertical positions under the wing. Pressure data were taken at angles of attack of 0°, 5°, and 10° and an inboard flap was deflected to ±7.5° for part of the investigation. This report presents tabulated pressure coefficients for the wing and nacelle for the complete range of test conditions.

INTRODUCTION

There have been many investigations conducted by the National Advisory Committee for Aeronautics to determine the loads on wing-body models at transonic speeds (for example, ref. 1). It was known that the addition of engine nacelles and/or stores to the wing would seriously alter the wing loading obtained on these models. An example of this effect is shown in reference 2. In addition, it was shown in reference 3 that a jet issuing from a nacelle had large effects on the pressures on an adjacent flat plate. Since very little jet-interference data had been obtained on actual airplane configurations, some jet-interference effects on steady wing loads were shown in references 4 and 5.

Because of the immediate interest in the complete range of data, selected examples of which are shown in reference 5, the purpose of this paper is to present tabulations of the pressure coefficients measured

on all configurations of the 3-percent-thick 60° delta-wing model of reference 5. No analysis of these data is included in this paper.

The investigation covered a Mach number range from 0.80 to 1.05 at angles of attack of 0°, 5°, and 10°. The average Reynolds number based on wing mean aerodynamic chord was 10×10^6 . The hot jet exhaust was furnished by a hydrogen-peroxide turbojet engine simulator (see ref. 6) and the jet total-pressure ratio was varied from 1 to 5 and sometimes to 7 at each Mach number. Pressures were obtained on a deflected inboard flap for two of the configurations.

SYMBOLS

A	aspect ratio
b	wing or flap span
c	local chord
c'	mean aerodynamic chord
C _p	pressure coefficient, $\frac{p_{\text{local}} - p_{\infty}}{q_{\infty}}$
d _j	jet exit diameter
L	lower surface
M	Mach number
p _∞	free-stream static pressure
p _{t,j} /p _∞	jet total-pressure ratio
q _∞	free-stream dynamic pressure
r	body or nacelle radius
S	wing area
U	upper surface
x	longitudinal distance from body or nacelle nose and wing or flap leading edge (positive rearward)

- y spanwise distance from body center line
z vertical distance below wing chord plane to nacelle center line
 α angle of attack of body center line
 δ deflection of flap from wing chord plane (positive with trailing edge down)
 λ taper ratio
 ϕ meridian angle measured clockwise from top center line (looking forward)

Subscripts:

- e exit
f flap
n nacelle center line
 ∞ free stream

APPARATUS AND METHODS

Tunnel

This investigation was conducted in the Langley 16-foot transonic tunnel, which is a single-return atmospheric wind tunnel with a slotted test section. The model sting support system is arranged so that the model is located near the center line of the test section at all angles of attack.

Model

Figure 1 shows several photographs of the model configurations mounted in the tunnel test section. Figure 1(f) is a closeup of the nacelle base and shows the exit of the turbojet simulator. The development of this system and the simulation technique is described in reference 6. Figure 2 is a sketch showing details of the wing-body-tail model. The physical characteristics of the wing, body, horizontal tails, and vertical tail are shown in figure 2. The steel wing was mounted in a midwing position on the body and had no geometric incidence, twist, or dihedral. The horizontal tail was constructed of steel and was

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mounted with 0° incidence on the wing chord plane extended. The vertical tail was constructed of plastic reinforced with a steel core and had no incidence relative to the model center line. Body ordinates, location of wing orifices, airfoil sections, and sweepback angles of the aerodynamic surfaces are also shown in figure 2. Note that the orifice locations marked with asterisks at $y/b = 0.19$ were used only for the deflected flap and were located at $y/b = 0.22$. For convenience in making the machine tabulations, the pressure coefficients measured with the flap deflected are located at $y/b = 0.19$.

Figure 3 shows the details of the various pylon-nacelle configurations investigated and includes the spanwise, chordwise, and vertical positions of the jet exit. The nacelle orifice locations presented in figure 3 have been identified numerically for association with the corresponding tabulated nacelle pressure coefficients. The nacelle, which was common to all configurations, consisted of an ogival nose, a cylindrical center section, and a 10° boattailed afterbody. The hydrogen-peroxide jet simulator was mounted inside the nacelle and was supplied with liquid peroxide from a tube carried internally through the wing and pylon.

Instrumentation

Surface pressures were measured at 6 spanwise stations on the wing, 4 meridian rows on the nacelle, and at 3 meridian angles on the nacelle base as shown in figure 3(b). Surface pressures were measured on the flap and figure 4 gives the orifice locations in percent of wing chord and corresponding flap chord to facilitate the use of the tabulations. This wing is the same as that used in reference 1, and the pressure instrumentation is similar.

The jet total pressure was measured at a point 1 inch ahead of the sonic exit. The model angle of attack was measured by a calibrated-pendulum strain-gage angle-of-attack indicator located in the model nose. The angles of attack thus determined were independent of sting and balance deflections under load.

Tests

Pressures were obtained at Mach numbers of 0.80, 0.90, 1.00, and 1.05 at angles of attack of 0° , 5° , and 10° with the jet simulator

operating at pressure ratios of 1 (off), 3, 5, and sometimes 7. The flap was deflected 7.5° with the nacelle located at $y_n/b = 0.41$ and $\frac{z}{d_j} = 2$ and $\pm 7.5^\circ$ with the nacelle located at $y_n/b = 0.41$ and $\frac{z}{d_j} = 1.23$.

Reduction of Data

The wing and nacelle surface pressures were photographically recorded on three 100-tube mercury manometer boards. The jet total pressure was measured by an electrical pressure transducer which gave a visual indication to the simulator operator and the signal was also used to operate an analog-to-digital converter whose output was recorded on punch cards. The photographic film was read on an automatic film reader and recorded on punch cards. Automatic computing machines reduced the data from the punch cards to individual pressure coefficients and to jet total-pressure ratio. Automatic tabulating machines compiled the individual pressure coefficients for each configuration and test condition into the tables presented subsequently.

Accuracy

The following is the estimated accuracy of the quantities presented in this paper:

M	± 0.005
C_p	± 0.005
$p_{t,j}/p_\infty$	± 0.10
α , deg	± 0.10

It was shown in reference 1 that the aeroelastic twist of this wing was very small and no corrections to wing angle of attack were believed to be necessary. Reference 7 shows the deflection constants for the control surfaces on this wing and no corrections to the control deflection angles presented herein have been made. A complete final check of the machine tabulations presented herein has not been made and random errors due to orifice malfunction may be present. Such errors can usually be detected by visual inspection of the plotted data.

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RESULTS

The results of the investigation are presented in tables I to V. The tables show pressure coefficients for each orifice on the wing upper and lower surfaces and on the nacelle. The data in the tables are presented for increasing values of δ , M , α , and $p_{t,j}/p_\infty$ for each configuration shown in figure 3. The following index is provided for convenient reference to the tabulated data:

Table

Configuration 1: $y_n/b/2 = 0.41$; $x_e/c = 0.38$; $z/d_j = 4.0$	I
Configuration 2: $y_n/b/2 = 0.41$; $x_e/c = 0.38$; $z/d_j = 2.0$	II
Configuration 3: $y_n/b/2 = 0.41$; $x_e/c = 0.69$; $z/d_j = 2.0$	III
Configuration 4: $y_n/b/2 = 0.41$; $x_e/c = 0.69$; $z/d_j = 1.23$	IV
Configuration 5: $y_n/b/2 = 0.69$; $x_e/c = 0.62$; $z/d_j = 4.0$	V

Langley Aeronautical Laboratory,
National Advisory Committee for Aeronautics,
Langley Field, Va., September 27, 1957.

- [REDACTED]
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 4. Judd, Joseph H., and Falanga, Ralph A.: Free-Flight Investigation at Transonic Speeds of the Power-On Characteristics Including Some Effects of Sonic Propulsive Jets of a Four-Engine Delta-Wing Configuration. NACA RM L57E31, 1957.
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 6. Runckel, Jack F., and Swihart, John M.: A Hydrogen Peroxide Turbojet-Engine Simulator for Wind-Tunnel Powered-Model Investigations. NACA RM L57H15, 1957.
 7. Critzos, Chris C., and Foss, Willard E., Jr.: Effect of Full-Span Trailing-Edge Elevons on the Transonic Longitudinal Aerodynamic Characteristics of a Wing-Body Combination Having a 3-Percent-Thick Triangular Wing With 60° Leading-Edge Sweep. NACA RM L57G03, 1957.

TABLE I

X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp
.01	.079	-.074					.096	-.005					1	-.104
.03	-.081	-.081	-.069	-.042			.067	-.031	-.142	-.122	-.105		2	-.161
.05	.016	-.076					.044	-.047		-.102	-.087	-.090	3	-.186
.08	-.011	-.082	-.052				.034	-.064	-.108				4	-.093
.10	-.017	-.091		-.057	-.045		.006			-.089			5	-.114
.13	-.016	-.104	-.068		-.057		.021	-.206	-.099		-.084		6	-.186
.20	-.061						.049						7	-.174
.25	-.059	-.084	-.072	-.074	-.045	-.076	-.089	-.126	-.070	-.055	-.058	-.032	8	-.085
.30	-.078	-.083	-.070	-.068	-.065	-.082	-.120	-.134	-.055	-.061	-.080	-.064	9	-.164
.35	-.067	-.106	-.075	-.085	-.085	-.095	-.112	-.134	-.063	-.065	-.089	-.089	10	-.177
.40	-.055	-.126	-.081	-.100	-.091	-.113	-.101	-.127	-.058	-.097	-.120	-.111	11	-.069
.45	-.092	-.103	-.088	-.092	-.073	-.111	-.116	-.126	-.064	-.084	-.113	-.096	12	-.078
.50	-.082	-.107	-.087	-.094	-.088	-.106	-.097	-.092	-.074	-.061	-.101	-.099	13	-.074
.55	-.082	-.105	-.095	-.080	-.070	-.080	-.080	-.070	-.072	-.045	-.082			
.60	-.076	-.101	-.061	-.079	-.070	-.093	-.052	-.062	-.064	-.053	-.066	-.073		
.65	-.144	-.096	-.070	-.047	-.074		-.047	-.067	-.063	-.062	-.054			
.68											-.075			
.70	-.077	-.086	-.060	-.068	-.065	-.063	-.051	-.061	-.051	-.068		-.054		
.75	-.072	-.068	-.047				-.049	-.064	-.050			-.055		
.77											-.043			
.79											-.027			
.80	-.064	-.059					-.046	-.055			-.028	-.026		
.82											-.038			
.83											-.031	-.025		
.85	-.046	-.064					-.039	-.054				-.016		
.87											-.025			
.88											-.023	-.013		
.90											-.051			
.91											-.013	-.003	-.001	-.002
.93											-.052			
.95	-.012	-.019					-.048	-.039			-.051			
.97											-.026			
.98											-.019			
											-.023			

$\delta = 0.0$						$M = 0.8C$						$\alpha = 0.0$						Nacelle	
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p					
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90							
.01	.076	-.076	-.070	-.050	-.032	-.055	.082	-.010	.053	-.036	-.149	-.118	-.102	1	-.091				
.03	-.084	-.084	-.070	-.050	-.032	-.055	-.053	-.036	-.053	-.053	-.119	-.096	-.087	2	-.151				
.05	.011	-.073	-.050	-.032	-.055	-.026	-.067	-.119	-.097	-.095	3	-.164							
.08	-.010	-.082	-.058	-.072	-.061	-.018	-.035	-.205	-.201	-.089	4	-.093							
.10	-.019	-.096	-.072	-.061	-.071	-.061	-.061	-.126	-.060	-.078	5	-.111							
.15	-.014	-.101	-.076	-.061	-.055	-.120	-.091	-.126	-.049	-.077	6	-.173							
.20	-.066	-.101	-.076	-.061	-.055	-.132	-.091	-.126	-.077	-.087	7	-.149							
.25	-.060	-.087	-.084	-.088	-.058	-.120	-.091	-.126	-.049	-.068	8	-.077							
.30	-.079	-.085	-.084	-.081	-.057	-.120	-.091	-.127	-.077	-.045	9	-.146							
.35	-.072	-.106	-.092	-.096	-.101	-.120	-.104	-.127	-.076	-.095	10	-.161							
.40	-.057	-.127	-.093	-.112	-.111	-.103	-.116	-.122	-.074	-.116	11	-.083							
.45	-.096	-.104	-.098	-.102	-.094	-.122	-.115	-.122	-.087	-.120	12	-.075							
.50	-.084	-.108	-.096	-.109	-.104	-.113	-.095	-.080	-.101	-.083	13	-.067							
.55	-.082	-.110	-.108	-.089	-.089	-.077	-.079	-.092	-.075	-.091	14	-.077							
.60	-.083	-.103	-.077	-.097	-.089	-.067	-.064	-.081	-.085	-.075	15	-.079							
.65	-.153	-.100	-.382	-.361	-.094	-.105	-.067	-.052	-.079	-.064	16	-.084							
.68	-.070	-.089	-.074	-.085	-.078	-.069	-.072	-.077	-.070	-.085	17	-.059							
.70	-.092	-.089	-.074	-.085	-.078	-.069	-.066	-.080	-.068	-.066	18	-.054							
.75	-.079	-.069	-.061	-.061	-.061	-.066	-.066	-.068	-.059	-.056	19	-.056							
.77	-.057	-.069	-.051	-.071	-.051	-.062	-.063	-.063	-.029	-.022	20	-.022							
.79	-.070	-.063	-.059	-.059	-.042	-.033	-.062	-.063	-.050	-.041	21	-.022							
.82	-.057	-.067	-.044	-.036	-.025	-.055	-.064	-.063	-.038	-.029	22	-.022							
.85	-.057	-.067	-.044	-.036	-.025	-.055	-.064	-.063	-.029	-.022	23	-.022							
.87	-.054	-.027	-.013	-.013	-.010	-.058	-.063	-.058	-.029	-.022	24	-.022							
.89	-.049	-.013	-.001	-.001	-.010	-.063	-.060	-.015	-.006	-.003	25	-.003							
.90	-.027	-.023	-.016	-.022	-.010	-.059	-.044	-.004	-.007	-.007	26	-.007							
.91	-.031	-.001	-.001	-.001	-.010	-.063	-.063	-.013	-.006	-.003	27	-.003							
.93	-.029	-.001	-.001	-.001	-.010	-.060	-.060	-.015	-.004	-.002	28	-.002							
.95	-.027	-.023	-.016	-.022	-.010	-.059	-.044	-.004	-.007	-.007	29	-.007							
.97	-.018	-.016	-.016	-.016	-.010	-.035	-.035	-.004	-.007	-.007	30	-.007							

TABLE I.- Continued

TABLE I.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1

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TABLE I.- Continued

TABLE I. - Continued

TABLE I - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1

$$\delta = 0.0 \quad M = 0.90 \quad \alpha = -0.1 \quad p_{t,i}/p_{cm} = 5.1$$

TABLE I.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1

TABLE I. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1

		$\delta = 0.0$						$M = 0.90$						$\alpha = 5^\circ 0$		$P_{t,j}/P_{\infty} = 3.2$			
X	C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle					
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p				
.01	-0.03	-0.740	-0.801	-0.814	-0.751	-0.668		.272	.314					1	.015				
.02	-0.10	-0.807	-0.801	-0.853	-0.751	-0.668		.257	.283	.197	.240	.241	.208	2	-0.038				
.03	-0.181	-0.807	-0.801	-0.853	-0.751	-0.668		.257	.250	.193	.207	.208		3	-0.044				
.04	-0.168	-0.445	-0.497	-0.500	-0.500	-0.500		.196	.243	.134				4	-0.010				
.05	-0.175	-0.190	-0.213	-0.213	-0.213	-0.213		.190			.149			5	-0.017				
.06	-0.218	-0.370	-0.213	-0.213	-0.213	-0.213		.173	.125	.089	.120			6	-0.022				
.07	-0.244	-0.201	-0.224	-0.203	-0.726	-0.647		.158						7	-0.075				
.08	-0.255	-0.193	-0.226	-0.213	-0.485	-0.480		.113	-0.014	.099	.083	.097	.141	8	-0.008				
.09	-0.177	-0.213	-0.242	-0.248	-0.593	-0.655		.058	-0.021	.097	.070	.047	.113	9	-0.113				
.10	-0.176	-0.229	-0.257	-0.280	-0.476	-0.658		.011	-0.006	.064	.052	.044	.070	10	-0.056				
.11	-0.195	-0.210	-0.265	-0.265	-0.351	-0.647		.003	-0.003	.050	.015	.007	.037	11	-0.001				
.12	-0.182	-0.226	-0.280	-0.289	-0.282	-0.634		.019	-0.005	.024	.011	.002	.033	12	-0.001				
.13	-0.189	-0.224	-0.263	-0.263	-0.240	-0.214		.004	-0.034	.001	.026	.002	.016	13	-0.012				
.14	-0.188	-0.229	-0.228	-0.224	-0.194	-0.617		.021	-0.024	-0.004	.030	.015							
.15	-0.296	-0.219	-0.219	-0.180	-0.169	-0.169		.014	-0.023	-0.011	-0.002	.011							
.16	-0.199	-0.197	-0.190	-0.180	-0.174	-0.596		.003	-0.006	-0.010	-0.013	-0.017							
.17	-0.179	-0.162	-0.159	-0.159	-0.087			.006	-0.015	-0.008	-0.031			1	-0.010				
.18	-0.149	-0.133	-0.115	-0.087	-0.082	-0.554		.009	-0.009	-0.022	-0.020	-0.001	-0.038	2	-0.004				
.19	-0.113	-0.110	-0.098	-0.087	-0.029			.009	-0.018	-0.016	-0.008	-0.004	-0.004	3	-0.008				
.20	-0.091	-0.070	-0.028					.023	-0.023	-0.015	-0.004	-0.002	-0.002	4	-0.051				
.21	-0.080							.025	-0.025	-0.009	-0.006	-0.013	-0.079	5	-0.001				
.22	-0.044	-0.044	-0.002	-0.009	-0.495			.033	-0.033	-0.005	-0.013	-0.079		6	-0.090				
.23	-0.044	-0.021	-0.021	-0.027	-0.027	-0.049		.049	-0.024	-0.010	-0.018	-0.001		7	-0.064				
.24	-0.017		-0.006	-0.025					-0.015							-0.055			
		$\delta = 0.0$						$M = 0.90$						$\alpha = 5^\circ 0$		$P_{t,j}/P_{\infty} = 5.2$			
X	C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle					
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p				
.01	-0.076	-0.743	-0.807	-0.802	-0.843	-0.736	-0.668	.271	.334					1	.026				
.02	-0.145	-0.805	-0.807	-0.802	-0.843	-0.736	-0.668	.254	.247	.205	.294	.244	.202	2	-0.016				
.03	-0.144	-0.804	-0.807	-0.802	-0.843	-0.736	-0.668	.258	.249	.200	.290	.243	.203	3	-0.027				
.04	-0.166	-0.428	-0.487	-0.487	-0.852	-0.773	-0.683	.196	.283	.143				4	-0.051				
.05	-0.170	-0.187	-0.207	-0.207	-0.207	-0.207	-0.207	.191			.151			5	-0.011				
.06	-0.149	-0.372	-0.207	-0.207	-0.207	-0.207	-0.207	.171	.121	.100	.123			6	-0.090				
.07	-0.212							.288						7	-0.064				
.08	-0.232	-0.202	-0.216	-0.189	-0.714	-0.641		.113	-0.013	.104	.089	.096	.144	8	-0.055				
.09	-0.257	-0.190	-0.210	-0.205	-0.644	-0.643		.058	-0.020	.097	.074	.067	.119	9	-0.088				
.10	-0.173	-0.212	-0.237	-0.239	-0.584	-0.647		.014	-0.062	.067	.056	.048	.077	10	-0.022				
.11	-0.152	-0.232	-0.249	-0.270	-0.446	-0.648		.004	-0.003	.045	.021	.011	.042	11	-0.088				
.12	-0.192	-0.212	-0.259	-0.260	-0.387	-0.637		.016	-0.008	.021	.017	.005	.037	12	-0.075				
.13	-0.178	-0.232	-0.255	-0.241	-0.279	-0.622		.011	-0.024	.008	.051	.005	.017	13	-0.104				
.14	-0.184	-0.228	-0.257	-0.233	-0.211			.021	-0.111	-0.004	.038	.016							
.15	-0.186	-0.234	-0.219	-0.217	-0.180	-0.608		.026	-0.005	.005	.020	.019	.009						
.16	-0.293	-0.220	-0.209	-0.172	-0.164			.007	-0.030	-0.004	.002	.015							
.17	-0.197	-0.201	-0.182	-0.171	-0.123	-0.588		.005	-0.019	-0.002	-0.007	-0.016							
.18	-0.176	-0.163	-0.150	-0.076				.011	-0.023	-0.004	-0.026			1	-0.005				
.19	-0.149	-0.136	-0.097	-0.078	-0.039	-0.542		.013	-0.016	-0.026	-0.009	-0.004	-0.033	2	-0.004	-0.033			
.20	-0.124	-0.109	-0.090	-0.050	-0.010			.009	-0.026	-0.012	-0.002	-0.004		3	-0.009				
.21	-0.111	-0.113	-0.063	-0.020	-0.010			.009	-0.029	-0.007	.003	.009		4	-0.009				
.22	-0.081	-0.037	-0.011	-0.008	-0.018	-0.487		.031	-0.033	-0.003	.015	.016	-0.071	5	-0.001				
.23	-0.056	-0.049	-0.011	-0.008	-0.018	-0.487		.032	-0.039	-0.001	.015			6	-0.005				
.24	-0.043	-0.033	-0.017	-0.017	-0.038	-0.052		.029	-0.029	-0.028	.017			7	-0.017				
.25	-0.017	-0.010		-0.017					-0.019	-0.017				8	-0.005				

TABLE I. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1

5 = 0.0 M = 0.90 $\alpha = 5^\circ$ $P_{t,j}/P_{\infty} = 7.1$								
X C	Wing upper surface stations, fraction of semispan						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.090	-0.791	-0.829	-0.826			1	-0.021
.03	-0.150	-0.774	-0.829	-0.826			2	-0.024
.05	-0.180	-0.851					3	-0.012
.08	-0.179	-0.487	-0.511				4	-0.025
.10	-0.185	-0.205					5	-0.015
.15	-0.222						6	-0.035
.20	-0.222						7	-0.040
.25	-0.254	-0.220	-0.234	-0.200	-0.701	-0.652		
.30	-0.252	-0.208	-0.233	-0.218	-0.701	-0.652		
.35	-0.186	-0.227	-0.240	-0.222	-0.652	-0.457		
.40	-0.144	-0.244	-0.244	-0.253	-0.652	-0.442		
.45	-0.204	-0.227	-0.270	-0.267	-0.378	-0.645		
.50	-0.191	-0.245	-0.263	-0.274	-0.301	-0.635		
.55	-0.199	-0.244	-0.265	-0.245	-0.219			
.60	-0.198	-0.244	-0.232	-0.233	-0.189	-0.621		
.65	-0.305	-0.236	-0.220	-0.186	-0.171			
.68								
.70	-0.208	-0.213	-0.192	-0.183	-0.183	-0.598		
.73	-0.188	-0.181	-0.165		-0.087			
.75					-0.108			
.77					-0.091			
.79						-0.038		
.80	-0.162	-0.150			-0.054	-0.539		
.82			-0.122				-0.027	
.83			-0.101	-0.064				
.85	-0.121	-0.128			-0.022			
.87			-0.106	-0.032				
.88	-0.096						-0.042	
.90			-0.045		-0.001	-0.497		
.91	-0.073			-0.002			-0.047	
.93	-0.061						-0.052	
.95	-0.058	-0.046			-0.024			
.96					-0.059	-0.041		
.97						-0.007		
.98	-0.033					-0.057		
5 = 0.0 M = 0.90 $\alpha = 10^\circ$ $P_{t,j}/P_{\infty} = 1.0$								
X C	Wing upper surface stations, fraction of semispan						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.323	-1.025					1	-0.101
.03	-0.450	-1.016	-0.929	-0.883			2	-0.047
.08	-0.413	-1.135		-0.891	-0.900	-0.624	3	-0.007
.09	-0.405	-1.269		-0.975			4	-0.012
.10	-0.397	-1.352			-0.928	-0.913	5	-0.075
.15	-0.342	-1.115		-1.006		-0.927	6	-0.007
.20	-0.381						7	-0.030
.25	-0.498	-0.233	-0.804	-1.006	-0.937	-0.561		
.30	-0.511	-0.568	-0.714	-1.012	-0.956	-0.552		
.35	-0.279	-0.465	-0.656	-1.018	-0.975	-0.540		
.40	-0.262	-0.418	-0.593	-1.022	-0.580	-0.526		
.45	-0.399	-0.343	-0.554	-1.001	-0.566	-0.512		
.50	-0.324	-0.350	-0.512	-0.967	-0.561	-0.504		
.55	-0.305	-0.325	-0.484	-0.891	-0.531	-0.490		
.60	-0.295	-0.361	-0.421	-0.804	-0.530	-0.479		
.65	-0.386	-0.337	-0.385	-0.649	-0.516			
.68								
.70	-0.317	-0.320	-0.322	-0.581	-0.904	-0.451		
.73	-0.296	-0.258	-0.256		-0.880			
.75					-0.372			
.77					-0.316			
.80	-0.233	-0.185			-0.873	-0.428		
.82			-0.173					
.83			-0.151	-0.211				
.85	-0.151	-0.135			-0.771			
.87	-0.108	-0.135						
.88	-0.097							
.90	-0.063	-0.081		-0.070	-0.686	-0.411		
.91	-0.063	-0.053						
.93	-0.066	-0.036			-0.559			
.95					-0.014			
.97						-0.001		
.98	-0.018					-0.001		

TABLE I.- Continued

TABLE I. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1

		$\delta = 0.0$						$M = 0.90$						$\alpha = 10.0$						$P_{t,j}/P_{\infty} = 7.0$					
$\frac{x}{c}$		Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle											
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p										
.01	-0.327	-1.034	-0.931	-0.898	-0.890	-0.891	-0.892	-0.390	-0.421	-0.422	-0.423	-0.424	-0.425	1	0.125										
.02	-0.419	-1.143	-1.027	-0.978	-0.910	-0.911	-0.921	-0.428	-0.443	-0.424	-0.425	-0.426	-0.427	2	0.094										
.03	-0.413	-1.125	-1.025	-0.978	-0.910	-0.911	-0.921	-0.429	-0.442	-0.424	-0.425	-0.426	-0.427	3	0.060										
.04	-0.402	-1.134	-1.025	-0.978	-0.940	-0.923	-0.943	-0.430	-0.444	-0.425	-0.426	-0.427	-0.428	4	0.182										
.05	-0.396	-1.126	-1.018	-0.978	-0.940	-0.923	-0.943	-0.431	-0.445	-0.426	-0.427	-0.428	-0.429	5	0.099										
.06	-0.387	-1.143	-1.026	-0.978	-0.940	-0.923	-0.943	-0.432	-0.446	-0.427	-0.428	-0.429	-0.430	6	0.029										
.07	-0.427	-0.243	-0.828	-1.022	-0.980	-0.980	-0.980	-0.294	-0.145	-0.227	-0.207	-0.212	-0.223	7	0.047										
.08	-0.517	-0.377	-0.767	-1.027	-0.971	-0.947	-0.947	-0.295	-0.106	-0.204	-0.184	-0.174	-0.215	8	0.169										
.09	-0.275	-0.444	-0.649	-1.025	-0.986	-0.986	-0.986	-0.296	-0.175	-0.111	-0.167	-0.168	-0.164	9	-0.013										
.10	-0.247	-0.417	-0.621	-1.040	-0.988	-0.950	-0.950	-0.302	-0.132	-0.127	-0.150	-0.124	-0.116	10	0.106										
.11	-0.403	-0.349	-0.576	-1.025	-0.977	-0.977	-0.977	-0.303	-0.162	-0.122	-0.124	-0.115	-0.107	11	0.225										
.12	-0.329	-0.356	-0.537	-0.995	-0.970	-0.970	-0.970	-0.304	-0.122	-0.095	-0.101	-0.121	-0.097	12	0.266										
.13	-0.312	-0.366	-0.501	-0.921	-0.959	-0.959	-0.959	-0.305	-0.184	-0.085	-0.075	-0.127	-0.184	13	0.214										
.14	-0.305	-0.348	-0.498	-0.847	-0.941	-0.941	-0.941	-0.306	-0.099	-0.075	-0.085	-0.103	-0.096	14	0.064										
.15	-0.399	-0.347	-0.402	-0.700	-0.926	-0.926	-0.926	-0.307	-0.046	-0.067	-0.076	-0.079	-0.049	15											
.16	-0.321	-0.333	-0.377	-0.609	-0.914	-0.914	-0.914	-0.308	-0.076	-0.057	-0.064	-0.051	-0.021	16											
.17	-0.311	-0.272	-0.268	-0.590	-0.890	-0.890	-0.890	-0.309	-0.037	-0.038	-0.047	-0.022	-0.046	17											
.18	-0.277	-0.272	-0.268	-0.420	-0.867	-0.867	-0.867	-0.310	-0.057	-0.039	-0.028	-0.035	-0.029	18											
.19	-0.249	-0.196	-0.185	-0.852	-0.891	-0.891	-0.891	-0.311	-0.028	-0.026	-0.024	-0.017	-0.017	19											
.20	-0.226	-0.191	-0.159	-0.250	-0.800	-0.800	-0.800	-0.312	-0.039	-0.013	-0.012	-0.016	-0.016	20											
.21	-0.191	-0.141	-0.113	-0.121	-0.170	-0.170	-0.170	-0.313	-0.029	-0.013	-0.012	-0.012	-0.012	21											
.22	-0.191	-0.101	-0.087	-0.087	-0.720	-0.423	-0.423	-0.314	-0.007	-0.006	-0.006	-0.009	-0.117	22											
.23	-0.072	-0.040	-0.040	-0.040	-0.017	-0.017	-0.017	-0.315	-0.008	-0.000	-0.000	-0.010	-0.028	23											
.24	-0.029	-0.024	-0.024	-0.024	-0.024	-0.024	-0.024	-0.316	-0.000	-0.000	-0.000	-0.000	-0.008	24											
.25	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.317	-0.007	-0.006	-0.005	-0.004	-0.004	25											
.26	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.318	-0.007	-0.006	-0.005	-0.004	-0.004	26											
.27	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.319	-0.007	-0.006	-0.005	-0.004	-0.004	27											
.28	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.320	-0.007	-0.006	-0.005	-0.004	-0.004	28											
.29	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.321	-0.007	-0.006	-0.005	-0.004	-0.004	29											
.30	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.322	-0.007	-0.006	-0.005	-0.004	-0.004	30											
.31	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.323	-0.007	-0.006	-0.005	-0.004	-0.004	31											
.32	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.324	-0.007	-0.006	-0.005	-0.004	-0.004	32											
.33	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.325	-0.007	-0.006	-0.005	-0.004	-0.004	33											
.34	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.326	-0.007	-0.006	-0.005	-0.004	-0.004	34											
.35	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.327	-0.007	-0.006	-0.005	-0.004	-0.004	35											
.36	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.328	-0.007	-0.006	-0.005	-0.004	-0.004	36											
.37	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.329	-0.007	-0.006	-0.005	-0.004	-0.004	37											
.38	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.330	-0.007	-0.006	-0.005	-0.004	-0.004	38											
.39	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.331	-0.007	-0.006	-0.005	-0.004	-0.004	39											
.40	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.332	-0.007	-0.006	-0.005	-0.004	-0.004	40											
.41	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.333	-0.007	-0.006	-0.005	-0.004	-0.004	41											
.42	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.334	-0.007	-0.006	-0.005	-0.004	-0.004	42											
.43	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.335	-0.007	-0.006	-0.005	-0.004	-0.004	43											
.44	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.336	-0.007	-0.006	-0.005	-0.004	-0.004	44											
.45	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.337	-0.007	-0.006	-0.005	-0.004	-0.004	45											
.46	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.338	-0.007	-0.006	-0.005	-0.004	-0.004	46											
.47	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.339	-0.007	-0.006	-0.005	-0.004	-0.004	47											
.48	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.340	-0.007	-0.006	-0.005	-0.004	-0.004	48											
.49	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.341	-0.007	-0.006	-0.005	-0.004	-0.004	49											
.50	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.342	-0.007	-0.006	-0.005	-0.004	-0.004	50											
.51	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.343	-0.007	-0.006	-0.005	-0.004	-0.004	51											
.52	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.344	-0.007	-0.006	-0.005	-0.004	-0.004	52											
.53	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.345	-0.007	-0.006	-0.005	-0.004	-0.004	53											
.54	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.346	-0.007	-0.006	-0.005	-0.004	-0.004	54											
.55	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.347	-0.007	-0.006	-0.005	-0.004	-0.004	55											
.56	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.348	-0.007	-0.006	-0.005	-0.004	-0.004	56											
.57	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.349	-0.007	-0.006	-0.005	-0.004	-0.004	57											
.58	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.350	-0.007	-0.006	-0.005	-0.004	-0.004	58											
.59	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.351	-0.007	-0.006	-0.005	-0.004	-0.004	59											
.60	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.352	-0.007	-0.006	-0.005	-0.004	-0.004	60											
.61	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.353	-0.007	-0.006	-0.005	-0.004	-0.004	61											
.62	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.023	-0.354	-0.007	-0.006	-0.005	-0.004	-0.004	62											

TABLE I. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION I

$\delta = 0.0$ $M = 1.00$ $\alpha = 0.0$ $P_{t,1}/P_{\infty} = 5.2$

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	.126	-.021					.146	.095					1	-.015
.03	.098	-.019	-.021	.003			.120	.056	-.169	-.196	-.067		2	-.075
.05	.080	-.008			-.009	.017	.141	.049		-.150	-.051	-.127	3	-.234
.08	.064	-.015		-.006			.080	.050	-.137				4	-.261
.10	.052	-.020			-.034	.012	.070			-.166			5	-.075
.15	.059	-.108	-.015		-.035		.052	-.064	-.027		-.049		6	-.134
.20	.007						.038						7	-.127
.25	-.009	-.044	-.032	-.055	-.024	-.026	.015	-.290	-.137	-.009	-.055	-.104	8	-.127
.30	-.091	-.027	-.028	-.082	-.077	-.139	-.083	-.074	-.017	-.082	-.130		9	-.221
.35	-.031	-.048	-.053	-.075	-.110	-.157	-.137	-.025	-.050	-.042	-.136	-.162	10	-.232
.40	-.006	-.071	-.071	-.089	-.123	-.176	-.149	-.056	-.014	-.095	-.133	-.171	11	-.097
.45	-.046	-.049	-.081	-.095	-.110	-.189	-.036	-.189	-.021	-.114	-.136	-.177	12	-.056
.50	-.029	-.072	-.085	-.114	-.137	-.200	-.086	.022	-.067	-.076	-.145	-.200	13	-.098
.55	-.036	-.087	-.109	-.113	-.141		-.100	.017	-.057	-.091	-.143			
.60	-.050	-.094	-.085	-.127	-.136	-.226	-.029	-.014	-.082	-.103	-.127	-.188		
.65	-.154	-.099	-.110	-.112	-.162		-.000	-.075	-.108	-.116	-.129			
.68												-.136		
.70	-.084	-.099	-.117	-.137	-.178	-.229	-.040	-.067	-.101	-.146		-.188		
.75	-.091	-.093	-.111		-.177		-.082	-.095	-.107			-.153		
.77					-.145							-.158		
.80	-.102	-.101			-.161	-.211	-.075	-.097				-.153		
.82			-.146									-.140		
.83			-.137	-.138								-.130		
.85	-.106	-.139			-.156		-.083	-.106				-.147		
.87			-.128									-.107		
.88			-.124									-.128		
.90			-.123									-.124		
.91			-.103									-.116		
.93			-.107	-.098								-.139		
.95	-.079	-.090			-.089	-.130		-.140	-.139			-.143		
.96					-.060							-.089		
.97					-.080							-.124		

$\delta = 0.0$ $M = 1.00$ $\alpha = 0.0$ $P_{t,1}/P_{\infty} = 5.2$

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	.124	-.028					.137	.072					1	-.028
.03	.094	-.027	-.037	.022	.011	.027	-.055	.044	-.186	-.194	-.048		2	-.075
.05	.076	-.016					.129	.036		-.124	-.048	-.128	3	-.264
.08	.060	-.020		-.004			.073	.040	-.149				4	.005
.10	.047	-.024		-.015	-.003		.063	-.068		-.097			5	-.039
.15	.053	-.110	-.003		-.027		.041	-.068	-.025		-.064		6	-.135
.20	.001						.029						7	-.212
.25	-.011	-.043	-.021	-.036	-.031	-.120	-.007	-.294	-.126	-.004	-.088	-.089	8	.019
.30	-.095	-.039	-.008	-.044	-.049	-.133	-.088	-.151	-.046	-.030	-.092	-.113	9	-.202
.35	-.046	-.047	-.027	-.041	-.105	-.149	-.134	-.023	-.050	-.080	-.052	-.153	10	-.224
.40	-.012	-.065	-.049	-.077	-.117	-.172	-.184	-.052	-.008	-.140	-.178	-.179	11	.056
.45	-.044	-.042	-.084	-.085	-.107	-.183	-.043	-.167	-.060	-.054	-.135	-.189	12	.081
.50	-.026	-.063	-.071	-.107	-.125	-.193	-.060	-.023	-.127	-.059	-.137	-.210	13	.084
.55	-.030	-.072	-.099	-.108	-.129		-.034	-.015	-.037	-.087	-.137			
.60	-.042	-.050	-.079	-.114	-.125	-.217	-.024	-.058	-.085	-.097	-.137	-.184		
.65	-.134	-.088	-.102	-.104	-.153		-.023	-.051	-.098	-.121	-.128			
.68												-.129		
.70	-.075	-.094	-.110	-.127	-.167	-.222	-.040	-.069	-.096	-.128		-.180		
.73	-.083	-.092	-.102		-.164		-.058	-.102	-.092			-.141		
.77				-.134								-.142		
.79				-.125								-.156		
.80	-.097	-.100			-.153	-.205	-.078	-.087				-.145	-.185	
.82				-.135								-.194		
.83				-.126	-.130							-.120	-.197	
.85	-.103	-.137			-.120	-.126	-.079	-.102				-.117	-.191	
.87				-.131								-.105		
.88				-.122								-.114		
.90				-.105								-.112	-.123	-.136
.93				-.100	-.089							-.157	-.117	
.95	-.077	-.084			-.081	-.123		-.141	-.133			-.092	-.136	
.96					-.053							-.079		
.97					-.077							-.112		

TABLE I.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1

TABLE I.- Continued

TABLE I - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1

x c	δ = 0.0 M = 1.00						α = 5.0	p _{t,j} /p _∞ = 7.3	Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90				
Wing upper surface stations, fraction of semispan										
.01	-0.020	-0.620					.422	.457		
.03	-0.071	-0.335	-0.723	-0.699	-0.747	-0.605	-0.640	.475		
.05	-0.084	-0.618					.523	.299		
.08	-0.091	-0.345	-0.456				.242	.321		
.10	-0.097	-0.073					.238			
.13	-0.076	-0.235	-0.146	-0.778	-0.630		.227	.172		
.20	-0.125						.219			
.25	-0.127	-0.209	-0.139	-0.116	-0.615	-0.642	.169	-0.017		
.30	-0.228	-0.193	-0.128	-0.121	-0.596	-0.657	.094	.019		
.35	-0.317	-0.139	-0.149	-0.187	-0.545	-0.680	.018	.054		
.40	-0.129	-0.144	-0.175	-0.221	-0.462	-0.698	.037	.058		
.45	-0.119	-0.118	-0.190	-0.228	-0.373	-0.698	.035	.065		
.50	-0.094	-0.138	-0.200	-0.244	-0.319	-0.698	.066	.060		
.55	-0.098	-0.146	-0.226	-0.244	-0.260		.075	.055		
.60	-0.110	-0.165	-0.212	-0.248	-0.254	-0.706	.065	.051		
.65	-0.225	-0.187	-0.232	-0.234	-0.264		.058	-0.064		
.68							.044	-0.019		
.70	-0.167	-0.193	-0.233	-0.257	-0.262	-0.703	.015	.021		
.73	-0.184	-0.194	-0.231				.025	.025		
.75							.074			
.77							.025			
.79							.060			
.80	-0.200	-0.207					.0302			
.82							.012			
.83							.052			
.85	-0.205	-0.233	-0.244	-0.249			.032	-0.042		
.87	-0.229	-0.242	-0.244				.014	.009		
.88	-0.223						.006			
.90							.004			
.91	-0.203						.014			
.93	-0.199	-0.200					.039			
.95	-0.199	-0.188					.027			
.96							.017			
.97							.051			
.98	-0.148						.063			
Wing lower surface stations, fraction of semispan										
x c	δ = 0.0 M = 1.00						α = 10.0	p _{t,j} /p _∞ = 1.0	Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90				
Wing upper surface stations, fraction of semispan										
.01	-0.178	-0.817					.429	.501		
.03	-0.274	-0.855	-0.926	-0.845	-0.877	-0.810	-0.797	.477	.451	
.05	-0.283	-0.927					.551	.505		
.08	-0.275	-1.025	-0.998				.459	.521		
.10	-0.264	-1.043		-0.887	-0.991		.452			
.15	-0.205	-0.957	-0.991				.426	.398		
.20	-0.292						.409			
.25	-0.248	-0.043	-0.407	-0.971	-0.879	-0.808	.353	.169		
.30	-0.342	-0.213	-0.353	-0.985	-0.916	-0.827	.291	.128		
.35	-0.459	-0.280	-0.322	-0.991	-0.956	-0.846	.211	.132		
.40	-0.163	-0.382	-0.333	-0.997	-0.972	-0.861	.160	.117		
.45	-0.204	-0.316	-0.342	-0.910	-0.978	-0.859	.129	.108		
.50	-0.283	-0.286	-0.348	-0.781	-0.988	-0.855	.129	.145		
.55	-0.261	-0.280	-0.363	-0.626	-0.989	-0.855	.137	.159		
.60	-0.234	-0.284	-0.344	-0.500	-0.992	-0.862	.161	.152		
.65	-0.325	-0.500	-0.360	-0.379	-1.008		.153	.111		
.68							.070			
.70	-0.275	-0.300	-0.361	-0.385	-1.002	-0.857	.132	.105		
.73	-0.291	-0.294	-0.384		-0.993		.114	.084		
.77							.054			
.79							.064			
.80	-0.306	-0.313					.064			
.82							.048			
.85							.056			
.87	-0.311	-0.327					.040			
.88	-0.330	-0.365	-0.322	-0.902			.034			
.90	-0.330	-0.347		-0.833	-0.656		.039			
.91	-0.311			-0.292			.019			
.93	-0.306	-0.294					.002			
.95	-0.283	-0.287					.012			
.96							.007			
.97							.014			
.98	-0.252						.048			

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TABLE I.- Continued

$\delta = 0.0$						$M = 1.00$						$\alpha = 10.0$						$P_{t,j}/P_{\infty} = 5.2$					
z	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle										
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p									
.01	-0.167	-0.809					0.439	0.508					1	0.163									
.03	-0.263	-0.882	-0.917	-0.846			0.484	0.516	0.414	0.386	0.330		2	0.117									
.05	-0.267	-0.922					0.553	0.505		0.345	0.334	0.255	3	0.077									
.08	-0.260	-1.021	-0.986	-0.856	-0.814	-0.794	0.443	0.525	0.341	0.380			4	0.142									
.10	-0.252	-1.035					0.458			0.380			5	0.132									
.15	-0.200	-0.941	-0.984	-0.890	-0.830		0.432	0.401	0.290	0.274			6	0.052									
.20	-0.245						0.415						7	0.054									
.25	-0.243	-0.039	-0.403	-0.976	-0.879	-0.808	0.360	0.177	0.279	0.252	0.244	0.230	8	0.151									
.30	-0.334	-0.206	-0.350	-0.988	-0.920	-0.825	0.294	0.243	0.265	0.233	0.210	0.221	9	-0.001									
.35	-0.450	-0.270	-0.322	-0.996	-0.956	-0.847	0.220	0.155	0.226	0.208	0.191	0.164	10	0.112									
.40	-0.151	-0.370	-0.332	-1.003	-0.975	-0.860	0.174	0.158	0.201	0.168	0.148	0.140	11	0.173									
.45	-0.194	-0.307	-0.344	-0.906	-0.962	-0.856	0.150	0.150	0.175	0.159	0.134	0.136	12	0.183									
.50	-0.274	-0.278	-0.349	-0.791	-0.989	-0.855	0.159	0.164	0.149	0.159	0.124	0.113	13	0.183									
.55	-0.250	-0.277	-0.364	-0.629	-0.993		0.170	0.153	0.132	0.164	0.129												
.60	-0.227	-0.280	-0.346	-0.507	-0.936	-0.843	0.170	0.128	0.152	0.137	0.113	0.096											
.65	-0.326	-0.300	-0.359	-0.384	-1.012		0.143	0.097	0.104	0.112	0.094	0.068											
.70	-0.276	-0.299	-0.360	-0.390	-1.011	-0.856	0.125	0.102	0.100	0.078			0.070		0.081								
.75	-0.296	-0.293	-0.362		-1.000		0.109	0.078	0.086				0.072										
.77					-0.384								0.045										
.79					-0.336								0.061										
.80	-0.307	-0.310				-0.971	-0.756	0.091	0.077				0.063	0.020									
.82						-0.369							0.043										
.83						-0.368	-0.331	-0.913					0.055	0.059	0.047								
.85	-0.312	-0.328					0.076	0.035															
.87						-0.325	-0.362	-0.323	-0.913				0.033	0.051	0.039								
.88							0.035																
.90													0.029	0.036	0.017	0.021	-0.058						
.91													0.010	-0.001									
.93																							
.95	-0.262	-0.287				-0.748		0.018	-0.014						0.018								
.96															-0.027								
.97						-0.224									-0.043								
.98						-0.258									-0.061								

TABLE I.- Continued WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1													
		$\delta = 0.0$			$M = 1.00$			$\alpha = 10.0$		$P_{t,j}/P_{\infty} = 7.3$			
X	C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan					
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.01	-170	-807	-916	-851	-820	-795		.438	.509				
.03	-264	-843	-920	-861	-820	-795		.484	.515	.418	.386	.326	1 .173
.05	-267	-820						.557	.508		.360	.331	2 .130
.08	-264	-1019	-989					.484	.525	.350			3 .103
.10	-255	-1035		-893	-830			.459					4 .212
.15	-201	-943	-989		-890			.434	.403	.298	.322		5 .140
.20	-242							.420					6 .068
.25	-244	-836	-421	-977	-882	-808		.365	.184	.277	.244	.233	7 .086
.30	-333	-202	-361	-993	-921	-825		.299	.149	.220	.226	.205	8 .217
.35	-440	-222	-329	-1016	-958	-844		.227	.126	.216	.208	.187	9 .033
.40	-448	-372	-329	-1016	-977	-851		.178	.108	.197	.167	.144	10 .149
.45	-448	-109	-631	-1020	-985	-856		.156	.156	.156	.135	.128	11 .275
.50	-277	-277	-355	-800	-992	-894		.171	.153	.154	.125	.116	12 .271
.55	-247	-276	-371	-647	-994			.157	.142	.140	.159		13 .266
.60	-221	-279	-353	-523	-995	-861		.163	.126	.122	.137	.122	14 .104
.65	-325	-297	-366	-398	-1013			.142	.082	.086	.115	.089	
.70	-272	-300	-366	-6401	-1010	-852		.120	.112	.112	.1077		15 .058
.75	-292	-294	-368			-1001		.110	.058	.100			16 .072
.77				-366							.023		
.79				-343							.040		
.80	-304	-913		-373		-978	-733	.091	.086				
.82				-371	-336	-918				.008			
.83				-325	-367	-925				.004	-0.058		
.85	-309	-325		-325	-367	-932				.002	.028		
.87				-325	-367	-932				.009	.038	.062	
.88				-322	-344	-892	-651			.001	.063	.045	-0.046
.90				-303	-4294	-824				.002	.018	.051	
.93				-295	-296	-757				.022	.026	.045	
.95	-273	-279		-284	-284	-757					.019	.008	
.96				-228									
.97				-244									
.98													
		$\delta = 0.0$			$M = 1.05$			$\alpha = 0.0$		$P_{t,j}/P_{\infty} = 1.0$			
X	C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan					
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.01	.142	-021						.154	.100				
.03	.102	-025	-087	-048				.127	.056	.233	.156	.243	1 .170
.05	.076	-021		-037	-049	-047		.141	.031	.120	.229	.144	2 .121
.08	.055	-031	-085					.072	.037	.170			3 .251
.10	.043	-031		-091	-080			.071					4 .233
.15	.055	-075	-079		-099			.057	-012	.140	.249		5 .189
.20								.118					6 .186
.25	-009	-112	-116	-111	-393	-130		.020	-020	-106	.267	.178	7 .287
.30	-071	-100	-080	-116	-120	-144		.044	-038	-157	.216	.055	8 .206
.35	-106	-101	-093	-112	-154	-165		.200	-134	-261	.137	.025	9 .522
.40	-051	-147	-111	-141	-161	-166		.066	-075	-164	.046	.164	10 .273
.45	-119	-100	-126	-149	-150	-152		.171	-152	-186	.054	.107	11 .191
.50	-088	-119	-139	-163	-171	-211		.061	-028	-046	.044	.127	12 .194
.55	-093	-124	-135	-165	-172	-172		.149	-025	-050	.073	.134	13 .198
.60	-095	-123	-136	-167	-175	-238		.247	-037	-075	.095	.124	
.65	-193	-136	-157	-153	-193			.139	-022	.096	.110	.14	
.68													
.70	-121	-137	-158	-277	-196	-246		.012	-058	.102	.134		14 .200
.73	-133	-133	-150		-195			.057	-082	.109			
.77				-193									
.79				-176									
.80	-140	-141			-189	-234		.080	-089				
.82				-179									
.83				-174	-167								
.85	-143	-167			-169	-183		.092	-108				
.87				-167									
.88				-161	-152	-173	-222						
.91				-140	-148	-173	-222						
.93				-144	-145	-175							
.95	-117	-130			-150	-175		.134	-142				
.96													
.97				-138									
.98				-126									

TABLE I.- Continued

TABLE I. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1

δ = 0.0 M = 1.05						α = 0.0	P _{t,j} /P _∞ = 9.8	Nacelle	Orifice number	C _p		
x c	Wing upper surface stations, fraction of semispan					Wing lower surface stations, fraction of semispan						
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.01	-0.135	-0.030	-0.086	-0.041	-0.054	-0.104	-0.150	-0.101	-0.122	-0.021	1	-0.160
.03	-0.049	-0.032	-0.086	-0.024	-0.054	-0.104	-0.124	-0.062	-0.187	-0.046	2	-0.146
.05	-0.049	-0.026	-0.087	-0.024	-0.054	-0.104	-0.129	-0.030	-0.121	-0.046	3	-0.124
.08	-0.051	-0.034	-0.087	-0.024	-0.054	-0.104	-0.121	-0.040	-0.167	-0.046	4	-0.111
.10	-0.059	-0.032	-0.077	-0.024	-0.054	-0.104	-0.120	-0.070	-0.167	-0.046	5	-0.109
.15	-0.048	-0.077	-0.077	-0.092	-0.087	-0.082	-0.055	-0.015	-0.124	-0.107	6	-0.170
.20	-0.051	-0.112	-0.113	-0.088	-0.078	-0.100	-0.021	-0.221	-0.124	-0.107	7	-0.037
.25	-0.012	-0.112	-0.113	-0.088	-0.078	-0.100	-0.032	-0.249	-0.096	-0.082	8	-0.225
.30	-0.072	-0.090	-0.087	-0.092	-0.093	-0.113	-0.037	-0.289	-0.099	-0.104	9	-0.179
.35	-0.105	-0.099	-0.092	-0.096	-0.103	-0.121	-0.020	-0.127	-0.020	-0.149	10	-0.197
.40	-0.048	-0.147	-0.107	-0.120	-0.152	-0.201	-0.048	-0.242	-0.060	-0.084	11	-0.166
.45	-0.116	-0.103	-0.109	-0.128	-0.146	-0.214	-0.164	-0.089	-0.199	-0.300	12	-0.194
.50	-0.092	-0.116	-0.115	-0.151	-0.165	-0.220	-0.043	-0.054	-0.306	-0.075	13	-0.154
.55	-0.090	-0.120	-0.137	-0.151	-0.173	-0.227	-0.026	-0.058	-0.127	-0.059	14	-0.118
.60	-0.093	-0.120	-0.120	-0.159	-0.177	-0.243	-0.013	-0.216	-0.041	-0.106	15	-0.142
.65	-0.183	-0.129	-0.136	-0.145	-0.199	-0.228	-0.128	-0.336	-0.127	-0.155	16	-0.218
.70	-0.117	-0.123	-0.144	-0.172	-0.206	-0.246	-0.290	-0.036	-0.171	-0.190	17	-0.233
.75	-0.123	-0.115	-0.138	-0.180	-0.205	-0.205	-0.028	-0.117	-0.176	-0.204	18	-0.209
.77	-0.196	-0.180	-0.196	-0.195	-0.233	-0.233	-0.111	-0.165	-0.202	-0.186	19	-0.247
.79	-0.122	-0.122	-0.170	-0.182	-0.176	-0.195	-0.154	-0.171	-0.157	-0.165	20	-0.172
.82	-0.150	-0.150	-0.165	-0.177	-0.183	-0.195	-0.154	-0.149	-0.131	-0.121	21	-0.193
.83	-0.150	-0.150	-0.162	-0.177	-0.183	-0.195	-0.154	-0.120	-0.117	-0.124	22	-0.261
.85	-0.148	-0.148	-0.149	-0.170	-0.221	-0.221	-0.122	-0.136	-0.119	-0.130	23	-0.151
.91	-0.127	-0.133	-0.158	-0.174	-0.174	-0.174	-0.122	-0.144	-0.117	-0.151	24	-0.144
.93	-0.127	-0.124	-0.154	-0.174	-0.174	-0.174	-0.122	-0.147	-0.117	-0.151	25	-0.147
.95	-0.127	-0.124	-0.154	-0.174	-0.174	-0.174	-0.122	-0.144	-0.117	-0.151	26	-0.144
.96	-0.141	-0.120	-0.141	-0.141	-0.141	-0.141	-0.120	-0.147	-0.117	-0.151	27	-0.147

δ = 0.0 M = 1.05						α = 5.0	P _{t,j} /P _∞ = 1.0	Nacelle	Orifice number	C _p		
x c	Wing upper surface stations, fraction of semispan					Wing lower surface stations, fraction of semispan						
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.01	-0.005	-0.565	-0.746	-0.745	-0.745	-0.745	-0.281	-0.352	-0.134	-0.168	1	-0.018
.03	-0.058	-0.596	-0.746	-0.757	-0.645	-0.643	-0.272	-0.314	-0.103	-0.103	2	-0.034
.05	-0.079	-0.564	-0.757	-0.757	-0.645	-0.643	-0.317	-0.297	-0.103	-0.098	3	-0.181
.08	-0.093	-0.326	-0.383	-0.745	-0.652	-0.667	-0.232	-0.229	-0.069	-0.016	4	-0.175
.10	-0.098	-0.102	-0.745	-0.652	-0.667	-0.667	-0.227	-0.194	-0.075	-0.120	5	-0.021
.15	-0.075	-0.198	-0.236	-0.667	-0.667	-0.667	-0.217	-0.171	-0.075	-0.120	6	-0.180
.20	-0.136	-0.212	-0.216	-0.220	-0.588	-0.650	-0.184	-0.096	-0.039	-0.122	7	-0.203
.25	-0.187	-0.264	-0.226	-0.231	-0.549	-0.665	-0.103	-0.153	-0.103	-0.078	8	-0.171
.30	-0.220	-0.242	-0.240	-0.477	-0.680	-0.680	-0.027	-0.008	-0.071	-0.047	9	-0.264
.35	-0.176	-0.244	-0.248	-0.256	-0.584	-0.696	-0.092	-0.011	-0.107	-0.040	10	-0.185
.40	-0.176	-0.244	-0.248	-0.256	-0.600	-0.696	-0.053	-0.110	-0.079	-0.028	11	-0.137
.45	-0.238	-0.204	-0.252	-0.260	-0.366	-0.696	-0.001	-0.208	-0.049	-0.001	12	-0.195
.50	-0.180	-0.203	-0.250	-0.270	-0.280	-0.696	-0.120	-0.012	-0.029	-0.028	13	-0.140
.55	-0.173	-0.212	-0.258	-0.270	-0.263	-0.696	-0.157	-0.076	-0.017	-0.016	14	-0.024
.60	-0.173	-0.214	-0.239	-0.275	-0.283	-0.696	-0.078	-0.037	-0.008	-0.012	15	-0.031
.65	-0.251	-0.256	-0.263	-0.301	-0.301	-0.301	-0.056	-0.022	-0.012	-0.034	16	-0.036
.70	-0.220	-0.242	-0.258	-0.283	-0.307	-0.696	-0.056	-0.022	-0.012	-0.034	17	-0.057
.75	-0.231	-0.230	-0.254	-0.305	-0.305	-0.305	-0.033	-0.003	-0.021	-0.052	18	-0.052
.77	-0.298	-0.281	-0.303	-0.643	-0.643	-0.643	-0.008	-0.014	-0.059	-0.055	19	-0.055
.79	-0.236	-0.233	-0.271	-0.294	-0.303	-0.643	-0.008	-0.014	-0.049	-0.049	20	-0.057
.82	-0.226	-0.245	-0.263	-0.271	-0.294	-0.643	-0.011	-0.043	-0.062	-0.053	21	-0.045
.85	-0.228	-0.237	-0.262	-0.269	-0.287	-0.606	-0.011	-0.048	-0.055	-0.056	22	-0.110
.87	-0.221	-0.203	-0.251	-0.254	-0.287	-0.606	-0.070	-0.074	-0.055	-0.057	23	-0.050
.88	-0.221	-0.210	-0.237	-0.288	-0.288	-0.606	-0.067	-0.067	-0.061	-0.050	24	-0.050
.91	-0.216	-0.215	-0.243	-0.288	-0.288	-0.606	-0.058	-0.075	-0.055	-0.057	25	-0.050
.93	-0.215	-0.215	-0.243	-0.288	-0.288	-0.606	-0.058	-0.075	-0.055	-0.057	26	-0.050
.95	-0.221	-0.203	-0.237	-0.288	-0.288	-0.606	-0.070	-0.074	-0.061	-0.061	27	-0.050
.97	-0.186	-0.210	-0.210	-0.210	-0.210	-0.210	-0.085	-0.085	-0.067	-0.067	28	-0.050

TABLE I. - Continued

$$\delta = 0.0 \quad M = 1.05 \quad \alpha = 5.0 \quad p_{L,i}/p_{cm} = 5.3$$

$$\delta = 0.0 \quad M = 1.05 \quad \alpha = 5.0 \quad p_{t,i}/p_{\text{rec}} = 9.8$$

TABLE I.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 1

X C	S = 0.0 M = 1.05						$\alpha = 10.0 P_{r,l}/P_{\infty} = 1.0$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.140	-0.767	-0.943	-0.875	-0.894	-0.794	-0.401	-0.509	-0.523	-0.514	-0.362	-0.342	1	0.111
.03	-0.248	-0.798	-0.879	-0.879	-0.894	-0.794	-0.453	-0.514	-0.520	-0.530	-0.320	-0.330	2	0.049
.05	-0.248	-0.879	-0.930	-0.983	-0.983	-0.894	-0.554	-0.502	-0.520	-0.530	-0.320	-0.330	3	-0.137
.08	-0.258	-0.930	-0.983	-0.983	-0.983	-0.894	-0.551	-0.520	-0.520	-0.520	-0.320	-0.330	4	-0.001
.10	-0.259	-0.932	-0.983	-0.983	-0.983	-0.894	-0.558	-0.520	-0.520	-0.520	-0.320	-0.330	5	0.091
.15	-0.210	-0.889	-0.967	-0.911	-0.854	-0.855	-0.416	-0.393	-0.395	-0.395	-0.297	-0.276	6	0.006
.20	-0.254	-0.889	-0.967	-0.911	-0.854	-0.855	-0.400	-0.393	-0.395	-0.395	-0.276	-0.276	7	-0.144
.25	-0.236	-0.282	-0.400	-0.924	-0.877	-0.792	-0.451	-0.111	-0.204	-0.252	-0.247	-0.248	8	-0.029
.30	-0.294	-0.285	-0.360	-0.926	-0.901	-0.810	-0.526	-0.079	-0.230	-0.240	-0.215	-0.279	9	-0.227
.35	-0.396	-0.318	-0.339	-0.930	-0.925	-0.820	-1.03	-0.105	-0.226	-0.216	-0.195	-0.191	10	-0.079
.40	-0.433	-0.304	-0.342	-0.931	-0.936	-0.833	-1.04	-0.084	-0.220	-0.179	-0.158	-0.168	11	0.049
.45	-0.240	-0.316	-0.349	-0.869	-0.926	-0.851	-0.058	-0.022	-0.192	-0.169	-0.149	-0.168	12	0.044
.50	-0.226	-0.349	-0.349	-0.744	-0.943	-0.832	-0.064	-0.137	-0.268	-0.260	-0.139	-0.153	13	0.041
.55	-0.270	-0.334	-0.363	-0.849	-0.944	-0.843	-0.074	-0.169	-0.251	-0.274	-0.240	-0.245		
.60	-0.301	-0.327	-0.343	-0.844	-0.942	-0.843	-0.161	-0.164	-0.143	-0.151	-0.126	-0.126		
.65	-0.360	-0.314	-0.360	-0.831	-0.950	-0.846	-0.166	-0.127	-0.126	-0.125	-0.108	-0.108		
.68	-0.298	-0.309	-0.362	-0.850	-0.954	-0.862	-0.145	-0.126	-0.119	-0.087	-0.117	-0.117		
.70	-0.294	-0.300	-0.359	-0.852	-0.952	-0.862	-0.130	-0.111	-0.097	-0.091	-0.101	-0.108		
.75	-0.294	-0.300	-0.359	-0.846	-0.952	-0.862	-0.130	-0.111	-0.097	-0.098	-0.108	-0.108		
.77	-0.294	-0.300	-0.359	-0.846	-0.952	-0.862	-0.130	-0.111	-0.097	-0.098	-0.108	-0.108		
.79	-0.294	-0.300	-0.359	-0.846	-0.952	-0.862	-0.130	-0.111	-0.097	-0.098	-0.108	-0.108		
.80	-0.298	-0.307	-0.359	-0.846	-0.952	-0.862	-0.112	-0.100	-0.085	-0.085	-0.085	-0.085		
.82	-0.301	-0.315	-0.353	-0.846	-0.952	-0.862	-0.095	-0.091	-0.095	-0.095	-0.095	-0.095		
.83	-0.301	-0.315	-0.353	-0.846	-0.952	-0.862	-0.095	-0.091	-0.095	-0.095	-0.095	-0.095		
.85	-0.301	-0.315	-0.353	-0.846	-0.952	-0.862	-0.095	-0.091	-0.095	-0.095	-0.095	-0.095		
.87	-0.304	-0.305	-0.321	-0.846	-0.952	-0.862	-0.095	-0.091	-0.095	-0.095	-0.095	-0.095		
.88	-0.300	-0.305	-0.321	-0.846	-0.952	-0.862	-0.095	-0.091	-0.095	-0.095	-0.095	-0.095		
.90	-0.279	-0.344	-0.302	-0.787	-0.860	-0.860	-0.065	-0.080	-0.082	-0.085	-0.085	-0.085		
.91	-0.265	-0.311	-0.302	-0.787	-0.860	-0.860	-0.052	-0.046	-0.046	-0.046	-0.046	-0.046		
.95	-0.261	-0.269	-0.225	-0.888	-0.869	-0.869	-0.040	-0.040	-0.040	-0.040	-0.040	-0.040		
.98	-0.252	-0.256	-0.221	-0.805	-0.805	-0.805	-0.005	-0.005	-0.005	-0.005	-0.005	-0.005		

X C	S = 0.0 M = 1.05						$\alpha = 10.0 P_{r,l}/P_{\infty} = 3.0$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.197	-0.766	-0.946	-0.882	-0.882	-0.790	-0.382	-0.504	-0.524	-0.548	-0.368	-0.378	1	0.120
.03	-0.246	-0.813	-0.946	-0.882	-0.882	-0.790	-0.447	-0.512	-0.532	-0.552	-0.393	-0.394	2	0.054
.05	-0.264	-0.894	-0.946	-0.882	-0.882	-0.790	-0.595	-0.496	-0.516	-0.536	-0.393	-0.394	3	-0.046
.08	-0.271	-0.946	-0.969	-0.920	-0.920	-0.826	-0.447	-0.513	-0.529	-0.534	-0.324	-0.320	4	-0.095
.10	-0.273	-0.947	-0.969	-0.920	-0.920	-0.826	-0.448	-0.514	-0.529	-0.534	-0.324	-0.320	5	-0.100
.15	-0.220	-0.907	-0.976	-0.855	-0.855	-0.855	-0.401	-0.393	-0.393	-0.393	-0.276	-0.276	6	0.016
.20	-0.260	-0.894	-0.976	-0.855	-0.855	-0.855	-0.336	-0.112	-0.266	-0.255	-0.251	-0.253	7	-0.109
.25	-0.239	-0.276	-0.391	-0.943	-0.882	-0.790	-0.266	-0.082	-0.244	-0.229	-0.242	-0.242	8	-0.101
.30	-0.296	-0.287	-0.359	-0.949	-0.904	-0.803	-0.466	-0.082	-0.235	-0.213	-0.203	-0.206	9	-0.030
.35	-0.403	-0.314	-0.354	-0.958	-0.982	-0.879	-1.161	-0.118	-0.235	-0.213	-0.193	-0.196	10	0.010
.40	-0.433	-0.322	-0.344	-0.943	-0.982	-0.879	-1.162	-0.103	-0.211	-0.180	-0.162	-0.172	11	-0.117
.45	-0.237	-0.352	-0.350	-0.882	-0.947	-0.829	-0.096	-0.121	-0.183	-0.156	-0.172	-0.172	12	0.121
.50	-0.229	-0.351	-0.350	-0.870	-0.982	-0.831	-0.101	-0.174	-0.156	-0.182	-0.144	-0.155	13	0.126
.55	-0.276	-0.350	-0.364	-0.577	-0.982	-0.831	-0.150	-0.164	-0.144	-0.178	-0.147	-0.147		
.60	-0.299	-0.311	-0.344	-0.446	-0.946	-0.838	-0.176	-0.141	-0.140	-0.154	-0.130	-0.147		
.65	-0.347	-0.309	-0.362	-0.342	-0.980	-0.860	-0.153	-0.109	-0.125	-0.127	-0.114	-0.098		
.70	-0.287	-0.309	-0.362	-0.357	-0.967	-0.856	-0.130	-0.116	-0.117	-0.089	-0.104	-0.117		
.75	-0.289	-0.300	-0.355	-0.329	-0.985	-0.862	-0.120	-0.102	-0.096	-0.098	-0.109	-0.109		
.77	-0.289	-0.300	-0.355	-0.329	-0.985	-0.862	-0.120	-0.102	-0.096	-0.098	-0.109	-0.109		
.79	-0.289	-0.300	-0.355	-0.329	-0.985	-0.862	-0.120	-0.102	-0.096	-0.098	-0.109	-0.109		
.80	-0.299	-0.311	-0.355	-0.318	-0.979	-0.791	-0.108	-0.094	-0.068	-0.089	-0.095	-0.081		
.82	-0.299	-0.311	-0.355	-0.318	-0.979	-0.791	-0.108	-0.094	-0.068	-0.089	-0.095	-0.081		
.85	-0.299	-0.317	-0.352	-0.315	-0.973	-0.793	-0.087	-0.045	-0.050	-0.081	-0.088	-0.102		
.87	-0.306	-0.302	-0.352	-0.315	-0.973	-0.793	-0.087	-0.045	-0.050	-0.081	-0.088	-0.102		
.88	-0.302	-0.341	-0.296	-0.801	-0.853	-0.853	-0.059	-0.074	-0.055	-0.080	-0.084	-0.081		
.90	-0.281	-0.281	-0.305	-0.895	-0.895	-0.895	-0.044	-0.034	-0.044	-0.049	-0.049	-0.084		
.93	-0.286	-0.286	-0.280	-0.895	-0.895	-0.895	-0.044	-0.034	-0.044	-0.049	-0.049	-0.084		
.95	-0.261	-0.268	-0.221	-0.800	-0.800	-0.800	-0.005	-0.037	-0.005	-0.049	-0.049	-0.084		
.96	-0.256	-0.256	-0.221	-0.800	-0.800	-0.800	-0.005	-0.037	-0.005	-0.049	-0.049	-0.084		
.98	-0.256	-0.256	-0.221	-0.800	-0.800	-0.800	-0.005	-0.037	-0.005	-0.049	-0.049	-0.084		

TABLE I - Concluded

x r	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.10	0.32	0.53	0.65	0.74	0.90	0.10	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-+157	-+782					+387	+508					1	+128
.03	-+244	-+829	-+944	-+879			+449	+515	+374	+379	+329		2	+075
.05	-+262	-+892		-+886	-+835	-+790	+550	+500		+355	+331	+266	3	+029
.08	-+271	-+942	-+987				+454	+523	+301				4	+136
.10	-+268	-+944		-+922	-+835		+441			+323			5	+099
.15	-+219	-+807	-+974		-+854		+417	+395	+255		+275		6	+020
.20	-+257						+405						7	-+066
.25	-+237	-+276	-+388	-+950	-+882	-+790	+341	+117	+274	+247	+247	+250	8	+146
.30	-+292	-+281	-+346	-+955	-+909	-+801	+270	+091	+264	+237	+218	+237	9	-+166
.35	-+399	-+304	-+332	-+965	-+937	-+821	+168	+126	+228	+213	+201	+190	10	+074
.40	-+424	-+304	-+337	-+970	-+942	-+835	+119	+119	+202	+176	+159	+166	11	+174
.45	-+229	-+330	-+345	-+885	-+947	-+832	+108	+136	+180	+169	+154	+166	12	+180
.50	-+226	-+344	-+347	-+750	-+952	-+832	+123	+175	+154	+181	+144	+151	13	+187
.55	-+273	-+321	-+362	-+578	-+951		+162	+159	+154	+175	+148			
.60	-+294	-+303	-+343	-+445	-+945	-+841	+175	+139	+146	+153	+129	+143		
.65	-+341	-+304	-+358	-+341	-+961		+153	+110	+121	+131	+105			
.68											+76			
.70	-+281	-+306	-+359	-+357	-+968	-+857	+133	+119	+115	+092		+114		
.73														
.75	-+284	-+296	-+353		-+965		+123	+103	+110		+073			
.77											+081	+102		
.79														
.80	-+294	-+307		-+353		-+764	+111	+096		+041		+109	+078	
.82				-+345					+056	+094				
.83				-+329										
.85	-+294	-+312		-+345	-+318	-+875	+097	+053		+043	+077	+094	+102	
.87				-+302	-+348	-+315				+045				
.88				-+296										
.90				-+275	-+336	-+800	-+651							
.91				-+280	-+300	-+293								
.93				-+256	-+265		-+696	+047	+048		+050	+070	+085	+007
.95						-+276								
.97					-+217					+039		+052	+085	
.98				-+250					+014					

TABLE II
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

		$\delta = 0.0$ $M = 0.80$						$\alpha = 0.0$		$P_{t,j}/P_{\infty} = 1.0$						Nacelle	
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p			
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			1	-0.077	
.01	-0.082	-0.074					-0.078	-0.062					2	-0.160			
.03	-0.044	-0.081	-0.042	-0.043	-0.044	-0.054	-0.056	-0.019	-0.145	-0.145	-0.112	-0.102	3	-0.226			
.05	-0.017	-0.042					-0.027	-0.042			-0.113	-0.088	4	-0.132			
.08	-0.002	-0.041	-0.061				-0.006	-0.033	-0.141		-0.098	-0.102	5	-0.163			
.10	-0.017	-0.046					-0.017				-0.091		6	-0.232			
.15	-0.018	-0.045	-0.074	-0.065	-0.066	-0.036	-0.104	-0.154					7	-0.212			
.20	-0.059						-0.061				-0.071		8	-0.111			
.25	-0.053	-0.031	-0.077	-0.075	-0.056	-0.083	-0.081	-0.102	-0.110	-0.065	-0.058	-0.071	9	-0.199			
.30	-0.064	-0.049	-0.077	-0.070	-0.082	-0.090	-0.096	-0.153	-0.037	-0.067	-0.081	-0.100	10	-0.215			
.35	-0.067	-0.068	-0.085	-0.087	-0.097	-0.095	-0.106	-0.152	-0.046	-0.068	-0.094	-0.104	11	-0.095			
.40	-0.075	-0.070	-0.089	-0.105	-0.099	-0.106	-0.159	-0.151	-0.039	-0.099	-0.110	-0.116	12	-0.094			
.45	-0.092	-0.061	-0.094	-0.099	-0.089	-0.110	-0.150	-0.108	-0.054	-0.092	-0.108	-0.110	13	-0.095			
.50	-0.079	-0.066	-0.094	-0.106	-0.101	-0.116	-0.120	-0.035	-0.068	-0.072	-0.134	-0.108	14	-0.095			
.55	-0.078	-0.068	-0.100	-0.083	-0.092	-0.100	-0.005	-0.070	-0.046	-0.087	-0.084	-0.084	15	-0.095			
.60	-0.081	-0.061	-0.069	-0.086	-0.081	-0.108	-0.051	-0.066	-0.063	-0.068	-0.069	-0.084	16	-0.095			
.65	-0.077	-0.057	-0.073	-0.056	-0.089	-0.038	-0.031	-0.063	-0.063	-0.068	-0.059	-0.079	17	-0.095			
.70	-0.070	-0.047	-0.066	-0.077	-0.072	-0.073	-0.041	-0.015	-0.056	-0.069	-0.063	-0.065	18	-0.095			
.73							-0.079	-0.023	-0.051				19	-0.053			
.75	-0.071	-0.030	-0.052				-0.046	-0.007			-0.068		20	-0.036			
.77							-0.032	-0.013	-0.041	-0.035	-0.021		21	-0.032			
.79							-0.009	-0.030	-0.018				22	-0.005			
.80	-0.061	-0.022					-0.016	-0.016	-0.016	-0.016	-0.016	-0.016	23	-0.016			
.82							-0.006	-0.006	-0.006	-0.006	-0.006	-0.006	24	-0.006			
.83							-0.003	-0.003	-0.003	-0.003	-0.003	-0.003	25	-0.003			
.85	-0.048	-0.024					-0.001	-0.001	-0.001	-0.001	-0.001	-0.001	26	-0.001			
.87							-0.001	-0.001	-0.001	-0.001	-0.001	-0.001	27	-0.001			
.88							-0.013	-0.013	-0.013	-0.013	-0.013	-0.013	28	-0.013			
.90							-0.013	-0.013	-0.013	-0.013	-0.013	-0.013	29	-0.013			
.91							-0.013	-0.013	-0.013	-0.013	-0.013	-0.013	30	-0.013			
.93							-0.014	-0.014	-0.014	-0.014	-0.014	-0.014	31	-0.014			
.95	-0.018	-0.024					-0.018	-0.018	-0.018	-0.018	-0.018	-0.018	32	-0.018			
.96							-0.027	-0.027	-0.027	-0.027	-0.027	-0.027	33	-0.027			
.97							-0.034	-0.034	-0.034	-0.034	-0.034	-0.034	34	-0.034			
.98													35	-0.034			
		$\delta = 0.0$ $M = 0.80$						$\alpha = -0.1$		$P_{t,j}/P_{\infty} = 3.6$						Nacelle	
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p			
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			1	-0.069	
.01	-0.050	-0.073					-0.076	-0.049					2	-0.179			
.03	-0.043	-0.075	-0.076	-0.043	-0.061	-0.051	-0.049	-0.066	-0.155	-0.137	-0.116	-0.116	3	-0.179			
.05	-0.016	-0.073					-0.019	-0.055			-0.104	-0.093	4	-0.091			
.08	-0.001	-0.079	-0.066	-0.045	-0.061	-0.051	-0.002	-0.076	-0.143		-0.089	-0.116	5	-0.150			
.10	-0.022	-0.078					-0.024				-0.141	-0.141	6	-0.204			
.15	-0.022	-0.080	-0.080	-0.067	-0.065	-0.072	-0.043	-0.141			-0.100		7	-0.174			
.20	-0.055						-0.072						8	-0.078			
.25	-0.057	-0.068	-0.083	-0.082	-0.061	-0.082	-0.082	-0.141	-0.067	-0.075	-0.071	-0.075	9	-0.170			
.30	-0.060	-0.085	-0.082	-0.078	-0.086	-0.090	-0.106	-0.182	-0.014	-0.082	-0.094	-0.095	10	-0.182			
.35	-0.074	-0.105	-0.091	-0.092	-0.101	-0.096	-0.112	-0.176	-0.055	-0.086	-0.104	-0.099	11	-0.079			
.40	-0.077	-0.108	-0.096	-0.110	-0.102	-0.105	-0.154	-0.165	-0.068	-0.116	-0.125	-0.126	12	-0.069			
.45	-0.097	-0.099	-0.101	-0.101	-0.101	-0.109	-0.143	-0.134	-0.089	-0.109	-0.121	-0.108	13	-0.059			
.50	-0.083	-0.107	-0.104	-0.10	-0.105	-0.119	-0.104	-0.133	-0.130	-0.088	-0.116	-0.109	14	-0.059			
.55	-0.082	-0.105	-0.109	-0.089	-0.099	-0.095	-0.082	-0.059	-0.098	-0.060	-0.095	-0.060	15	-0.059			
.60	-0.087	-0.101	-0.079	-0.091	-0.085	-0.111	-0.054	-0.074	-0.085	-0.081	-0.075	-0.081	16	-0.059			
.65	-0.084	-0.096	-0.083	-0.066	-0.092	-0.056	-0.056	-0.100	-0.084	-0.077	-0.068	-0.087	17	-0.059			
.70	-0.077	-0.085	-0.073	-0.082	-0.078	-0.080	-0.067	-0.076	-0.072	-0.080	-0.071	-0.065	18	-0.059			
.73							-0.067	-0.079	-0.063		-0.057		19	-0.057			
.75	-0.077	-0.069	-0.062		-0.064		-0.100	-0.079	-0.063		-0.076		20	-0.056			
.77							-0.050						21	-0.056			
.79							-0.041						22	-0.056			
.80	-0.067	-0.063		-0.054		-0.042	-0.039	-0.062	-0.058		-0.060	-0.039	23	-0.056			
.82							-0.042						24	-0.056			
.83							-0.042						25	-0.056			
.85	-0.054	-0.061		-0.042		-0.028	-0.024	-0.046	-0.057		-0.053	-0.043	26	-0.055			
.87							-0.049	-0.026	-0.021		-0.057	-0.036	27	-0.055			
.88							-0.046				-0.053		28	-0.055			
.90							-0.028	-0.013	-0.003	-0.009	-0.047	-0.024	29	-0.008			
.91							-0.023				-0.055	-0.010	30	-0.011			
.93							-0.023				-0.043		31	-0.011			
.95	-0.023	-0.016					-0.017	-0.023			-0.007		32	-0.006			
.96													33	-0.006			
.97													34	-0.006			
.98													35	-0.006			

TABLE II.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

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TABLE II - Continued

TABLE II - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

TABLE II. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

$\delta = 0.0 \quad M = 0.80$							$\alpha = 10.0 \quad P_{t,j}/P_{\infty} = 1.0$							Nacelle	
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	Cp	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			
.01	-1.450	-1.514					.378	.398					1	.185	
.03	-1.540	-1.579	-1.330	-1.026			.401	.426	.384	.325	.264		2	.088	
.05	-1.518	-1.553					.408	.410		.317	.290		3	-.010	
.08	-1.504	-1.554	-1.508				.378	.415	.305				4	.004	
.10	-1.467	-1.664		-1.128			.363						5	.049	
.15	-1.422	-1.739	-1.571				.328	.283	.232				6	-.019	
.20	-1.426						.293						7	.029	
.25	-1.304	-1.451	-1.471	-1.272	-1.946	-1.463	.253	.114	.236	.224	.216	.209	8	.015	
.30	-1.284	-1.448	-1.358	-1.425	-1.921	-1.471	.198	.049	.239	.203	.181	.197	9	-.076	
.35	-1.335	-1.405	-1.298	-1.327	-1.886	-1.480	.162	.034	.219	.177	.161	.139	10	.008	
.40	-1.372	-1.384	-1.295	-1.210	-1.855	-1.482	.084	.018	.198	.140	.131	.104	11	.014	
.45	-1.320	-1.346	-1.282	-1.911	-1.858	-1.477	.046	.036	.165	.131	.117	.093	12	.014	
.50	-1.288	-1.332	-1.259	-1.633	-1.906	-1.484	.060	.104	.134	.127	.096	.067	13	.010	
.55	-1.277	-1.309	-1.239	-1.364	-1.001		.078	.115	.113	.139	.100				
.60	-1.268	-1.282	-1.181	-1.196	-1.947	-1.479	.094	.100	.103	.096	.089	.037			
.65	-1.240	-1.252	-1.163	-0.056	-1.833		.097	.064	.085	.082	.067				
.68												.040			
.70	-1.217	-1.217	-1.135	-0.068	-1.716	-1.474	.072	.069	.074	.056			14	-.012	
.73													15		
.75	-1.190	-1.175	-1.118				.079	.046	.060				16		
.77													17		
.79													18		
.80	-1.162	-1.152					.053						19		
.82													20		
.83													21		
.85	-1.127	-1.139											22		
.87													23		
.88													24		
.90													25		
.91	-0.091												26		
.93	-0.082	-0.068											27		
.95	-0.082	-0.070											28		
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$\delta = 0.0 \quad M = 0.80$							$\alpha = 10.0 \quad P_{t,j}/P_{\infty} = 3.1$							Nacelle	
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	Cp	
	0.19	0.32	0.53	0.85	0.74	0.90	0.19	0.32	0.53	0.85	0.74	0.90			
.01	-1.436	-1.170	-1.286	-1.012			.381	.412	.395	.331	.267		1	.202	
.03	-1.510	-1.241	-1.320	-1.417	-1.908	-1.468	.410	.435	.424	.325	.292	.252	2	.037	
.05	-1.491	-1.318		-1.022	-1.906		.413	.424					3	.057	
.08	-1.471	-1.519	-1.486				.385	.431	.318				4	.054	
.10	-1.459	-1.615		-1.113	-1.913		.356	.297					5	.077	
.15	-1.388	-1.539	-1.539	-1.948	-1.948		.267	.139	.263	.223	.218	.217	6	.016	
.20	-1.286	-1.421	-1.440	-1.270	-1.934	-1.446	.214	.079	.256	.203	.184	.205	7	.013	
.25	-1.264	-1.423	-1.320	-1.417	-1.908	-1.454	.188	.049	.215	.179	.167	.147	8	-.034	
.30	-1.240	-1.424	-1.294	-1.294	-1.874	-1.446	.158	.021	.187	.145	.133	.113	9	.047	
.35	-1.210	-1.425	-1.215	-1.215	-1.848	-1.451	.130	.009	.151	.120	.120	.103	10	.047	
.40	-1.180	-1.426	-1.247	-1.247	-1.897	-1.465	.093	.014	.146	.128	.131	.102	11	.047	
.45	-1.150	-1.427	-1.220	-1.220	-1.840	-1.499	.060	.054	.122	.104	.102	.076	12	.025	
.50	-1.127	-1.310	-1.247	-1.247	-1.897	-1.465	.034	.027	.094	.074	.074	.056	13	.089	
.55	-1.255	-1.286	-1.227	-1.227	-1.840	-1.499	.016						14		
.60	-1.254	-1.259	-1.171	-1.183	-1.947	-1.459</									

TABLE II.- Continued

$$\delta = 0.0 \quad M = 0.80 \quad \alpha = 10.0 \quad P_{t,i}/P_{\infty} = 6.3$$

TABLE II.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

		$\delta = 0.0$		$M = 0.90$		$\alpha = 0.0$		$P_{t,j}/P_{\infty} = 1.0$							
X c	Wing upper surface stations, fraction of semispan	Wing lower surface stations, fraction of semispan						Nacelle							
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp
.01	.113	-.079	-.090	-.093	-.041	-.041	-.049	-.063	-.059	-.035	-.157	-.195	-.125	1	-.068
.03	.069	-.079	-.081	-.075	-.095	-.041	-.049	-.063	-.080	-.062	-.157	-.144	-.119	2	-.175
.05	.040	-.079	-.075	-.072	-.095	-.041	-.049	-.063	-.055	-.076	-.157	-.144	-.119	3	-.359
.08	.019	-.075	-.072	-.072	-.095	-.041	-.049	-.063	-.024	-.058	-.174	-.127	-.119	4	-.123
.10	-.002	-.082	-.083	-.082	-.082	-.062	-.062	-.063	-.001	-.121	-.103	-.125	-.119	5	-.159
.15	-.014	-.078	-.083	-.083	-.070	-.062	-.062	-.063	-.024	-.152	-.172	-.092	-.119	6	-.250
.20	-.059								-.059					7	-.323
.25	-.054	-.067	-.082	-.079	-.067	-.093	-.093		-.071	-.133	-.123	-.062	-.065	8	-.102
.30	-.068	-.069	-.081	-.075	-.095	-.041	-.049	-.063	-.083	-.223	-.038	-.063	-.091	9	-.302
.35	-.088	-.069	-.089	-.092	-.092	-.112	-.109		-.154	-.194	-.057	-.072	-.108	10	-.348
.40	-.082	-.111	-.098	-.119	-.117	-.117	-.121		-.178	-.239	-.027	-.104	-.115	11	-.079
.45	-.096	-.104	-.105	-.111	-.166	-.128	-.136		-.185	-.161	-.046	-.04	-.145	12	-.078
.50	-.085	-.112	-.110	-.121	-.123	-.136	-.136		-.171	-.068	-.071	-.084	-.122	13	-.076
.55	-.086	-.114	-.117	-.103	-.114	-.114	-.114		-.111	-.018	-.081	-.064	-.105		
.60	-.094	-.111	-.088	-.107	-.101	-.125	-.125		-.046	-.024	-.078	-.088	-.084		-.099
.65	-.092	-.107	-.091	-.074	-.107	-.029	-.066		-.029	-.068	-.085	-.072			
.68															-.099
.70	-.088	-.100	-.083	-.097	-.091	-.085			-.039	-.065	-.074	-.087	-.070		
.73															
.75	-.086	-.079	-.076	-.065					-.093	-.072	-.069				-.063
.77															
.79															
.80	-.036	-.058													
.82															
.83															
.85	-.060	-.072													
.87															
.88															
.90															
.91															
.93															
.95	-.018	-.013													
.96															
.97															
.98															
		$\delta = 0.0$		$M = 0.90$		$\alpha = 0.0$		$P_{t,j}/P_{\infty} = 3.2$							
X c	Wing upper surface stations, fraction of semispan	Wing lower surface stations, fraction of semispan						Nacelle							
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp
.01	.115	-.066	-.076	-.083	-.026	-.057	-.058		.166	-.044	-.157	-.151	-.115	1	-.064
.03	.076	-.079	-.083	-.031	-.051	-.057	-.058		.081	-.066	-.157	-.151	-.115	2	-.162
.05	.046	-.070	-.080	-.077	-.080	-.096	-.101		.048	-.079	-.106	-.095	-.124	3	-.226
.08	.024	-.058	-.060						.022	-.059	-.161			4	-.083
.10	.004	-.073	-.073	-.056	-.066				.001					5	-.155
.15	-.008	-.073	-.073	-.073	-.069				.028	-.152	-.159	-.101		6	-.234
.20	-.058								.057					7	-.198
.25	-.048	-.065	-.077	-.084	-.065	-.095			.073	-.136	-.092	-.067	-.073	8	-.067
.30	-.059	-.080	-.077	-.080	-.096	-.101			.085	-.219	-.018	-.076	-.098	9	-.195
.35	-.060	-.101	-.087	-.095	-.101	-.108			.155	-.196	-.021	-.088	-.112	10	-.242
.40	-.075	-.108	-.098	-.114	-.108	-.122			.172	-.209	-.052	-.116	-.134	11	-.055
.45	-.091	-.108	-.105	-.109	-.104	-.120			.169	-.098	-.081	-.116	-.133	12	-.056
.50	-.083	-.108	-.108	-.119	-.120	-.136			.125	-.004	-.104	-.095	-.128	13	-.053
.55	-.080	-.113	-.112	-.102	-.112	-.112			.078	-.039	-.104	-.075	-.111		
.60	-.092	-.107	-.086	-.104	-.100	-.125			.034	-.070	-.095	-.095	-.090		-.093
.65	-.088	-.106	-.091	-.071	-.106				.050	-.018	-.093	-.090	-.076		
.68															
.70	-.085	-.095	-.082	-.095	-.090	-.086			.067	-.090	-.082	-.091	-.067		
.73															
.75	-.084	-.076	-.069						.115	-.093	-.075				
.77															
.79															
.80	-.075	-.060													
.82															
.83															
.85	-.061	-.071													
.87															
.88															
.90															
.91															
.93															
.95	-.018	-.014													
.96															
.97															
.98															

CONFIDENTIAL

TABLE II - Continued

TABLE II - Continued

x c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number		C _p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.55	0.74	0.90	1	2		
	-.078	-.683	-.540	-.807	-.740	-.755	-.721	-.653	.263	.291	.244	.184	.216	.240	.209	.053
.01	-.141	-.540	-.462	-.807	-.740	-.755	-.721	-.653	.253	.291	.244	.184	.216	.240	.209	.048
.05	-.154	-.462	-.420	-.721	-.721	-.721	-.721	-.653	.254	.291	.209	.180	.198	.209	.209	.131
.08	-.165	-.420	-.420	-.721	-.721	-.721	-.721	-.653	.205	.219	.101	.101	.101	.101	.101	.047
.10	-.174	-.428	-.428	-.773	-.773	-.773	-.773	-.653	.187	.187	.187	.140	.140	.140	.140	.060
.15	-.174	-.230	-.215	-.215	-.215	-.215	-.215	-.215	.154	.154	.107	.046	.122	.122	.122	.144
.25	-.224	-.224	-.224	-.224	-.224	-.224	-.224	-.224	.128	.128	.128	.128	.128	.128	.128	.139
.30	-.185	-.217	-.217	-.217	-.217	-.217	-.217	-.217	.108	.108	.017	.076	.104	.106	.137	.023
.35	-.172	-.226	-.226	-.226	-.226	-.226	-.226	-.226	.105	.105	.056	.056	.073	.073	.116	.011
.40	-.204	-.240	-.240	-.247	-.247	-.247	-.247	-.247	.001	.001	.088	.097	.086	.086	.077	.092
.45	-.105	-.217	-.217	-.256	-.256	-.256	-.256	-.256	.059	.059	.109	.094	.035	.024	.037	.012
.50	-.101	-.229	-.229	-.249	-.249	-.249	-.249	-.249	.070	.070	.064	.063	.026	.017	.035	.009
.55	-.103	-.229	-.229	-.246	-.246	-.246	-.246	-.246	.049	.049	.018	.027	.035	.007	.015	.013
.60	-.203	-.223	-.223	-.287	-.287	-.287	-.287	-.287	.021	.021	.059	.023	.019	.016	.005	.013
.65	-.197	-.212	-.212	-.197	-.197	-.197	-.197	-.197	.042	.042	.011	.007	.007	.015	.009	.015
.68	-.188	-.190	-.190	-.170	-.171	-.171	-.171	-.171	.026	.026	.007	.006	.007	.007	.007	.015
.73	-.175	-.153	-.153	-.136	-.136	-.136	-.136	-.136	.002	.002	-.013	.001	.001	.001	.001	.001
.77	-.175	-.153	-.153	-.136	-.136	-.136	-.136	-.136	.002	.002	-.013	.001	.001	.001	.001	.001
.79	-.175	-.153	-.153	-.136	-.136	-.136	-.136	-.136	.020	.020	-.010	.010	.010	.010	.010	.010
.80	-.146	-.128	-.128	-.091	-.091	-.091	-.091	-.091	.005	.005	-.007	.005	.004	.004	.004	.036
.82	-.072	-.072	-.072	-.055	-.055	-.055	-.055	-.055	.005	.005	-.007	.005	.004	.004	.004	.036
.83	-.089	-.089	-.089	-.048	-.048	-.048	-.048	-.048	.004	.004	-.021	.007	.001	.001	.007	.036
.85	-.101	-.111	-.111	-.022	-.022	-.022	-.022	-.022	.004	.004	-.021	.007	.001	.001	.007	.036
.87	-.089	-.089	-.089	-.048	-.048	-.048	-.048	-.048	.004	.004	-.021	.007	.001	.001	.007	.036
.88	-.084	-.084	-.084	-.029	-.029	-.029	-.029	-.029	.021	.021	-.021	.002	.002	.002	.002	.036
.90	-.020	-.020	-.020	.001	.001	.001	.001	.001	.005	.005	-.029	.008	.015	.009	.009	.078
.91	-.051	-.051	-.051	.001	.001	.001	.001	.001	.035	.035	-.039	.008	.015	.009	.009	.078
.93	-.043	-.043	-.043	.001	.001	.001	.001	.001	.035	.035	-.039	.008	.015	.009	.009	.078
.95	-.043	-.030	-.030	.030	.029	.029	.029	.029	.035	.035	-.032	.030	.030	.004	.004	.078
.96	-.014	-.014	-.014	.026	.026	.026	.026	.026	.016	.016	-.022	.030	.030	.004	.004	.078
.97	-.014	-.014	-.014	.014	.014	.014	.014	.014	.016	.016	-.016	.016	.016	.004	.004	.078
.98	-.014	-.014	-.014	.014	.014	.014	.014	.014	.016	.016	-.016	.016	.016	.004	.004	.078

TABLE II-- Continued

TABLE II. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						$\alpha = 10^\circ 0$	$P_{t,j}/P_{\infty} = 1.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.80				
.01	-1.325	-1.042	-1.200	-1.976	-1.993	-1.929	-1.615			
.03	-1.419	-1.087	-1.200	-1.976	-1.993	-1.929	-1.615			
.05	-1.417	-1.177	-1.323	-1.323						
.06	-1.416	-1.329	-1.323							
.10	-1.389	-1.370		-1.030	-1.957					
.15	-1.358	-1.050	-1.418		-1.010					
.20	-1.443									
.25	-1.426	-1.401	-1.685	-1.213	-1.000	-1.567				
.30	-1.213	-1.443	-1.452	-1.249	-1.099	-1.563				
.35	-1.281	-1.415	-1.378	-1.263	-1.015	-1.566				
.40	-1.418	-1.405	-1.377	-1.276	-1.048	-1.560				
.45	-1.324	-1.365	-1.390	-1.164	-1.060	-1.546				
.50	-1.311	-1.374	-1.395	-1.986	-1.041	-1.540				
.55	-1.318	-1.374	-1.392	-1.730	-1.001	-1.535				
.60	-1.346	-1.374	-1.332	-1.493	-1.952	-1.526				
.65	-1.342	-1.363	-1.284	-1.260	-1.918					
.68										
.70	-1.353	-1.319	-1.219	-1.210	-1.866	-1.514				
.73										
.75	-1.294	-1.245	-1.167		-1.816					
.77					-1.107					
.79					-1.095					
.80	-1.226	-1.186			-1.741	-1.486				
.82					-1.123					
.83					-1.112	-1.097				
.85	-1.154	-1.150			-1.112					
.87					-1.095					
.88					-1.091					
.90					-1.081					
.91					-1.071					
.93	-1.088				-1.071					
.95	-1.076	-1.060			-1.536	-1.474				
.96	-1.084	-1.069			-1.411					
.97					-1.053					
.98					-1.042					
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						$\alpha = 10^\circ 0$	$P_{t,j}/P_{\infty} = 3.2$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.80				
.01	-1.332	-1.043	-1.200	-1.983	-1.993	-1.935	-1.433			
.03	-1.420	-1.094	-1.198	-1.983	-1.001	-1.935	-1.433			
.05	-1.421	-1.183								
.08	-1.417	-1.338	-1.323							
.10	-1.390	-1.380		-1.040	-1.963					
.15	-1.362	-1.055	-1.421		-1.013					
.20	-1.447									
.25	-1.428	-1.406	-1.700	-1.223	-1.980	-1.445				
.30	-1.212	-1.444	-1.459	-1.264	-1.951	-1.444				
.35	-1.292	-1.412	-1.385	-1.276	-1.951	-1.489				
.40	-1.421	-1.406	-1.583	-1.276	-1.951	-1.500				
.45	-1.337	-1.368	-1.396	-1.160	-1.939	-1.508				
.50	-1.307	-1.378	-1.395		-1.953	-1.901	-1.514			
.55	-1.320	-1.377	-1.377	-1.648	-1.895					
.60	-1.347	-1.378	-1.280	-1.367	-1.786	-1.523				
.65	-1.344	-1.355	-1.200	-1.132	-1.744					
.68										
.70	-1.320	-1.288	-1.136	-1.120	-1.711	-1.531				
.73										
.75	-1.263	-1.208	-1.116		-1.687					
.77					-1.123					
.79					-1.142					
.80	-1.199	-1.163			-1.654	-1.523				
.82					-1.133					
.83					-1.136	-1.193				
.85	-1.147	-1.149			-1.219	-1.615				
.87					-1.135	-1.411				
.88					-1.128					
.91					-1.135	-1.212				
.93					-1.059	-1.120				
.95	-1.103	-1.087			-1.196	-1.512				
.96										
.97					-1.091					
.98					-1.067					

TABLE II. - Continued

TABLE II. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

$\delta = 0.0 \quad M = 1.00$							$\alpha = 0.0 \quad P_{t,j}/P_{\infty} = 1.0$							Nacelle	
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			
.01	.127	-.039					.125	.015					1	-.031	
.03	.093	-.064	-.048	-.032			.107	-.045	-.045	-.148	-.255		2	-.087	
.05	.075	-.128		-.036	-.034	-.068	.086	-.113		-.152	-.092	-.131	3	-.324	
.08	.059	-.080	-.034				.069	-.019	-.135				4	-.279	
.10	.043	-.046		-.056	-.038		.058						5	-.180	
.15	.031	-.036	-.057		-.049		.016	-.126	-.081		-.057		6	-.197	
.20	-.063						.038						7	-.312	
.25	-.016	-.010	-.059	-.093	-.049	-.129	.025	-.040	-.244	.015	-.046	-.117	8	-.205	
.30	-.017	-.042	-.060	-.093	-.079	-.143	.059	-.107	-.239	.011	-.085	-.140	9	-.307	
.35	-.012	-.068	-.083	-.098	-.139	-.154	.065	-.271	-.106	.019	-.085	-.167	10	-.282	
.40	-.023	-.075	-.099	-.123	-.119	-.171	.073	-.153	-.016	-.057	-.146	-.170	11	-.156	
.45	-.053	-.064	-.110	-.129	-.111	-.192	.193	-.330	.014	-.070	-.124	-.186	12	-.155	
.50	-.050	-.084	-.118	-.123	-.126	-.205	.152	-.158	.016	-.065	-.144	-.201	13	-.154	
.55	-.056	-.100	-.122	-.116	-.149		.274	-.011	-.041	-.064	-.132				
.60	-.076	-.112	-.097	-.123	-.139	-.231	.057	.026	-.057	-.097	-.123	-.195			
.65	-.091	-.121	-.116	-.112	-.168		.009	-.018	-.051	-.110	-.122				
.68												-.130			
.70	-.102	-.119	-.121	-.141	-.180	-.233	.004	-.034	-.090	-.132		-.195			
.75	-.118	-.107	-.113		-.181		.063	-.070	-.100			-.153			
.77												-.151			
.79												-.164			
.80	-.114	-.110					.164	-.216				-.149			
.82												-.152			
.85												-.199			
.88	-.113	-.144													
.87															
.85	-.132	-.132	-.137												
.86															
.90															
.91															
.93															
.95	-.088	-.110	-.104		-.196										
.96															
.97															
.98															
$\delta = 0.0 \quad M = 1.00$							$\alpha = 0.0 \quad P_{t,j}/P_{\infty} = 3.3$							Nacelle	
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			
.01	.157	-.013					.154	.026					1	-.032	
.03	.129	-.038	-.036	-.027	-.033	-.011	.139	-.061	-.145	-.048			2	-.089	
.05	.104	-.099					.099	.022	-.129				3	-.323	
.08	.090	-.008	-.023				.056						4	-.276	
.10	.073	-.003		-.052	-.034		.034	-.114	-.078	-.204	-.053		5	-.172	
.15	.055	-.006	-.057		-.049		.001						6	-.197	
.20	-.016						.001	-.028	-.238	.005	-.054	-.123	7	-.307	
.25	.010	-.035	-.057	-.070	-.094	-.147	.028	-.104	.016	-.009	-.083	-.144	8	-.210	
.30	.009	-.372	-.080	-.083	-.113	-.155	.056	-.264	.068	-.038	-.098	-.169	9	-.300	
.35							.062	-.149	.024	-.084	-.131	-.173	10	-.280	
.40	-.011	-.077	-.056	-.079	-.100	-.171	.184	-.323	.010	-.097	-.131	-.183	11	-.169	
.45	-.051	-.065	-.081	-.098	-.117	-.148	.208	.119	.053	-.075	-.077	-.145	12	-.174	
.50	-.045	-.065	-.081	-.098	-.117	-.160	.157	.032	-.069	-.085	-.142		13	-.174	
.55	-.051	-.096	-.111	-.116	-.160		.030	-.007	-.091	-.109	-.128	-.185			
.60	-.072	-.111	-.088	-.125	-.156	-.230	.068	-.071	-.1C1	-.120	-.134				
.65	-.088	-.108	-.110	-.110	-.167							-.133			
.70	-.093	-.104	-.114	-.136	-.180	-.230	.028	-.076	-.108	-.140		-.191			
.75	-.096	-.099	-.108		-.181		.104	-.101	-.109						
.77															
.79												-.167			
.80	-.103	-.104					.082	-.1C1					15	-.153	
.82															
.83															
.85	-.106	-.137					.080	-.114							
.87															
.88															
.90															
.91															
.93															
.95	-.083	-.089			-.128		.136	-.149			-.097	-.144			
.96															
.97															
.98															

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TABLE II - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

$\delta = 0.0$ $M = 1.00$ $\alpha = 0.0$ $P_{t,j}/P_{\infty} = 5.3$

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	.124	-.061					.119	-.002					1	-.037
.03	.089	-.073	-.047	-.031			.059	-.046	-.074	-.161	-.068		2	-.098
.05	.070	-.141			-.038	-.032	-.087		.001	-.124	-.183	-.058	3	-.327
.08	.055	-.106	-.032				.065	-.041		-.141			4	-.057
.10	.039	-.052		-.059	-.047		.053			-.199			5	-.174
.15	.023	-.035		-.062	-.064		.004	-.136	-.088		-.082		6	-.201
.20	-.078						-.060						7	-.304
.25	-.013	-.014	-.063	-.071	-.090		-.031	-.049	-.249	-.020	-.026	-.128	8	-.047
.30	-.011	-.048		-.066	-.075	-.110	-.055	-.118	.043	-.050	-.058	-.144	9	-.297
.35	-.010	-.071		-.089	-.092	-.133	-.165	-.073	-.276	.049	-.101	-.080	10	-.287
.40	-.021	-.087		-.094	-.104	-.128	-.177	-.079	-.161	-.002	-.116	-.126	11	-.010
.45	-.058	-.073		-.094	-.111	-.142	-.195	-.196	-.296	-.096	-.081	-.145	12	-.004
.50	-.057	-.094		-.104	-.133	-.161	-.213	-.129	-.015	-.071	-.088	-.120	13	-.002
.55	-.061	-.106		-.120	-.130	-.174		-.101	.018	-.060	-.097	-.158		
.60	-.084	-.117		-.098	-.139	-.172	-.237	-.008	-.101	-.110	-.132	-.150		-.195
.65	-.097	-.113		-.123	-.126	-.184		-.033	-.057	-.128	-.139	-.145		
.68												-.144		
.70	-.094	-.117		-.130	-.149	-.199	-.242	-.044	-.091	-.120	-.161			-.200
.73												-.164		
.75	-.107	-.113		-.121		-.198		-.114	-.133	-.119				-.161
.77						-.156						-.178		
.79						-.151						-.170		
.80	-.117	-.117				-.126	-.227	-.104	-.110			-.163	-.200	
.82						-.159						-.157		
.83						-.151	-.151					-.139	-.155	
.85	-.123	-.158				-.142	-.173	-.093	-.123			-.137	-.151	
.87						-.148						-.125		
.88						-.141						-.136		
.90						-.125						-.131	-.146	-.223
.91						-.121						-.131		
.93						-.126						-.162	-.136	
.95	-.098	-.109				-.115						-.168		-.155
.96						-.106						-.108		
.97						-.081						-.106		
.98						-.101						-.106		
												-.131		

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	.154	-.017	-.035	-.022			.154	-.037					1	-.019
.03	.126	-.047					.138	-.025	-.051	-.133	-.068		2	-.078
.05	.102	-.106		-.025	-.043	-.088	.118	-.083	-.084	-.089	-.118		3	-.137
.08	.087	-.038		-.021			.098	.016	-.120				4	-.126
.10	.070	-.009			-.018	-.053	.087						5	-.157
.15	.035	-.001	-.054		-.065		.035	-.102	-.067		-.130		6	-.184
.20	-.030						-.007						7	-.088
.25	-.019	.008	-.050	-.063	-.074	-.135	.003	-.031	-.111	-.111	-.011	-.128	8	-.140
.30	-.014	-.024	-.045	-.049	-.083	-.146	.018	-.096	-.089	-.154	-.048	-.148	9	-.146
.35	-.009	-.059	-.054	-.084	-.110	-.195	.058	-.251	-.067	-.008	-.097	-.171	10	-.217
.40	-.003	-.068	-.066	-.103	-.118	-.169	.056	-.131	-.210	-.036	-.145	-.171	11	-.160
.45	-.010	-.041	-.079	-.110	-.120	-.188	.172	.027	.043	-.084	-.160	-.174	12	-.161
.48	-.038	-.073	-.094	-.120	-.142	-.201	.083	.032	-.098	-.107	-.164	-.174	13	-.182
.50	-.041	-.076	-.113	-.118	-.156		.016	-.207	-.106	-.117	-.158			
.55	-.059	-.084	-.093	-.123	-.152	-.226	.041	-.057	-.128	-.129	-.124	-.155		
.60	-.036	-.093	-.111	-.110	-.166		.042	-.049	-.113	-.107	-.114			
.65												-.115		
.70	-.074	-.102	-.114	-.134	-.181	-.226	-.023	-.128	-.083	-.125		-.170		
.73												-.142		
.75	-.095	-.098	-.107		-.180		-.145	-.068	-.095			-.136		
.77					-.158							-.147		
.79					-.133							-.133		
.80	-.105	-.090				-.161	-.210	-.061	-.078			-.131	-.169	
.82						-.143						-.120		
.83						-.134	-.133					-.119	-.128	
.85	-.102	-.135				-.153		-.061	-.093			-.120	-.131	
.87						-.125	-.133					-.097		
.88						-.118						-.097		
.90						-.109	-.146	-.185				-.115		
.91						-.108						-.115	-.130	-.193
.93						-.104	-.098	-.110				-.133	-.109	-.121
.95	-.076	-.086				-.091	-.134		-.128	-.136		-.091		-.130
.96												-.077		
.97						-.063						-.115		
.98						-.076								

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TABLE II. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

$\delta = 0.0$						$M = 1.00$						$\alpha = 5.0$						$P_{t,j}/P_{\infty} = 1.0$					
X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle										
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p	1	110	2	019					
.01	-0.017	-0.607	-0.423	-0.637	-0.614	-0.644	-0.282	-0.335	-0.283	-0.228	-0.272	-0.270	1	110									
.03	-0.068	-0.423	-0.429	-0.637	-0.624	-0.618	-0.282	-0.283	-0.242	-0.236	-0.229	-0.204	2	019									
.05	-0.054	-0.429	-0.429	-0.637	-0.624	-0.618	-0.277	-0.242	-0.228	-0.229	-0.204	-0.204	3	-0.091									
.06	-0.073	-0.582	-0.607	-0.637	-0.607	-0.644	-0.254	-0.264	-0.146	-0.201	-0.144	-0.144	4	-0.058									
.10	-0.058	-0.357	-0.342	-0.214	-0.214	-0.628	-0.634	-0.210	-0.158	-0.091	-0.144	-0.144	5	-0.017									
.15	-0.085	-0.342	-0.214	-0.214	-0.214	-0.673	-0.673	-0.162	-0.030	-0.169	-0.134	-0.134	6	-0.030									
.20	-0.103	-0.149	-0.132	-0.132	-0.132	-0.655	-0.650	-0.102	-0.058	-0.192	-0.114	-0.090	7	-0.057									
.25	-0.210	-0.113	-0.124	-0.229	-0.229	-0.641	-0.646	-0.072	-0.022	-0.153	-0.050	-0.049	8	-0.070									
.30	-0.092	-0.189	-0.166	-0.231	-0.231	-0.596	-0.679	-0.019	-0.065	-0.117	-0.051	-0.037	9	-0.153									
.35	-0.086	-0.179	-0.190	-0.231	-0.231	-0.500	-0.689	-0.012	-0.017	-0.012	-0.017	-0.019	10	-0.022									
.40	-0.134	-0.161	-0.203	-0.239	-0.239	-0.397	-0.706	-0.032	-0.043	-0.080	-0.039	-0.039	11	-0.083									
.45	-0.127	-0.178	-0.214	-0.248	-0.248	-0.336	-0.706	-0.013	-0.137	-0.049	-0.047	-0.010	12	-0.088									
.50	-0.134	-0.192	-0.231	-0.244	-0.244	-0.280	-0.706	-0.077	-0.095	-0.029	-0.051	-0.012	13	-0.092									
.60	-0.159	-0.193	-0.208	-0.247	-0.247	-0.271	-0.714	-0.106	-0.057	-0.030	-0.021	-0.013	14	-0.004									
.65	-0.160	-0.159	-0.226	-0.239	-0.239	-0.276	-0.714	-0.074	-0.015	-0.008	-0.004	-0.013	15	-0.016									
.70	-0.173	-0.205	-0.230	-0.265	-0.275	-0.711	-0.038	-0.009	-0.011	-0.019	-0.023	-0.026	16										
.75	-0.193	-0.197	-0.223	-0.282	-0.282	-0.606	-0.606	-0.018	-0.004	-0.004	-0.047	-0.031	17										
.77	-0.193	-0.197	-0.223	-0.257	-0.257	-0.632	-0.632	-0.009	-0.013	-0.042	-0.042	-0.047	18										
.79	-0.212	-0.210	-0.250	-0.254	-0.254	-0.271	-0.632	-0.009	-0.013	-0.030	-0.028	-0.010	19	-0.063									
.82	-0.209	-0.246	-0.244	-0.254	-0.254	-0.262	-0.601	-0.001	-0.032	-0.035	-0.026	-0.029	20	-0.018									
.83	-0.226	-0.237	-0.250	-0.262	-0.262	-0.271	-0.601	-0.001	-0.032	-0.035	-0.026	-0.029	21	-0.018									
.85	-0.225	-0.225	-0.225	-0.225	-0.225	-0.262	-0.601	-0.001	-0.034	-0.034	-0.026	-0.029	22	-0.018									
.87	-0.205	-0.220	-0.231	-0.260	-0.260	-0.545	-0.642	-0.042	-0.032	-0.034	-0.018	-0.122	23										
.88	-0.205	-0.191	-0.205	-0.253	-0.253	-0.545	-0.655	-0.070	-0.039	-0.078	-0.015	-0.015	24										
.90	-0.147	-0.147	-0.190	-0.265	-0.265	-0.545	-0.655	-0.079	-0.046	-0.043	-0.043	-0.043	25										
.93	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.154	-0.079	-0.079	-0.079	-0.079	-0.079	26										

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TABLE II.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

GOALS

TABLE II.- Continued

TABLE II. - Continued

TABLE II-- Continued

TABLE II-- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

 $\delta = 0.0 \quad M = 1.05 \quad \alpha = 0.0 \quad P_{t,j}/P_{\infty} = 5.1$

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	.135	.004					.130	.055					1	-.076
.03	.098	-.021	-.081	-.058			.111	-.015	-.071	-.143	-.230		2	-.083
.05	.052	-.032					.088	-.087		-.116	-.152	-.152	3	-.297
.08	.026						.062	-.086	-.136				4	-.214
.10	.041	-.108					.063		-.165				5	-.179
.15	.058	-.102					.044	-.095	-.172		-.108		6	-.182
.20	-.056						-.054						7	-.308
.25	-.002	-.053	-.106	-.106	-.100	-.144	.047	-.130	-.194	-.077	-.058	-.127	8	-.162
.30	-.077	-.061	-.096	-.110	-.118	-.156	.080	-.131	-.142	-.068	-.092	-.134	9	-.310
.35	-.059	-.090	-.102	-.120	-.138	-.166	.165	-.250	-.058	-.028	-.131	-.164	10	-.259
.40	-.028	-.112	-.113	-.131	-.140	-.179	.120	-.148	-.076	-.104	-.159	-.175	11	-.102
.45	-.036	-.057	-.122	-.138	-.156	-.196	.170	-.303	-.022	-.130	-.147	-.185	12	-.090
.50	-.084	-.111	-.134	-.149	-.153	-.208	.172	-.040	-.109	-.121	-.142	-.203	13	-.103
.55	-.035	-.121	-.151	-.148	-.166		.224	-.078	-.103	-.084	-.150			
.60	-.108	-.127	-.125	-.143	-.162	-.232	.050	-.022	-.083	-.123	-.145	-.205		
.65	-.116	-.128	-.138	-.130	-.181		.011	-.130	-.116	-.136	-.157			
.68														
.70	-.116	-.133	-.140	-.154	-.187	-.239	-.069	-.075	-.118	-.150		-.199		
.73														
.75	-.127	-.132	-.131		-.193		-.106	-.110	-.123					
.77														
.79														
.80	-.135	-.130			-.183	-.232	-.087	-.112						
.82														
.83														
.85	-.135	-.156			-.152	-.176	-.093	-.127						
.87														
.88														
.90														
.91														
.93														
.95	-.106	-.121			-.163		-.149	-.166	-.142					
.96														
.97														
.98														

 $\delta = 0.0 \quad M = 1.05 \quad \alpha = 0.0 \quad P_{t,j}/P_{\infty} = 7.6$

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	.133	-.007					.132	.056					1	-.075
.03	.100	-.029	-.095	-.058			.110	-.016	-.079	-.144	-.056		2	-.089
.05	.067	-.103					.085	-.101		-.123	-.076	-.122	3	-.299
.08	.045	-.144	-.073				.059	-.094	-.144				4	-.090
.10	.036	-.113					.064		-.105	-.174			5	-.187
.15	.010	-.085					.055						6	-.191
.20	-.057						.053	-.136	-.203	-.087	-.176	-.136	7	-.301
.25	-.097	-.058	-.114	-.110	-.113	-.151	.066	-.139	-.086	-.146	-.161	-.159	8	-.033
.30	-.079	-.073	-.106	-.109	-.131	-.162	.066	-.255	-.048	-.208	-.077	-.196	9	-.306
.35	-.062	-.098	-.112	-.123	-.152	-.167	.056	-.127	-.154	-.198	-.167	-.212	10	-.247
.40	-.042	-.121	-.123	-.134	-.156	-.187	.048	-.121	-.154	-.271	-.292	-.084	11	-.071
.45	-.102	-.106	-.130	-.138	-.158	-.203	.176	-.031	-.031	-.173	-.231		12	-.052
.50	-.088	-.121	-.138	-.155	-.185	-.222	.176	-.189	-.106	-.134	-.180		13	-.057
.55	-.090	-.130	-.150	-.158	-.198		.027	-.080	-.156	-.151	-.161	-.223		
.60	-.112	-.139	-.129	-.158	-.195	-.244	.027	-.192	-.135	-.133	-.154	-.143		
.65	-.121	-.141	-.145	-.146	-.200		.181	-.052						
.68														
.70	-.122	-.141	-.149	-.171	-.208	-.251	.037	-.142	-.118	-.143		-.200		
.73							.145	-.127	-.131					
.75	-.136	-.132	-.143		-.207									
.77														
.79														
.80	-.136	-.134			-.198	-.243	.108	-.120						
.82														
.83														
.85	-.136	-.165			-.163	-.188	.105	-.139						
.87														
.88														
.90														
.91														
.93														
.95	-.115	-.134			-.146	-.229	.154	-.149	-.134	-.159	-.175	-.217		
.96														
.97														
.98														

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TABLE II. - Continued

TABLE II-- Continued

$$\delta = 0.0 \quad M = 1.05 \quad a = 5.0 \quad p_{t_1}/p_\infty = 7.6$$

TABLE II. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 2

X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.80	Orifice number	C _p
.01	-0.154	-0.780					+0.04	+0.10					1	+212
.03	-0.231	-0.806	-1.080	-0.851			+0.448	+0.501	+0.358	+0.340	+0.330		2	+134
.05	-0.265	-0.900		-0.860	-0.802	-0.786	+0.477	+0.462	+0.320	+0.330	+0.264		3	-0.159
.08	-0.278	-0.977	-1.060				+0.450	+0.466	+0.266				4	+0.15
.10	-0.268	-0.958		-0.885	-0.810		+0.438		+0.306				5	+0.57
.15	-0.215	-0.913	-0.994		-0.837		+0.396	+0.318	+0.177		+0.273		6	+0.11
.20	-0.266						+0.354						7	-0.198
.25	-0.342	-0.242	-0.565	-0.971	-0.878	-0.785	+0.512	+0.122	+0.219	+0.266	+0.249	+0.245	8	+0.12
.30	-0.361	-0.401	-0.329	-0.976	-0.916	-0.797	+0.226	+0.026	+0.263	+0.217	+0.234		9	+205
.35	-0.214	-0.382	-0.310	-0.986	-0.946	-0.813	+0.197	+0.016	+0.271	+0.221	+0.199	+0.192	10	+105
.40	-0.215	-0.387	-0.328	-1.000	-0.958	-0.820	+0.086	+0.018	+0.265	+0.184	+0.168	+0.167	11	+0.64
.45	-0.380	-0.320	-0.335	-0.958	-0.962	-0.821	+0.002	+0.049	+0.227	+0.174	+0.158	+0.168	12	+0.070
.50	-0.301	-0.327	-0.342	-0.858	-0.970	-0.826	+0.032	+0.149	+0.189	+0.178	+0.142	+0.152	13	+0.65
.55	-0.259	-0.334	-0.353	-0.691	-0.970	-0.826	+0.050	+0.199	+0.168	+0.183	+0.145			
.60	-0.285	-0.342	-0.344	-0.518	-0.962	-0.834	+0.175	+0.187	+0.156	+0.147	+0.133	+0.144		
.65	-0.301	-0.327	-0.350	-0.265	-0.981		+0.180	+0.142	+0.138	+0.132	+0.116			
.68												+0.094		
.70	-0.302	-0.323	-0.352	-0.360	-0.991	-0.811	+0.151	+0.128	+0.130	+0.100		+0.114		
.75	-0.303	-0.314	-0.347				+0.160	+0.104	+0.115			+0.100		
.77												+0.107		
.79														
.80	-0.318	-0.327					+0.118	+0.104						
.82														
.85														
.87	-0.309	-0.335	-0.349	-0.313	-0.919		+0.104	+0.069	+0.098	+0.094		+0.099		
.88	-0.335	-0.349	-0.313				+0.076	+0.090	+0.089					
.90														
.91														
.93	-0.309	-0.319												
.95	-0.281	-0.303												
.96														
.97														
.98	-0.277													
X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.80	Orifice number	C _p
.01	-0.167	-0.798					+0.386	+0.304					1	+223
.03	-0.248	-0.822	-1.090	-0.866			+0.435	+0.496	+0.360	+0.372	+0.325		2	+126
.05	-0.279	-0.921		-0.876	-0.817	-0.789	+0.468	+0.459	+0.354	+0.328	+0.263		3	-0.060
.08	-0.292	-1.022	-1.069				+0.445	+0.465	+0.275				4	+110
.10	-0.261	-1.014		-0.903	-0.824		+0.432		+0.328				5	+0.88
.15	-0.222	-0.931	-1.006		-0.849		+0.388	+0.317	+0.216		+0.273		6	+0.08
.20	-0.272						+0.349						7	-0.113
.25	-0.362	-0.263	-0.579	-0.997	-0.891	-0.789	+0.306	+0.113	+0.294	+0.257	+0.249	+0.245	8	+122
.30	-0.367	-0.405	-0.326	-0.995	-0.930	-0.804	+0.222	+0.019	+0.311	+0.236	+0.217	+0.236	9	-0.143
.35	-0.191	-0.393	-0.306	-1.014	-0.961	-0.822	+0.185	+0.045	+0.265	+0.212	+0.200	+0.195	10	+0.029
.40	-0.236	-0.394	-0.327	-1.034	-0.974	-0.826	+0.088	+0.022	+0.234	+0.179	+0.170	+0.168	11	+136
.45	-0.402	-0.318	-0.338	-0.982	-0.980	-0.828	+0.039	+0.117		+0.167	+0.159	+0.168	12	+141
.50	-0.302	-0.333	-0.346	-0.873	-0.989	-0.833	+0.056	+0.217	+0.169	+0.175	+0.141	+0.152	13	+146
.55	-0.257	-0.333	-0.358	-0.703	-0.978		+0.159	+0.180	+0.155	+0.180	+0.145			
.60	-0.266	-0.335	-0.337	-0.536	-0.972	-0.840	+0.185	+0.145	+0.151	+0.145	+0.133	+0.142		
.65	-0.296	-0.324	-0.353	-0.285	-0.999		+0.163	+0.105	+0.135	+0.129	+0.113			
.68												+0.093		
.70	-0.297	-0.327	-0.356	-0.374	-1.015	-0.791	+0.127	+0.113	+0.127	+0.097		+0.110		
.73	-0.307	-0.319	-0.349				+0.141	+0.094	+0.113					
.77														
.79														
.80	-0.320	-0.331					+0.104	+0.093						
.82														
.85	-0.316	-0.341					+0.087	+0.059						
.87	-0.359	-0.353	-0.316				+0.067	+0.064	+0.083	+0.054		+0.096		
.88	-0.340						+0.067							
.90														
.91	-0.316	-0.343		-0.877	-0.644									
.93	-0.312	-0.318												
.95	-0.288	-0.306			-0.790		+0.049	+0.023						
.96					-0.298									
.97					-0.240									
.98	-0.291											-0.010		

TABLE II. - Concluded

TABLE III
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

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TABLE III. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

		$\delta = 0.0$ $M = 0.80$						$\alpha = 0.0$ $P_{t,j}/P_{\infty} = 5.1$							
x/c		Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.057	-.028						.103	.032					1	-.172
.03	.052	-.039	-.113	-.087				.058	.021	-.071	-.100	-.100		2	-.207
.05	.025	-.034		-.072	-.079	-.060		.040	.023		-.085	-.092	-.092	3	-.140
.08	.012	-.044	-.056					.027	.019	-.095				4	-.024
.10	-.001	-.053		-.078	-.072			.006			-.081			5	-.148
.15	-.011	-.067	-.087		-.076			.012	-.008	-.127	-.102			6	-.201
.20	.023							.018						7	-.151
.25	-.037	-.072	-.071	-.073	-.059	-.080		.013	-.119	-.090	-.097	-.071	-.067	8	-.024
.30	-.067	-.080	-.071	-.067	-.083	-.081		.023	-.140	-.081	-.106	-.058	-.091	9	-.134
.35	-.057	-.093	-.086	-.086	-.099	-.098		.010	-.174	-.110	-.112	-.102	-.092	10	-.172
.40	-.062	-.104	-.094	-.112	-.099	-.101		.006	-.183	-.120	-.133	-.111	-.110	11	.005
.45	-.085	-.089	-.101	-.098	-.089	-.098		.012	-.122	-.135	-.129	-.108	-.093	12	.021
.50	-.079	-.096	-.100	-.111	-.099	-.095		.015	-.125	-.103	-.158	-.094	-.102	13	.019
.55	-.077	-.096	-.104	-.104	-.087	-.084		.017	-.121	-.140	-.140	-.050	-.077		
.60	-.084	-.090	-.072	-.091	-.081	-.079		.010	-.108	-.134	-.090	-.066	-.065		
.65	-.075	-.082	-.074	-.056	-.091			.007	-.097	-.164	-.050	-.060	-.054		
.68													-.070		
.70	-.076	-.070	-.067	-.078	-.073	-.060		.009	-.093	-.031	-.031	-.058	-.054		
.73															
.75	-.069	-.052	-.055					.063	-.030	-.024					
.77															
.79															
.80	-.061	-.045						.031	-.019						
.82															
.83															
.85	-.044	-.050						.044	-.018						
.87															
.88															
.90															
.91	-.010														
.93	-.012	-.013													
.95	-.010	-.002													
.96															
.97															
.98															
		$\delta = 0.0$ $M = 0.80$						$\alpha = 0.0$ $P_{t,j}/P_{\infty} = 6.3$							
x/c		Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.056	-.029						.105	.038					1	-.162
.03	.052	-.038	-.120	-.087				.052	.021	-.064	-.087	-.089		2	-.149
.05	.031		-.067	-.081	-.060			.045	.023	-.076	-.084	-.079		3	-.123
.08	.018	-.042	-.085					.025	.023	-.090				4	-.038
.10	-.011	-.051	-.073	-.068	-.071			.018		-.074				5	-.135
.15	-.010	-.051	-.081					.021	-.001	-.118	-.094			6	-.180
.20	.023							.025						7	-.120
.25	-.035	-.057	-.066	-.068	-.054	-.078		.020	-.107	-.083	-.086	-.064	-.063	8	-.027
.30	-.066	-.076	-.068	-.062	-.080	-.089		.016	-.153	-.072	-.095	-.090	-.087	9	-.131
.35	-.052	-.051	-.078	-.082	-.081	-.096		.014	-.126	-.097	-.100	-.090	-.088	10	-.143
.40	-.059	-.101	-.089	-.107	-.061	-.099		.008	-.175	-.110	-.122	-.107	-.109	11	.061
.45	-.081	-.059	-.097	-.064	-.063	-.098		.010	-.120	-.144	-.127	-.108	-.103	12	.110
.50	-.072	-.094	-.096	-.103	-.093	-.097		.016	-.116	-.096	-.143	-.082	-.084	13	.092
.55	-.071	-.091	-.059	-.063	-.076			.012	-.120	-.108	-.122	-.043	-.078		
.60	-.081	-.036	-.068	-.086	-.076	-.070		.008	-.123	-.073	-.064	-.061	-.067		
.65	-.071	-.079	-.071	-.052	-.083			.009	-.143	-.048	-.057	-.050	-.065		
.68															
.70	-.057	-.069	-.064	-.072	-.064	-.057		.003	-.074	-.046	-.046	-.054	-.051		
.73															
.75	-.063	-.050	-.048					.007	-.027	-.032	-.033	-.041			
.77															
.79															
.80	-.054	-.045						.003	-.027	-.017	-.018	-.018	-.025	-.029	
.82															
.83															
.85	-.035	-.047						.009	-.016	-.016	-.020	-.020	-.010		
.87															
.88															
.90															
.92															
.93															
.95	-.006	-.004	-.021												
.96															
.97															
.98															

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TABLE III.- Continued

TABLE III.- Continued

TABLE III.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

$\frac{x}{c}$	$\delta = 0.0 \quad M = 0.80$						$\alpha = 10.0 \quad P_{t,j}/P_{\infty} = 2.9$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.02	-445	-1.136	-0.938	-0.895	-0.810	-0.527	.373	.371	.188	.202	.217	1	-0.011	
.03	-513	-1.188	-0.938	-0.895	-0.825	-0.527	.390	.388	.278	.282	.245	2	-0.040	
.05	-520	-1.273	-0.938	-0.895	-0.825	-0.527	.396	.354	.278	.278	.269	3	-0.058	
.08	-457	-1.450	-0.927	-0.807	-0.433	-0.527	.363	.347	.244	.244	.243	4	-0.006	
.10	-448	-1.570	-0.900	-0.813	-0.514	-0.527	.348	.322	.206	.253	.221	5	.005	
.15	-417	-0.932	-0.969	-0.814	-0.527	-0.527	.323	.322	.206	.221	.221	6	-0.057	
.20	-385	-0.260	-0.943	-0.910	-0.797	-0.527	.298	.246	.193	.188	.202	7	-0.051	
.25	-379	-0.260	-0.943	-0.910	-0.797	-0.527	.265	.246	.193	.188	.202	8	-0.001	
.30	-400	-0.292	-1.021	-0.902	-0.811	-0.527	.271	.137	.180	.161	.172	9	-0.071	
.35	-330	-0.320	-1.082	-0.840	-0.807	-0.433	.230	.062	.139	.141	.155	10	-0.040	
.40	-342	-0.337	-1.061	-0.978	-0.800	-0.419	.167	.026	.120	.113	.125	11	-0.000	
.45	-324	-0.318	-0.974	-1.031	-0.790	-0.402	.085	.032	.093	.110	.121	12	-0.005	
.50	-308	-0.309	-0.328	-1.061	-0.798	-0.391	.083	.064	.063	.115	.103	13	-0.036	
.55	-294	-0.295	-0.555	-1.035	-0.825	-0.365	.068	.037	.059	.141	.110			
.60	-280	-0.262	-0.420	-1.029	-0.875	-0.365	.060	.014	.087	.109	.096			
.65	-250	-0.239	-0.266	-0.945	-0.919	-0.527	.041	-0.031	.107	.091	.077			
.68	-0.57	-0.074	-0.013	-0.059	-0.019	-0.527					.057			
.70	-234	-0.205	-0.132	-0.914	-0.919	-0.527	.023	.011	.117	.070	.004			
.73	-196	-0.164	-0.059	-0.883	-0.527	-0.527	.035	.057	.103		.045			
.75	-196	-0.164	-0.059	-0.883	-0.527	-0.527	.047				.047			
.77	-	-	-	-	-	-								
.79	-	-	-	-	-	-								
.80	-166	-0.135	-0.013	-0.877	-0.325	-0.527	.056	.085	.371	-0.08	-0.027	-0.052		
.82	-0.02	-0.441	-0.02	-0.877	-0.325	-0.527	.068	.044	.068					
.83	-0.02	-0.441	-0.02	-0.877	-0.325	-0.527	.065	.053	.062	.060	.035	.005		
.85	-119	-0.114	-0.000	-0.27	-0.269	-0.863	.037	.037	.037	.037	.035			
.87	-0.000	-0.27	-0.269	-0.863	-0.527	-0.527	.065	.053	.062	.060	.035			
.88	-0.001	-0.27	-0.269	-0.863	-0.527	-0.527	.065	.053	.062	.060	.035			
.90	-	-	-	-	-	-								
.91	-0.049	-0.122	-0.791	-0.308	-0.527	-0.527	.027	.026	.026	.026	.024	-0.143		
.93	-0.054	-0.063	-0.692	-0.527	-0.527	-0.527	.006	.009	.054	.019	.118			
.95	-0.035	-0.013	-0.074	-0.074	-0.527	-0.527	.006	.009	.068	.019	.118			
.96	-0.017	-0.074	-0.074	-0.074	-0.527	-0.527								
.97	-0.017	-0.074	-0.074	-0.074	-0.527	-0.527								
.98	-0.017	-0.074	-0.074	-0.074	-0.527	-0.527								
<hr/>														
$\frac{x}{c}$	$\delta = 0.0 \quad M = 0.80$						$\alpha = 10.0 \quad P_{t,j}/P_{\infty} = 5.1$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-451	-1.146	-0.940	-0.899	-0.815	-0.795	.371	.369	.203	.242	.213	1	-0.005	
.03	-517	-1.196	-0.940	-0.899	-0.831	-0.534	.379	.360	.275	.283	.267	2	-0.050	
.05	-531	-1.270	-0.930	-0.831	-0.527	-0.527	.388	.359	.275	.267	.239	3	-0.038	
.08	-463	-1.510	-0.934	-0.802	-0.802	-0.527	.368	.349	.244			4	-0.049	
.10	-454	-1.587	-0.906	-0.817	-0.817	-0.527	.349	.325	.255			5	-0.009	
.15	-423	-0.936	-0.969	-0.817	-0.817	-0.527	.317	.325	.208	.219		6	-0.049	
.20	-388	-0.263	-0.947	-0.815	-0.795	-0.527	.288	.254	.193	.188	.202	7	-0.033	
.25	-383	-0.263	-0.947	-0.815	-0.795	-0.527	.251	.254	.193	.188	.202	8	-0.054	
.30	-403	-0.276	-1.026	-0.908	-0.813	-0.527	.269	.139	.179	.164	.168	9	-0.056	
.35	-332	-0.317	-1.086	-0.950	-0.812	-0.527	.229	.073	.143	.145	.139	10	-0.023	
.40	-345	-0.357	-1.057	-0.966	-0.802	-0.527	.172	.052	.119	.114	.121	11	-0.083	
.45	-326	-0.308	-0.977	-1.060	-0.800	-0.527	.083	.042	.052	.111	.116	12	-0.087	
.50	-311	-0.308	-0.835	-1.064	-0.803	-0.527	.084	.047	.104	.111	.099	13	-0.092	
.55	-297	-0.293	-0.662	-1.040	-0.829	-0.527	.072	.050	.064	.139	.106			
.60	-285	-0.267	-0.425	-1.033	-0.880	-0.527	.065	.023	.085	.101	.090			
.65	-255	-0.239	-0.286	-0.952	-0.927	-0.527	.040	-0.015	.101	.085	.072			
.68	-0.57	-0.074	-0.074	-0.074	-0.074	-0.527					.050			
.70	-236	-0.202	-0.132	-0.920	-0.922	-0.527	.022	.026	.103	.066	.039	-0.003		
.73	-196	-0.163	-0.060	-0.893	-0.527	-0.527	.036	.070	.091		.039			
.75	-196	-0.163	-0.060	-0.893	-0.527	-0.527								
.77	-	-	-	-	-	-								
.79	-	-	-	-	-	-								
.80	-164	-0.131	-0.016	-0.898	-0.326	-0.527	.053	.080	.056	.041	.023	-0.059		
.82	-0.02	-0.447	-0.02	-0.898	-0.326	-0.527								
.83	-0.02	-0.447	-0.02	-0.898	-0.326	-0.527	.060	.047	.051	.027	.002			
.85	-125	-0.111	-0.004	-0.871	-0.527	-0.527	.060	.056	.046					
.87	-0.092	-0.271	-0.271	-0.871	-0.527	-0.527	.060	.041	.051	.027				
.88	-0.083	-0.271	-0.271	-0.871	-0.527	-0.527	.060	.042	.051	.027				
.90	-	-	-	-	-	-								
.91	-0.063	-0.125	-0.794	-0.313	-0.527	-0.527								
.93	-0.051	-0.064	-0.696	-0.527	-0.527	-0.527								
.95	-0.068	-0.031	-0.015	-0.015	-0.527	-0.527	.004	.009	.048	.017	.043	-0.149		
.96	-0.070	-0.070	-0.070	-0.070	-0.527	-0.527								
.97	-0.020	-0.020	-0.020	-0.020	-0.527	-0.527								
.98	-0.020	-0.020	-0.020	-0.020	-0.527	-0.527								

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TABLE III. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

$\delta = 0.0$						$M = 0.90$	$\alpha = 0.0$	$P_{t,j}/P_{\infty} = 1.0$							
x	c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.107	-.009						.112	.039					1	-.238
.03	.069	-.022	-.138	-.094				.067	.033	-.091	-.155	-.135		2	-.328
.05	.039	-.026		-.080	-.085	-.046		.049	.026		-.109	-.122	-.149	3	-.455
.08	.026	-.037	-.099					.037	.028	-.110				4	-.121
.10	.011	-.049		-.088	-.082			.016						5	-.120
.15	.002	-.068	-.103	-.091				.026						6	-.134
.20	-.020							.009						7	-.391
.25	-.038	-.091	-.087	-.093	-.074	-.085		.031	-.138	-.106	-.123	-.134	-.053	8	-.111
.30	-.046	-.097	-.087	-.083	-.102	-.096		.013	-.230	-.194	-.159	-.169	-.110	9	-.481
.35	-.067	-.104	-.100	-.105	-.119	-.110		.140	-.202	-.115	-.160	-.180	-.114	10	-.438
.40	-.072	-.117	-.112	-.131	-.112	-.112		.128	-.230	-.140	-.205	-.181	-.137	11	-.088
.45	-.100	-.105	-.121	-.119	-.107	-.114		.155	-.180	-.175	-.217	-.153	-.122	12	-.077
.50	-.094	-.115	-.124	-.128	-.122	-.115		.157	-.107	-.222	-.060	-.130	-.121	13	-.083
.55	-.095	-.119	-.126	-.109	-.101			.145	-.139	-.248	-.116	-.101			
.60	-.105	-.114	-.099	-.113	-.097	-.094		.125	-.180	-.285	-.087	-.077	-.089		
.65	-.100	-.103	-.101	-.075	-.107			.140	-.247	-.091	-.066	-.057			
.68															
.70	-.099	-.100	-.089	-.098	-.089	-.069		.177	-.294	-.039	-.054				
.73															
.75	-.093	-.078	-.073					.166	-.082	-.006					
.77															
.79															
.80	-.081	-.066						.041	-.004	-.056	-.001				
.82															
.83															
.85	-.059	-.068						.018		.031	.024	.005	-.017		
.87											.023	.007	-.004		
.88											.020				
.90												.013			
.91												.019			
.93												.002	.022		
.95	-.024	-.004						.028		.024	-.002		.041		
.96													.034		
.97								.036							
.98								.004							
$\delta = 0.0$						$M = 0.90$	$\alpha = 0.1$	$P_{t,j}/P_{\infty} = 2.0$							
x	c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.105	-.023						.115	.045					1	-.230
.03	.063	-.034	-.142	-.103				.067	.030	-.084	-.114	-.116		2	-.312
.05	.039	-.029		-.090	-.096	-.077		.051	.032		-.097	-.112	-.113	3	-.267
.08	.026	-.041	-.102					.035	.031	-.102				4	-.082
.10	.007	-.049		-.092	-.088			.019			-.091			5	-.194
.15	-.003	-.071	-.102	-.094				.026	.015	-.153		-.132		6	-.290
.20	-.024							.029						7	-.260
.25	-.039	-.093	-.088	-.095	-.076	-.100		.026	-.129	-.102	-.116	-.115	-.082	8	-.082
.30	-.079	-.097	-.088	-.089	-.108	-.111		.010	-.206	-.084	-.130	-.140	-.106	9	-.281
.35	-.071	-.106	-.103	-.110	-.119	-.123		.134	-.197	-.169	-.147	-.143	-.111	10	-.339
.40	-.080	-.114	-.116	-.131	-.116	-.125		.127	-.214	-.133	-.187	-.137	-.132	11	-.084
.45	-.099	-.107	-.122	-.122	-.108	-.125		.153	-.175	-.168	-.182	-.124	-.119	12	-.067
.50	-.097	-.115	-.124	-.129	-.126	-.125		.157	-.121	-.208	-.134	-.114	-.119	13	-.062
.55	-.097	-.121	-.124	-.110	-.107			.145	-.131	-.240	-.085	-.093			
.60	-.109	-.114	-.100	-.112	-.102	-.108		.125	-.167	-.168	-.085	-.075	-.090		
.65	-.111	-.109	-.100	-.082	-.112			.132	-.229	-.042	-.055	-.060			
.68															
.70	-.099	-.094	-.092	-.103	-.094	-.075		.153	-.184	-.004	-.054				
.73															
.75	-.094	-.074	-.077					.108	-.021	-.012					
.77															
.79															
.80	-.082	-.066						.043	-.005	-.035	.022				
.82															
.83															
.85	-.063	-.071						.020		.033	-.002				
.87															
.88															
.90															
.91															
.93															
.95	-.022	-.007						.024		-.054	-.024				
.96															
.97								.038							
.98								.001							

TABLE III. - Continued

TABLE III.- Continued

TABLE III. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

x c	δ = 0.0 M = 0.90						α = 5° C	P _{t,j} /P _∞ = 5.0	Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.061	-0.678					.284	.120	1	-1.02
.03	-0.114	-0.705	-0.567	-0.576			.253	.263	2	-1.153
.05	-0.139	-0.873					.235	.242	3	-1.093
.08	-0.131	-0.213	-0.565	-0.584	-0.592	-0.599	.217	.235	4	-0.37
.10	-0.142	-0.150					.203		5	-0.77
.15	-0.152	-0.235					.191	.204	6	-1.153
.20	-0.155						.177		7	-1.02
.25	-0.171	-0.190	-0.488	-0.532	-0.629	-0.542	.160	.095	8	-0.93
.30	-0.231	-0.201	-0.436	-0.477	-0.638	-0.548	.160	.014	9	-1.113
.35	-0.190	-0.226	-0.328	-0.429	-0.624	-0.560	.121	.063	10	-1.096
.40	-0.190	-0.227	-0.242	-0.397	-0.595	-0.555	.117	.087	11	-0.72
.45	-0.201	-0.227	-0.207	-0.315	-0.595	-0.545	.023	.024	12	-0.73
.50	-0.197	-0.218	-0.195	-0.310	-0.598	-0.531	.033	.009	13	-0.68
.55	-0.192	-0.222	-0.192	-0.259	-0.445		.038	.023		
.60	-0.205	-0.211	-0.149	-0.234	-0.368	-0.501	.020	.066		
.65	-0.192	-0.196	-0.158	-0.177	-0.302		.033	.110		
.68							.041	.029		
.70	-0.182	-0.174	-0.134	-0.172	-0.235	-0.457	-0.043	-0.046	.017	-0.010
.73							.014	.034	.046	.008
.75	-0.166	-0.139	-0.130				.001		.014	
.77							.018			
.79							.015			
.80	-0.134	-0.109					.027	.059		
.82							.008			
.83							.021			
.85	-0.091	-0.091					.016	.014		
.87							.004	.029	.019	
.89							.011			
.90							.021			
.91							.015			
.93							.027			
.95	-0.036	-0.009					.016			
.96							.022			
.97							.035			
.98							.042			
							.004			
x c	δ = C=0 M = 0.90						α = 10° C	P _{t,j} /P _∞ = 1.0	Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.325	-0.993	-0.904	-0.869	-0.895	-0.603	.398	.407	1	-0.943
.03	-0.411	-1.036	-0.904	-0.874	-0.884	-0.737	.406	.411	2	-1.112
.05	-0.439	-1.133					.410	.373	3	-1.121
.06	-0.376	-1.272	-0.887				.305	.364	4	-0.39
.10	-0.394	-1.295					.369	.234	5	-0.18
.15	-0.361	-0.991	-0.958	-0.885	-0.882		.339	.343	6	-1.07
.20	-0.351						.317	.195	7	-1.11
.25	-0.360	-0.376	-1.068	-0.937	-0.895	-0.603	.283	.259	8	-0.019
.30	-0.406	-0.354	-1.079	-0.944	-0.933	-0.578	.289	.136	9	-1.126
.35	-0.466	-0.363	-1.152	-0.982	-0.953	-0.556	.242	.049	10	-0.93
.40	-0.402	-0.356	-1.153	-1.043	-0.945	-0.534	.172	.011	11	-0.25
.45	-0.364	-0.343	-1.092	-1.120	-0.929	-0.511	.075	.025	12	-0.03
.50	-0.336	-0.354	-0.956	-1.157	-0.919	-0.486	.076	.068	13	-0.16
.55	-0.336	-0.363	-0.770	-1.140	-0.929		.663	.033		
.60	-0.357	-0.374	-0.543	-1.123	-0.936	-0.438	.056	.001		
.65	-0.363	-0.379	-0.415	-1.054	-0.946		.022	.066		
.68							.078	.396		
.70	-0.378	-0.366	-0.290	-1.007	-0.970	-0.401	.009	.047	.112	.033
.73							.083		.059	
.75	-0.344	-0.313	-0.157				.001	.023	.126	.062
.77							.059			
.79							.062			
.80	-0.274	-0.232					.038	.096	.049	-0.019
.82							.097			
.83							.392			
.85	-0.168	-0.142					.072	.083		
.87							.073			
.88							.070			
.90							.073			
.91							.070			
.92							.039			
.93							.020			
.95	-0.059	-0.029					.019			
.96							.028			
.97							.073			
.98							.021			

TABLE III. - Continued

N c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Crifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	1	-0.016
.01	-0.314	-0.986					.399	.412					2	-0.075
.03	-0.395	-1.030	-0.895	-0.858			.410	.415	.275	.289	.246		3	-0.066
.05	-0.429	-1.120	-0.852	-0.866	-0.869	-0.749	.414	.373	.275	.261	.228		4	-0.008
.08	-0.364	-1.258	-0.877				.382	.368	.240				5	-0.002
.10	-0.384	-1.281		-0.877	-0.869		.369		.253				6	-0.077
.15	-0.351	-0.976	-0.948		-0.880		.341	.346	.203		.213		7	-0.065
.20	-0.341						.314						8	-0.015
.25	-0.350	-0.367	-1.058	-0.926	-0.886	-0.616	.286	.268	.193	.181	.197	.222	9	-0.084
.30	-0.397	-0.346	-1.169	-0.932	-0.926	-0.595	.298	.258	.183	.156	.168	.220	10	-0.052
.35	-0.398	-0.352	-1.142	-0.979	-0.943	-0.567	.249	.222	.143	.153	.161		11	-0.017
.40	-0.392	-0.349	-1.137	-1.035	-0.941	-0.542	.176	.152	.115	.118	.127	.129	12	-0.019
.45	-0.357	-0.333	-1.068	-1.113	-0.927	-0.511	.083	.090	.087	.107	.123	.121	13	-0.024
.50	-0.326	-0.349	-0.945	-1.151	-0.920	-0.490	.083	.074	.056	.115	.105	.097		
.55	-0.326	-0.357	-0.756	-1.136	-0.928	-0.472	.072	.046	.047	.114	.114			
.60	-0.351	-0.369	-0.592	-1.123	-0.936	-0.442	.068	.016	.084	.116	.101	.072		
.65	-0.357	-0.372	-0.410	-1.055	-0.939	-0.404	.040	.037	.118	.102	.087			
.70	-0.374	-0.362	-0.293	-1.007	-0.961	-0.407	.017	.003	.131	.082		.064		
.75	-0.344	-0.313	-0.172		-0.979		.036	.082	.112			.060		
.77				-0.879								.059		
.79				-0.820								.055		
.80	-0.273	-0.208			-1.008	-0.374	.071	.103				.048	-0.019	
.82				-0.034								.074		
.83				-0.005	-0.632							.049	-0.055	
.85	-0.169	-0.138			-1.012		.082	.059				.031		
.87		-0.108	-0.031	-0.432								.051	-0.057	
.88		-0.094			-0.882	-0.357						.047	-0.045	
.90		-0.063		-0.155								.042	-0.036	
.91		-0.047	-0.075									.017	-0.000	-0.101
.93												.067	-0.064	
.95	-0.060	-0.033			-0.756		.015	.011				.029		
.96					-0.026									
.97					-0.094									
.98		-0.007										.011		

TABLE III.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

$\delta = 0.0$						$M = 1.00$						$\alpha = 0.0$						$P_{t,j}/P_{\infty} = 1.0$					
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle										
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp									
.01	.089	.065	-0.115	-0.053	-0.043	-0.026	-0.078	.091	.130	.085	-0.094	-0.059	1	=125									
.02	.066	.051	-0.115	-0.043	-0.026	-0.026	-0.078	.055	.094	-0.095	-0.095	-0.059	2	=155									
.03	.060	.051	-0.115	-0.043	-0.026	-0.026	-0.078	.060	.095	-0.095	-0.095	-0.059	3	=193									
.04	.065	.040	-0.136	-0.036	-0.026	-0.026	-0.078	.062	.104	-0.005	-0.005	-0.047	4	=381									
.10	.059	.032	-0.152	-0.052	-0.046	-0.046	-0.078	.062	.133	-0.133	-0.133	-0.097	5	=497									
.15	.092	.012	-0.24	-0.024	-0.016	-0.016	-0.078	.092	.105	-0.056	-0.056	-0.067	6	=180									
.20	.058	-0.083	-0.24	-0.083	-0.073	-0.073	-0.078	.100	-0.005	-0.005	-0.005	-0.005	7	=340									
.25	.042	-0.029	-0.030	-0.026	-0.016	-0.016	-0.078	.095	-0.185	-0.018	-0.018	-0.018	8	=315									
.30	.013	-0.32	-0.035	-0.029	-0.016	-0.016	-0.078	.095	-0.221	-0.050	-0.041	-0.041	9	=309									
.35	-0.002	-0.57	-0.061	-0.068	-0.112	-0.112	-0.078	.056	-0.173	-0.018	-0.069	-0.131	10	=329									
.40	-0.020	-0.27	-0.083	-0.083	-0.115	-0.115	-0.078	.072	-0.196	-0.028	-0.115	-0.169	11	=273									
.45	-0.039	-0.50	-0.095	-0.099	-0.111	-0.111	-0.078	.028	-0.231	-0.067	-0.134	-0.185	12	=261									
.50	-0.034	-0.071	-0.299	-0.115	-0.131	-0.131	-0.078	.129	-0.050	-0.118	-0.147	-0.205	13	=269									
.55	-0.041	-0.089	-0.109	-0.112	-0.134	-0.134	-0.078	.151	-0.048	-0.158	-0.163	-0.230											
.60	-0.056	-0.097	-0.089	-0.123	-0.122	-0.122	-0.078	.050	-0.084	-0.215	-0.237	-0.252											
.65	-0.081	-0.101	-0.110	-0.105	-0.160	-0.160	-0.078	.055	-0.146	-0.342	-0.284	-0.292											
.68	-0.081	-0.101	-0.110	-0.105	-0.160	-0.160	-0.078	-0.075															
.70	-0.093	-0.099	-0.116	-0.131	-0.172	-0.225	-0.078	.091	-0.234	-0.303	-0.315	-0.304											
.73	-0.098	-0.089	-0.107	-0.137	-0.174	-0.225	-0.078	.172	-0.384	-0.195	-0.216	-0.275											
.75	-0.098	-0.089	-0.107	-0.133	-0.174	-0.225	-0.078	.172	-0.384	-0.195	-0.216	-0.275											
.77	-0.104	-0.101	-0.141	-0.155	-0.215	-0.215	-0.078	.315	-0.212	-0.059	-0.140	-0.160	318										
.80	-0.104	-0.101	-0.133	-0.131	-0.145	-0.145	-0.078	.161	-0.025	-0.022	-0.041	-0.068											
.82	-0.128	-0.126	-0.131	-0.131	-0.145	-0.145	-0.078	.128	-0.019	-0.019	-0.041	-0.068											
.85	-0.128	-0.122	-0.128	-0.131	-0.139	-0.207	-0.078	.013	-0.055	-0.055	-0.055	-0.055											
.87	-0.109	-0.114	-0.114	-0.139	-0.207	-0.207	-0.078	.057	-0.057	-0.057	-0.057	-0.057											
.88	-0.102	-0.092	-0.072	-0.122	-0.207	-0.207	-0.078	-0.007	-0.007	-0.007	-0.007	-0.007											
.90	-0.046	-0.068	-0.068	-0.072	-0.072	-0.072	-0.078	-0.046	-0.046	-0.046	-0.046	-0.046											
$\delta = 0.0$						$M = 1.00$						$\alpha = 0.0$						$P_{t,j}/P_{\infty} = 3.0$					
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle										
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp									
.01	.100	.062	-0.124	-0.080	-0.061	-0.061	-0.109	.107	.120	.088	-0.055	-0.055	1	=140									
.03	.077	.047	-0.124	-0.080	-0.061	-0.061	-0.109	.078	.105	-0.087	-0.088	-0.055	2	=210									
.05	.069	.052	-0.124	-0.080	-0.061	-0.061	-0.109	.079	.114	-0.099	-0.062	-0.142	3	=405									
.08	.073	.043	-0.149	-0.097	-0.111	-0.111	-0.109	.078	.114	-0.012	-0.012	-0.012	4	=402									
.10	.065	.032	-0.124	-0.070	-0.064	-0.064	-0.109	.083	-0.135	-0.074	-0.074	-0.074	5	=119									
.15	.010	-0.042	-0.124	-0.067	-0.067	-0.067	-0.109	.104	.112	-0.062	-0.074	-0.074	6	=196									
.20	.060	-0.041	-0.027	-0.067	-0.068	-0.068	-0.109	.110	-0.009	-0.189	-0.036	-0.078	7	=354									
.25	.041	-0.027	-0.046	-0.070	-0.068	-0.068	-0.109	.101	-0.009	-0.189	-0.036	-0.078	8	=348									
.30	.009	-0.053	-0.046	-0.059	-0.059	-0.059	-0.109	.102	-0.218	-0.345	-0.052	-0.106	9	=316									
.35	-0.005	-0.059	-0.074	-0.089	-0.126	-0.126	-0.109	.157	-0.169	-0.028	-0.080	-0.140	10	=346									
.40	-0.023	-0.050	-0.096	-0.111	-0.133	-0.133	-0.109	.069	-0.201	-0.041	-0.124	-0.179	11	=339									
.45	-0.047	-0.048	-0.108	-0.115	-0.128	-0.128	-0.109	.124	-0.226	-0.078	-0.144	-0.192	12	=338									
.50	-0.038	-0.074	-0.123	-0.135	-0.150	-0.150	-0.109	.120	-0.63	-0.128	-0.156	-0.217	13	=346									
.55	-0.048	-0.091	-0.124	-0.130	-0.149	-0.149	-0.109	.153	-0.055	-0.165	-0.174	-0.240											
.60	-0.073	-0.106	-0.104	-0.136	-0.148	-0.148	-0.109	.055	-0.087	-0.227	-0.249	-0.262											
.65	-0.087	-0.105	-0.124	-0.123	-0.178	-0.178	-0.109	.060	-0.146	-0.349	-0.290	-0.280											
.68	-0.101	-0.103	-0.129	-0.147	-0.187	-0.241	-0.109	.094	-0.232	-0.092	-0.125	-0.225											
.70	-0.106	-0.096	-0.122	-0.189	-0.225	-0.225	-0.109	.175	-0.370	-0.028	-0.124	-0.112											
.75	-0.106	-0.096	-0.122	-0.151	-0.148	-0.189	-0.109	.175	-0.370	-0.028	-0.124	-0.112											
.77	-0.106	-0.096	-0.122	-0.151	-0.148	-0.189	-0.109	.175	-0.370	-0.028	-0.124	-0.112											
.79	-0.106	-0.096	-0.122	-0.151	-0.148	-0.189	-0.109	.175	-0.370	-0.028	-0.124	-0.112											
.80	-0.113	-0.103	-0.153	-0.146	-0.146	-0.169	-0.230	.262	.015	-0.070	-0.063	-0.083	150										
.82	-0.113	-0.136	-0.146	-0.146	-0.161	-0.161	-0.230	.044	-0.026	-0.035	-0.074	-0.091	100										
.85	-0.113	-0.131	-0.139	-0.145	-0.161	-0.161	-0.230	.089	-0.095	-0.040	-0.086	-0.089	105	-0.153									
.87	-0.126	-0.122	-0.125	-0.155	-0.220	-0.220	-0.230	.090	-0.090	-0.078	-0.086	-0.089	065										
.88	-0.109	-0.109	-0.092	-0.139	-0.139	-0.139	-0.230	.089	-0.095	-0.056	-0.056	-0.056	065										
.90	-0.108	-0.091	-0.062	-0.092	-0.092	-0.092	-0.230	.075	-0.048	-0.048	-0.048	-0.048											
.91	-0.105	-0.122	-0.125	-0.155	-0.220	-0.220	-0.230	.055	-0.055	-0.055	-0.055	-0.055											
.93	-0.110	-0.109	-0.092	-0.092	-0.092	-0.092	-0.230	.090	-0.090	-0.078	-0.078	-0.078											
.95	-0.108	-0.091	-0.062	-0.092	-0.092	-0.092	-0.230	.089	-0.095	-0.056	-0.056	-0.056											
.96	-0.075	-0.075	-0.062	-0.092	-0.092	-0.092	-0.230	.075	-0.048	-0.048	-0.048	-0.048											

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TABLE III. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

		$\delta = 0.0$	$M = 1.00$	$\alpha = -0.1$	$p_{t,i}/p_{\infty} = 5:1$										
X	C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	.108	.062	-	.132	-	.094	-	.116	.105	-	.102	-	.115	-	.151
.03	.089	.052	-	.132	-	.072	-	.084	.095	-	.102	-	.077	-	.221
.05	.079	.047	-	.142	-	.072	-	.084	.130	-	.123	-	.078	-	.417
.08	.086	.035	-	.060	-	.075	-	.082	.102	-	.026	-	.165	-	.192
.10	.067	.024	-	.123	-	.153	-	.084	.097	-	.157	-	.124	-	.124
.15	.090	-	.001	-	.056	-	.084	.098	.097	-	.079	-	.089	-	.209
.20	.053	-	-	-	-	-	-	.099	-	-	-	-	-	-	.363
.25	.032	-	.046	-	.059	-	.091	.088	-	.154	-	.027	-	.202	
.30	.001	-	.069	-	.062	-	.078	.116	-	.171	-	.232	-	.069	
.35	.016	-	.079	-	.091	-	.099	-	.043	.182	-	.053	-	.359	
.40	.036	-	.086	-	.110	-	.124	.145	-	.195	-	.211	-	.146	
.45	.059	-	.072	-	.127	-	.127	.144	-	.203	-	.221	-	.256	
.50	.054	-	.091	-	.129	-	.149	.165	-	.213	-	.142	-	.270	
.55	.065	-	.108	-	.138	-	.145	.169	-	.172	-	.075	-	.294	
.60	.093	-	.122	-	.117	-	.153	.163	-	.226	-	.081	-	.285	
.65	.106	-	.129	-	.136	-	.159	.192	-	.078	-	.165	-	.174	
.70	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.150
.73	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.189
.75	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.153
.77	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.127
.79	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.097
.80	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.106
.82	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.145
.83	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.029
.85	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.118
.87	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.131
.88	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.126
.90	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.139
.91	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.168
.93	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.129
.95	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.179
.96	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.096
.97	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.110
.98	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.054
								-	-	-	-	-	-	-	.086
								-	-	-	-	-	-	-	.086
		$\delta = 0.0$	$M = 1.00$	$\alpha = 0.0$	$p_{t,i}/p_{\infty} = 7:1$									Nacelle	
X	C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	.106	.067	-	.111	-	.072	-	.110	.123	-	.095	-	.085	-	.140
.03	.083	.054	-	.111	-	.059	-	.081	.112	-	.102	-	.061	-	.209
.05	.078	.058	-	.112	-	.060	-	.084	.117	-	.098	-	.161	-	.385
.08	.080	.047	-	.045	-	.055	-	.083	.119	-	.107	-	.053	-	.029
.10	.073	.037	-	.069	-	.060	-	.091	-	-	.135	-	.112	-	.112
.15	.094	.016	-	.039	-	.067	-	.111	.128	-	.060	-	.073	-	.194
.20	.065	-	-	-	-	-	-	.111	-	-	-	-	-	-	.343
.25	.044	-	.327	-	.043	-	.070	.147	.101	-	.033	-	.186	-	.014
.30	.017	.052	-	.048	-	.060	-	.105	.122	-	.048	-	.053	-	.317
.35	.003	.036	-	.072	-	.011	-	.059	.157	-	.030	-	.081	-	.343
.40	.021	.046	-	.095	-	.109	-	.064	.161	-	.039	-	.124	-	.052
.45	.042	.052	-	.107	-	.110	-	.203	.215	-	.077	-	.243	-	.148
.50	.035	.071	-	.112	-	.130	-	.121	.121	-	.067	-	.155	-	.204
.55	.046	.088	-	.124	-	.128	-	.146	.155	-	.050	-	.172	-	.145
.60	.074	.099	-	.135	-	.135	-	.146	.156	-	.052	-	.221	-	.091
.65	.084	.104	-	.123	-	.122	-	.156	.142	-	.039	-	.062	-	.154
.68	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.109
.70	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.183
.73	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.161
.75	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.173
.77	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.208
.79	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.218
.80	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.202
.82	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.215
.83	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.052
.85	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.052
.87	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.075
.88	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.062
.90	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.154
.91	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.252
.93	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.110
.95	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.079
.96	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.084
.97	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.079
.98	-	-	-	-	-	-	-	-	-	-	-	-	-	-	.084

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TABLE III.- Continued											
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3											
		$\delta = 0.0$		$M = 1.00$		$\alpha = 5.0$		$P_{t,j}/P_{\infty} = 1.0$			
X	C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan			
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65
.01		.001	-.551					.289	.379		
.02		-.044	-.557	-.528	-.508			.278	.338	.196	
.05		-.061	-.510		-.512	-.543	-.616	.283	.302	.188	.238
.08		-.051	-.127	-.522				.265	.301	.150	.136
.10		-.063	-.070		-.532	-.550		.266		.157	
.15		-.038		-.115	-.519	-.571		.251	.264	.070	.099
.20		-.069						.240			
.25		-.088		-.173	-.368	-.503	-.583	-.630	.225	.145	.086
.30		-.136		-.193	-.314	-.452	-.602	-.647	.223	.10	.053
.35		-.146		-.185	-.234	-.409	-.650	-.669	.173	-.044	.064
.40		-.176		-.189	-.228	-.386	-.590	-.683	.081	-.085	.037
.45		-.176		-.172	-.223	-.346	-.571	-.586	.004	-.036	.001
.50		-.156		-.183	-.228	-.326	-.553	-.588	.025	-.050	.095
.55		-.159		-.194	-.239	-.273	-.521		.004	-.008	.102
.60		-.178		-.203	-.208	-.291	-.485	-.670	.020	-.037	.182
.65		-.184		-.208	-.233	-.258	-.452		.010	-.105	.288
.68											.044
.70		-.200		-.254	-.239	-.290	-.427	-.612	.054	-.209	.056
.73										.016	.006
.75		-.213		-.217	-.232			.393	.147	.227	.030
.77								.273			.009
.79								.270			.014
.80		-.228		-.225				.352	-.516	.236	.001
.82								.259			.056
.85		-.222		-.249	-.259	-.269	-.313		.015	.055	.023
.87								.239	-.249	.262	.044
.88								.239	-.249	.062	.015
.90								.229	-.240	.281	-.426
.91								.221			.059
.93								.219	-.234		.013
.95		-.218		-.196				.193		.001	.015
.96								.175			.008
.97								.135			.024
.98										.003	
											.032
$\delta = 0.0$											
X	C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan			
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65
.01		-.024	-.558					.271	.376		
.03		-.063	-.576	-.528	-.495			.264	.337	.206	
.05		-.082	-.541		-.494	-.510	-.610	.272	.302	.199	.198
.08		-.065	-.192	-.516				.263	.295	.159	.196
.10		-.073	-.063		-.511	-.530		.259		.164	
.15		-.070	-.119	-.522				.251	.265	.081	.102
.20		-.076						.246			.011
.25		-.088	-.172	-.378	-.510	-.566	-.637		.220	.147	.096
.30		-.140	-.194	-.325	-.469	-.586		.221	.013	.105	.059
.35		-.146	-.188	-.259	-.438	-.590		.173	-.038	.071	.032
.40		-.178	-.188	-.223	-.411	-.579		.082	-.082	.047	.119
.45		-.176	-.165	-.212	-.368	-.566	-.698		.001	-.031	.048
.50		-.152	-.179	-.215	-.344	-.555	-.701		.023	-.038	.075
.55		-.155	-.185	-.224	-.301	-.528		.002	-.007	.093	-.098
.60		-.176	-.199	-.195	-.289	-.496		.023	-.036	.173	.051
.65		-.181	-.206	-.222	-.250	-.463		.008	-.101	.041	.066
.68											.007
.70		-.195	-.213	-.227	-.282	-.438	-.616		.050	.208	.107
.73										.042	.003
.75		-.211	-.213	-.220				.404	.137	.004	.088
.77								.258			.010
.79								.254			
.80		-.224	-.222					.365	-.513	.011	
.82								.248			.037
.83								.244			.020
.85		-.219	-.249					.325			.012
.87								.244			.004
.88								.244			.027
.90								.217			.027
.91								.222	-.223		.019
.93								.217			.113
.95		-.220	-.195					.173			.031
.96								.172			.017
.97								.195			.009
.98								.133			.055

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CONTINUATION

TABLE III - Continued

TABLE III. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

X c	5 = 0.0 M = 1.00						$\alpha = 10.0$	$P_{t,j}/P_{\infty} = 1.0$	Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.80			0.19	0.32	0.53	0.65	0.74	0.80		
.01	-.216	-.759	-.713	-.731	-.749	-.752	.441	.499	.322	.337	.285	.252	.252	.252	1	-.001
.03	-.231	-.796	-.707	-.725	-.756	-.756	.472	.495	.322	.316	.284	.252	.252	.252	2	-.098
.05	-.265	-.986	-.707	-.725	-.756	-.756	.489	.460	.322	.316	.284	.252	.252	.252	3	-.238
.08	-.230	-.977	-.707	-.720	-.733	-.733	.470	.445	.296	.290	.218	.218	.218	.218	4	.010
.10	-.231	-.996	-.707	-.720	-.733	-.733	.457	.425	.245	.218	.218	.218	.218	.218	5	.027
.15	-.205	-.809	-.761	-.749	-.749	-.749	.425	.422	.245	.218	.218	.218	.218	.218	6	-.074
.20	-.195	-.759	-.759	-.759	-.759	-.759	.399	.326	.237	.206	.197	.233	.233	.233	7	-.253
.25	-.200	-.246	-.878	-.765	-.749	-.752	.366	.326	.226	.175	.160	.225	.225	.225	8	.039
.30	-.246	-.258	-.940	-.782	-.774	-.766	.369	.190	.109	.184	.158	.175	.175	.175	9	-.250
.35	-.256	-.275	-.905	-.822	-.785	-.774	.312	.259	.252	.152	.098	.137	.146	.146	10	-.067
.40	-.304	-.291	-.852	-.874	-.798	-.782	.325	.252	.121	.104	.100	.139	.12	.12	11	.045
.45	-.283	-.262	-.879	-.908	-.806	-.786	.313	.131	.054	.118	.126	.121	.13	.13	12	.049
.50	-.280	-.271	-.824	-.935	-.834	-.790	.313	.127	.095	.002	.187	.136	.134	.134	13	.043
.55	-.255	-.279	-.716	-.927	-.866	-.798	.312	.052	.046	.129	.124	.110	.110	.110	14	-.074
.60	-.266	-.293	-.532	-.935	-.895	-.798	.302	.022	.116	.138	.138	.090	.090	.090	15	-.074
.65	-.280	-.300	-.390	-.915	-.920	-.920	.302	.022	.116	.138	.138	.090	.090	.090	16	-.074
.68	-.305	-.307	-.292	-.924	-.930	-.930	.025	.134	.159	.116	.086	.086	.086	.086	17	-.074
.72	-.309	-.302	-.252	-.944	-.944	-.944	.021	.071	.167	.090	.095	.095	.095	.095	18	-.074
.75	-.309	-.302	-.252	-.900	-.885	-.885	.021	.071	.167	.090	.095	.095	.095	.095	19	-.074
.77	-.327	-.318	-.261	-.955	-.603	-.603	.082	.142	.130	.125	.088	.078	.078	.078	20	-.026
.80	-.327	-.318	-.253	-.831	-.961	-.961	.130	.124	.102	.071	.063	.063	.063	.063	21	-.026
.82	-.326	-.334	-.265	-.770	-.961	-.961	.130	.124	.102	.071	.063	.063	.063	.063	22	-.026
.85	-.326	-.326	-.325	-.265	-.770	-.770	.130	.124	.102	.071	.063	.063	.063	.063	23	-.026
.87	-.326	-.326	-.325	-.265	-.770	-.770	.130	.124	.102	.071	.063	.063	.063	.063	24	-.026
.88	-.326	-.326	-.325	-.265	-.770	-.770	.130	.124	.102	.071	.063	.063	.063	.063	25	-.026
.90	-.308	-.257	-.688	-.959	-.544	-.544	.080	.082	.051	.051	.037	.040	.040	.040	26	-.026
.91	-.308	-.263	-.688	-.959	-.544	-.544	.080	.050	.053	.053	.010	.009	.009	.009	27	-.026
.93	-.303	-.283	-.593	-.848	-.848	-.848	.051	.012	.012	.012	.010	.009	.009	.009	28	-.026
.95	-.282	-.283	-.593	-.848	-.848	-.848	.021	.021	.021	.021	.021	.021	.021	.021	29	-.026
.97	-.237	-.257	-.593	-.848	-.848	-.848	.021	.021	.021	.021	.021	.021	.021	.021	30	-.026
.98	-.237	-.237	-.237	-.237	-.237	-.237	.021	.021	.021	.021	.021	.021	.021	.021	31	-.026
X c	5 = 0.0 M = 1.00						$\alpha = 10.0$	$P_{t,j}/P_{\infty} = 3.1$	Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.80			0.19	0.32	0.53	0.65	0.74	0.80		
.01	-.216	-.770	-.723	-.696	-.732	-.758	.434	.497	.329	.340	.290	.252	.252	.252	1	.013
.03	-.239	-.801	-.723	-.702	-.732	-.758	.467	.492	.324	.324	.294	.252	.252	.252	2	-.073
.05	-.248	-.991	-.711	-.711	-.732	-.758	.488	.456	.292	.292	.292	.252	.252	.252	3	-.056
.08	-.227	-.981	-.711	-.711	-.732	-.758	.464	.445	.292	.292	.292	.252	.252	.252	4	.063
.10	-.236	-.993	-.713	-.734	-.734	-.758	.455	.426	.253	.236	.236	.236	.236	.236	5	.039
.15	-.239	-.821	-.763	-.746	-.746	-.758	.419	.420	.253	.236	.236	.236	.236	.236	6	-.059
.20	-.200	-.759	-.759	-.759	-.759	-.759	.393	.326	.226	.218	.220	.230	.230	.230	7	-.080
.25	-.234	-.253	-.880	-.763	-.749	-.752	.361	.325	.245	.218	.220	.230	.230	.230	8	.069
.30	-.248	-.262	-.943	-.777	-.772	-.761	.369	.191	.233	.187	.190	.224	.224	.224	9	-.098
.35	-.256	-.276	-.907	-.814	-.789	-.774	.308	.110	.193	.159	.182	.173	.173	.173	10	-.034
.40	-.309	-.293	-.694	-.744	-.797	-.786	.230	.064	.164	.127	.156	.144	.144	.144	11	.077
.45	-.207	-.264	-.881	-.903	-.810	-.786	.129	.085	.126	.134	.153	.139	.12	.12	12	.077
.50	-.265	-.272	-.823	-.932	-.838	-.790	.133	.134	.080	.149	.135	.120	.120	.120	13	.079
.55	-.260	-.279	-.718	-.927	-.866	-.800	.128	.100	.056	.182	.141	.104	.104	.104	14	-.074
.60	-.273	-.291	-.536	-.937	-.897	-.800	.121	.060	.114	.156	.123	.104	.104	.104	15	-.074
.65	-.284	-.300	-.392	-.918	-.919	-.919	.082	-.011	.170	.141	.166	.184	.184	.184	16	-.074
.68	-.309	-.308	-.280	-.927	-.933	-.933	.037	.007	.185	.108	.070	.070	.070	.070	17	-.074
.72	-.314	-.307	-.248	-.941	-.941	-.941	.051	.132	.155	.075	.080	.080	.080	.080	18	-.074
.75	-.314	-.307	-.248	-.904	-.881	-.881	.051	.132	.155	.075	.080	.080	.080	.080	19	-.074
.77	-.314	-.307	-.248	-.881	-.881	-.881	.051	.132	.155	.075	.080	.080	.080	.080	20	-.074
.80	-.332	-.320	-.261	-.837	-.956	-.604	.118	.156	.104	.096	.076	.072	.072	.072	21	-.025
.82	-.326	-.252	-.837	-.963	-.963	-.963	.129	.087	.080	.079	.061	.055	.055	.055	22	-.025
.83	-.328	-.339	-.327	-.266	-.773	-.773	.129	.087	.080	.079	.061	.055	.055	.055	23	-.025
.85	-.328	-.339	-.327	-.266	-.773	-.773	.129	.087	.080	.079	.061	.055	.055	.055	24	-.025
.87	-.327	-.327	-.254	-.695	-.957	-.545	.074	.074	.065	.065	.051	-.041	-.041	-.041	25	-.025
.88	-.325	-.254	-.695	-.957	-.545	-.545	.074	.074	.065	.065	.051	-.041	-.041	-.041	26	-.025
.90	-.307	-.254	-.695	-.957	-.545	-.545	.074	.074	.065	.065	.051	-.041	-.041	-.041	27	-.025
.91	-.307	-.262	-.695	-.857	-.857	-.857	.029	-.013	.016	.016	.003	-.013	-.013	-.013	28	-.025
.93	-.329	-.262	-.695	-.857	-.857	-.857	.029	-.013	.016	.016	.003	-.013	-.013	-.013	29	-.025
.95	-.286	-.268	-.695	-.857	-.857	-.857	.029	-.013	.016	.016	.003	-.013	-.013	-.013	30	-.025
.96	-.237	-.258	-.695	-.857	-.857	-.857	.029	-.013	.016	.016	.003	-.013	-.013	-.013	31	-.025
.97	-.237	-.258	-.695	-.857	-.857	-.857	.029	-.013	.016	.016	.003	-.013	-.013	-.013	32	-.025
.99	-.245	-.245	-.245	-.245	-.245	-.245	.029	-.049	.016	.016	.003	-.013	-.013	-.013	33	-.025

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TABLE III - Continued

TABLE III.- Continued

TABLE III. - Continued

TABLE III.- Continued

TABLE III. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

$\delta = 0.0 \quad M = 1.05 \quad \alpha = 5.0 \quad p_{t,j}/p_{\infty} = 7.0$						$\delta = 0.0 \quad M = 1.05 \quad \alpha = 10.0 \quad p_{t,j}/p_{\infty} = 1.0$						Nacelle			
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			
.01	-0.051	-0.610					.215	.328					1	-0.088	
.03	-0.098	-0.582	-0.556	-0.531			.199	.307	.146	.165	.216		2	-0.129	
.05	-0.125	-0.405		-0.531	-0.566	-0.610	.191	.265		.100	.174	.152	3	-0.310	
.08	-0.119	-0.135	-0.518				.172	.265	.129				4	-0.051	
.10	-0.135	-0.128		-0.546	-0.569		.179			.096			5	-0.034	
.15	-0.107	-0.146	-0.449		-0.590		.198	.257	.055		.090		6	-0.113	
.20	-0.095						.250						7	-0.270	
.25	-0.133	-0.178	-0.381	-0.511	-0.598	-0.613	.215	.160	-0.060	.072	.063	.137	8	-0.029	
.30	-0.107	-0.209	-0.360	-0.459	-0.606	-0.622	.197	-0.533	.616	.054	.026	.102	9	-0.250	
.35	-0.125	-0.255	-0.319	-0.416	-0.597	-0.641	.177	.088	.046	.025	-0.005	.054	10	-0.224	
.40	-0.184	-0.265	-0.286	-0.386	-0.561	-0.654	.053	-0.155	.037	.016	-0.049	.016	11	-0.105	
.45	-0.228	-0.235	-0.268	-0.349	-0.522	-0.661	.101	-0.205	.008	-0.037	-0.065	-0.002	12	-0.093	
.50	-0.215	-0.225	-0.259	-0.333	-0.490	-0.670	.092	-0.020	.034	-0.058	-0.074	-0.024	13	-0.070	
.55	-0.197	-0.235	-0.258	-0.302	-0.437		.138	-0.077	.074	-0.068	-0.026				
.60	-0.206	-0.242	-0.222	-0.292	-0.396	-0.674	.008	-0.041	.129	.018	.021	-0.040			
.65	-0.214	-0.242	-0.239	-0.265	-0.361		.009	-0.094	.047	.027	-0.007				
.68												-0.032			
.70	-0.220	-0.242	-0.236	-0.289	-0.346	-0.670	.044	-0.178	.048	-0.030	-0.070				
.73												-0.081			
.75	-0.234	-0.238	-0.229				.324		.110	.045	-0.058				-0.094
.77							.271								
.79							.266								
.80	-0.242	-0.238					.303	-0.588	.002	-0.036					-0.085
.82							.250								
.83							.243	-0.263							
.85	-0.228	-0.255					.283		.045	-0.104	.019	.047	.009	-0.001	
.87							.248	-0.242							
.88							.247								
.90							.223	-0.274	.517						
.91							.236	-0.241							
.93							.235	-0.223							
.95	-0.225	-0.225					.257		.024	-0.051	-0.057				-0.031
.96							.229								
.97							.212								
.98							.193								
												.107			
$\delta = 0.0 \quad M = 1.05 \quad \alpha = 10.0 \quad p_{t,j}/p_{\infty} = 1.0$						$\delta = 0.0 \quad M = 1.05 \quad \alpha = 10.0 \quad p_{t,j}/p_{\infty} = 1.0$						Nacelle			
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			
.01	-0.216	-0.766					.310	.479					1	-0.015	
.03	-0.283	-0.791	-0.712	-0.702			.274	.471	.280	.327	.287		2	-0.065	
.05	-0.308	-0.886		-0.709	-0.739	-0.737	.424	.429		.303	.282	.268	3	-0.272	
.08	-0.257	-0.943	-0.702				.413	.424	.243				4	-0.095	
.10	-0.276	-0.936					.415			.280			5	-0.044	
.15	-0.245	-0.791	-0.745				.385	.409	.210		.215		6	-0.043	
.20							.368						7	-0.221	
.25	-0.215	-0.257	-0.864	-0.770	-0.746	-0.731	.346	.310	.220	.207	.178	.260	8	-0.123	
.30	-0.260	-0.305	-0.793	-0.763	-0.737	-0.737	.357	.154	.219	.176	.130	.253	9	-0.220	
.35	-0.254	-0.312	-0.871	-0.822	-0.777	-0.749	.290	.065	.181	.139	.105	.211	10	-0.192	
.40	-0.307	-0.341	-0.848	-0.852	-0.791	-0.758	.196	.008	.153	.093	.094	.184	11	-0.011	
.45	-0.347	-0.316	-0.841	-0.875	-0.809	-0.762	.078	.041	.119	.061	.131	.182	12	-0.002	
.50	-0.311	-0.306	-0.811	-0.894	-0.834	-0.763	.066	.135	.064	.031	.144	.167	13	-0.005	
.55	-0.293	-0.302	-0.734	-0.850	-0.850		.117	.103	.013	.078	.161				
.60	-0.320	-0.309	-0.594	-0.896	-0.865	-0.770	.127	.063	.071	.151	.156	.156			
.65	-0.294	-0.512	-0.456	-0.982	-0.878		.082	-0.006	.050	.166	.147				
.68												.133			
.70	-0.311	-0.312	-0.341	-0.879	-0.882	-0.645	.034	-0.115	.168	.152		.126			
.73												.133			
.75	-0.311	-0.306	-0.282				.048	-0.001	.191			.136			
.77												.132			
.79												.142			
.80	-0.324	-0.317					.084	-0.592	.016	.152		.133		-0.087	
.82												.178			
.83												.174			
.85	-0.322	-0.332					.169	.171		.137		.117			
.87												.163			
.88												.158			
.90												.145			
.91	-0.308	-0.256					.084	-0.533		.135		.096		-0.024	
.93	-0.311	-0.260					.104	.073		.106		.053			
.95	-0.284	-0.293										.022			
.96															
.97															
.98															

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TABLE III. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

X C	5 = 0.0 M = 1.05						$\alpha = 10.0$	$P_{t_1}/P_{\infty} = 3.0$	Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90			0.19	0.32	0.53	0.65	0.74	0.90			
.01	-.237	-.767							.525	.475						1	.015
.03	-.427	-.791	-.714	-.705					.579	.448	.277	.320	.284			2	-.070
.05	-.305	-.682			-.714	-.747	-.742		.433	.429		.302	.281	.269		3	-.258
.06	-.257	-.940			-.708				.613	.420	.241					4	.056
.10	-.427	-.933							.613	.420						5	.044
.15	-.427	-.753	-.751		-.726	-.741			.283	.407	.210		.212			6	-.042
.20	-.222								.368							7	-.219
.25	-.217	-.254	-.872	-.776	-.756	-.737			.339	.311	.217	.201	.188	.259		8	.026
.30	-.281	-.300	-.913	-.795	-.771	-.744			.557	.452	.214	.169	.142	.251		9	-.212
.35	-.255	-.308	-.880	-.597	-.784	-.754			.290	.059	.179	.133	.166	.208		10	-.176
.40	-.206	-.342	-.855	-.858	-.798	-.743			.197	.011	.151	.090	.161	.179		11	.083
.45	-.136	-.314	-.847	-.841	-.819	-.766			.08C	.063	.113	.068	.172	.176		12	.086
.50	-.312	-.307	-.818	-.966	-.842	-.4770			.088	.124	.061	.111	.157	.159		13	.089
.55	-.295	-.304	-.742	-.902	-.857				.117	.102	.011	.188	.166				
.60	-.302	-.312	-.603	-.937	-.874	-.778			.123	.060	.020	.175	.151	.149			
.65	-.296	-.312	-.461	-.892	-.887				.084	-.CC4	.186	.148	.134				
.68																117	
.70	-.314	-.312	-.346	-.889	-.892	-.655			.036	-.107	.211	.136				120	
.73																117	
.75	-.314	-.308	-.286			-.902			.026	.147	.184					122	
.77																136	
.79																113	
.80	-.329	-.319							.139	.188						116	.080
.82																134	
.83																129	
.85	-.324	-.337														112	
.87																131	
.88																101	
.90																105	
.91																085	.017
.93																070	
.95	-.298	-.298														085	
.96																044	
.97																048	
.98																053	
																005	
X C	5 = 0.0 M = 1.05						$\alpha = 10.0$	$P_{t_1}/P_{\infty} = 5.1$	Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90			0.19	0.32	0.53	0.65	0.74	0.90			
.01	-.214	-.769							.520	.474						1	.016
.03	-.281	-.791	-.714	-.709					.380	.466	.279	.326	.282			2	-.067
.05	-.308	-.887			-.714	-.744	-.749		.430	.429		.303	.279	.268		3	-.099
.08	-.256	-.943			-.705				.416	.418	.246					4	.124
.10	-.276	-.941							.615							5	.045
.15	-.244	-.779	-.748		-.752				.385	.403	.213					6	.045
.20	-.220								.371							7	-.196
.25	-.214	-.254	-.870	-.777	-.752	-.743			.341	.314	.218	.207	.205	.257		8	.126
.30	-.261	-.299	-.914	-.796	-.769	-.723			.358	.256	.216	.177	.185	.249		9	-.210
.35	-.255	-.311	-.877	-.828	-.778	-.761			.291	.054	.186	.143	.126	.206		10	-.073
.40	-.308	-.340	-.853	-.859	-.795	-.768			.095	.013	.154	.104	.169	.178		11	.164
.45	-.347	-.317	-.847	-.885	-.811	-.774			.020	.062	.116	.112	.175	.173		12	.169
.50	-.311	-.309	-.816	-.905	-.836	-.776			.095	.121	.080	.159	.157	.157		13	.161
.55	-.295	-.302	-.739	-.900	-.857				.117	.102	.023	.201	.146				
.60	-.296	-.311	-.598	-.964	-.872	-.785			.124	.053	.109	.178	.149	.146			
.65	-.294	-.311	-.456	-.899	-.885				.036	-.005	.189	.164	.131				
.68																111	
.70	-.308	-.315	-.337	-.885	-.892	-.658			.041	-.062	.205	.130				114	
.73																115	
.75	-.312	-.308	-.281			-.898			.026	.165	.173					121	
.77																107	
.79																119	
.80	-.326	-.319														112	.073
.82																120	
.83																137	
.85	-.322	-.334														120	
.87																137	
.88																120	
.90																137	
.91																114	
.93																102	
.95	-.288	-.292														069	.007
.96																078	
.97																048	
.98																020	
																020	
																045	

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TABLE III. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

X	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Office number	C _p
.01	-0.192	-0.759					0.330	0.480					1	0.025
.03	-0.260	-0.780	-0.700	-0.700			0.382	0.471	0.284	0.324	0.292		2	-0.045
.05	-0.269	-0.875					0.429	0.436	0.303	0.287	0.272		3	-0.011
.08	-0.249	-0.929	-0.686				0.422	0.424	0.245				4	-0.196
.10	-0.265	-0.925		-0.713	-0.725		0.419		0.277				5	0.055
.15	-0.230	-0.736	-0.731		-0.738		0.387	0.411	0.214		0.229		6	-0.034
.20	-0.206						0.414						7	-0.112
.25	-0.234	-0.246	-0.860	-0.756	-0.725	-0.727	0.366	0.316	0.219	0.207	0.223	0.256	8	-0.193
.30	-0.212	-0.289	-0.898	-0.775	-0.751	-0.737	0.326	0.150	0.217	0.177	0.197	0.245	9	-0.129
.35	-0.224	-0.317	-0.861	-0.806	-0.763	-0.747	0.308	0.062	0.183	0.149	0.192	0.200	10	-0.062
.40	-0.259	-0.343	-0.836	-0.839	-0.775	-0.753	0.190	0.023	0.158	0.120	0.172	0.168	11	-0.248
.45	-0.339	-0.311	-0.824	-0.859	-0.791	-0.758	0.079	0.067	0.124	0.140	0.167	0.161	12	-0.274
.50	-0.296	-0.301	-0.794	-0.877	-0.817	-0.761	0.098	0.133	0.077	0.163	0.156	0.145	13	-0.217
.55	-0.283	-0.294	-0.717	-0.871	-0.871		0.114	0.102	0.057	0.193	0.166			
.60	-0.295	-0.305	-0.573	-0.875	-0.849	-0.770	0.128	0.063	0.156	0.163	0.154	0.129		
.65	-0.287	-0.305	-0.441	-0.858	-0.859		0.086	0.003	0.183	0.154	0.134			
.68											0.111			
.70	-0.295	-0.305	-0.335	-0.855	-0.859	-0.643	0.047	0.020	0.154	0.134		0.093		
.73												0.100		
.75	-0.303	-0.302	-0.282		-0.863		0.072	0.172	0.176			0.096		
.77												0.097		
.79												0.094		
.80	-0.317	-0.313					0.143	0.122				0.077	0.055	
.82												0.137		
.83												0.109		
.85	-0.309	-0.327					0.167	0.166				0.063		
.87	-0.318	-0.253	-0.791		-0.866							0.051		
.88	-0.317	-0.259	-0.756		-0.866							0.051		
.90	-0.250		-0.703		-0.866	-0.530						0.051		
.91	-0.303											0.051		
.93	-0.304	-0.255										0.051		
.95	-0.278	-0.282										0.051		
.96												0.051		
.97												0.051		
.98	-0.234	-0.251										0.051		

TABLE III. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

x c	5 = 7.5 M = 0.80						$\alpha = 0.0$	$P_{t,j}/P_{\infty} = 1.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	.074	-.047					.063	.014		
.02	.034	-.055	-.143	-.096	-.106	-.169	.056	-.001	-.083	-.090
.03	.020	-.074					.020	.301	-.095	-.090
.04	.002	-.064	-.103				.007	.001	-.105	
.10	-.020									
.15	-.032	-.089	-.110							
.20	-.044									
.25	-.086	-.078	-.110	-.103	-.099	-.154	.016	-.125	-.083	-.066
.30	-.051	-.098	-.108	-.106	-.127	-.169	.016	-.176	-.074	-.073
.35	-.086	-.130	-.120	-.016	-.147	-.174	.097	-.195	-.103	-.076
.40	-.118	-.137	-.127	-.154	-.155	-.183	.113	-.192	-.111	-.077
.45	-.135	-.123	-.138	-.151	-.156	-.182	.131	-.162	-.124	-.088
.50	-.110	-.133	-.146	-.169	-.178	-.179	.173	-.102	-.138	-.047
.55	-.105	-.139	-.159	-.165	-.182		.135	-.121	.119	.004
.60	-.116	-.140	-.141	-.188	-.188	-.145	.115	-.131	.062	.010
.65	-.097	-.148	-.168	-.182	-.219		.097	-.150	.000	.049
.68										.016
.70	-.125	-.155	-.187	-.260	-.206	-.112	.091	-.079	.073	.006
.73	-.147	-.161	-.236		-.164		.040	.013	.152	.007
.77										.008
.79										
.80	-.184	-.213					.030	.124		
.82										
.83										
.85	-.267	-.446					.133	.241		
.87	-.324	-.156	-.106				.238	.146	.042	.003
.88	-.298						.203			
.89										
.90	-.291						.212			
.91	-.182						.179			
.92	-.204						.129	.146	.036	.006
.93	-.125	-.045					.090	.101	.081	
.94	-.138									
.95	-.102	-.081					.052	.068		
.96	-.102									
.97	-.078	-.037								
.98	-.078									
x c	5 = 7.5 M = 0.80						$\alpha = 0.0$	$P_{t,j}/P_{\infty} = 3.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	.092	-.026					.101	.032		
.02	.055	-.058	-.126	-.072	-.081	-.159	.058	.036	-.066	-.070
.03	.038	-.025					.028	.019	-.057	-.057
.04	.021	-.044	-.080				.028	.018	-.086	
.10	.001	-.053					.011			
.15	-.013	-.070	-.092	-.079	-.081	-.080	.018	.003	-.117	-.072
.20	-.021									
.25	-.362	-.057	-.093	-.279	-.074	-.138	.006	-.102	-.064	-.054
.30	-.031	-.078	-.092	-.384	-.099	-.149	.004	-.153	-.054	-.041
.35	-.056	-.107	-.102	-.312	-.120	-.164	.075	-.167	-.080	-.057
.40	-.099	-.114	-.113	-.123	-.131	-.169	.089	-.163	-.086	-.066
.45	-.084	-.104	-.122	-.128	-.128	-.167	.107	-.138	-.092	-.049
.50	-.088	-.114	-.129	-.144	-.152	-.164	.148	-.089	-.102	-.005
.55	-.084	-.117	-.144	-.140	-.157		.109	-.095	.077	.042
.60	-.056	-.119	-.123	-.161	-.163	-.195	.085	-.103	.019	.047
.65	-.076	-.127	-.146	-.157	-.195		.067	-.111	.054	.035
.68										.021
.72	-.108	-.133	-.171	-.234	-.182	-.094	.055	-.035	.113	.129
.73	-.128	-.141	-.215		-.138		.002	.072	.175	.018
.75										.021
.77										
.79										
.80	-.163	-.190					.068	.162		
.82										
.83										
.85	-.245	-.421					.158	.256		
.87	-.470	-.137	-.088				.213	.056		
.88	-.281						.209			
.89										
.90	-.274	-.076					.222			
.91	-.165						.184			
.92	-.187						.138			
.93	-.109	-.039					.107	.091		
.94	-.124						.091			
.95	-.066						.073			
.96	-.092						.056			
.97							.076			
.98	-.069	-.025					.050			

TABLE III. - Continued

x c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
	.01	.082	-0.036				.091	.026						
.03	.045	-0.049	-0.139	-0.090			.016	.010	-0.072	-0.080	-0.064			
.05	.079	-0.046		-0.090	-0.100	-0.177	.029	.011	-0.076	-0.089	-0.069	.001		
.06	.069	-0.057	-0.098				.019	.013	-0.095					
.10	-0.011	-0.068		-0.094	-0.095		.008		-0.079					
.15	-0.021	-0.052	-0.102		-0.093		.011	-0.005	-0.128		-0.082			
.20		-0.052					.008							
.25		-0.072	-0.068	-0.102	-0.093	-0.188	-0.146		-0.112	-0.072	-0.063	-0.049	-0.012	
.30	-0.041	-0.090	-0.101	-0.097	-0.114	-0.161	-0.23	-0.162	-0.065	-0.066	-0.053	-0.037		
.35	-0.076	-0.118	-0.111	-0.123	-0.186	-0.175	-0.09	-0.174	-0.092	-0.062	-0.053	-0.042	10	-0.109
.40	-0.106	-0.120	-0.125	-0.142	-0.139	-0.182	-0.094	-0.173	-0.083	-0.072	-0.052	-0.061	11	-0.053
.45	-0.094	-0.114	-0.132	-0.142	-0.139	-0.178	-0.111	-0.148	-0.100	-0.056	-0.045	-0.062	12	-0.065
.50	-0.090	-0.124	-0.140	-0.156	-0.166	-0.175	-0.153	-0.192	-0.108	-0.031	-0.031	-0.060	13	-0.065
.55	-0.094	-0.130	-0.153	-0.154	-0.167		-0.112	-0.092	-0.080	-0.034	-0.001			
.60	-0.104	-0.131	-0.134	-0.175	-0.174	-0.141	-0.090	-0.107	-0.020	-0.034	.015	-0.043		
.65	-0.085	-0.130	-0.158	-0.170	-0.205		-0.069	-0.110	-0.039	-0.072	.026			
.70		-0.115	-0.144	-0.182	-0.245	-0.194	-0.103	-0.057	-0.034	.097	.014			
.73														
.75	-0.135	-0.152	-0.230			-0.149		-0.006	.059	.163				
.77						-0.271								
.79														
.80	-0.173	-0.203				-0.102	-0.051	-0.060	.144					
.82														
.83														
.85	-0.255	-0.435	-0.306	-0.163		-0.067		-0.147	.245	.203	.078	.009		
.87														
.88														
.89														
.90	-0.289		-0.090		-0.034	-0.021		-0.174		.108	.043	.011	-0.006	
.91			-0.175		-0.060									
.92	-0.201							-0.126		.144				
.93			-0.122		-0.043					.095	.083			
.94	-0.138								-0.081					
.95	-0.103		-0.077		-0.020	-0.004			-0.044		.067			
.96											.038	.011		
.97														
.98	-0.079		-0.036							.048	.069			

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	Cp
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.094	-.028					.097	.034						
.03	.058	-.039	-.123	-.075			.026	.021	-.063	-.072	-.062			
.05	.037	-.052		-.075	-.087		.038	.022		-.068	-.063	.006		
.08	.022	-.045	-.087				.046	.021	-.083					
.10	-.001	-.052		-.081	-.085		.012			-.071				
.15	-.009	-.066	-.089		-.084		.017	.008	-.116		-.076			
.20	-.022						.012							
.25	-.065	-.057	-.091	-.078	-.077	-.138	.006	-.101	-.060	-.052	-.040	-.009		
.30	-.030	-.077	-.000	-.064	-.104	-.151	.016	-.148	-.050	-.056	-.047	-.033	9	-.652
.35	-.047	-.064	-.100	-.109	-.127	-.166	.070	-.062	-.072	-.056	-.047	-.034	10	-.077
.40	-.097	-.114	-.111	-.120	-.130	-.174	.068	-.082	-.065	-.055	-.040	-.037	11	-.075
.45	-.084	-.103	-.120	-.130	-.133	-.172	.105	-.135	-.075	-.065	-.047	-.036	12	-.091
.50	-.088	-.113	-.128	-.148	-.157	-.169	.143	-.078	-.095	-.067	-.024	-.035	13	-.091
.55	-.085	-.117	-.141	-.142	-.150	-.166	.106	-.089	-.066	-.042	-.004			
.60	-.095	-.118	-.123	-.126	-.123	-.135	.078	-.093	-.007	-.040	-.022	-.038		
.65	-.076	-.126	-.144	-.159	-.195		.062	-.036	.349	-.075	.030			
.70														
.73	-.109	-.130	-.169	-.235	-.185	-.099	-.046	-.018	.104	.124		.019		
.75	-.126	-.141	-.214		-.143		.002	.070	.171			.019		
.77														
.78														
.80	-.165	-.190					-.066	.153						
.82														
.85														
.85	-.246	-.418	-.292	-.149	-.061		.154	.257	.219	.085	.015			
.87														
.88														
.89														
.90	-.279		-.077	-.052	-.027	-.015	.219							
.91							.180							
.92	-.191		-.197	-.031			.133	.156	.115	.052	.019	.001		
.93														
.94	-.128													
.95														
.96	-.096		-.065		-.012	.004	.091	.107	.091					
.97														
.98	-.070		-.026		.010		.054	.075		.045	.017			
									.059	.079				

TABLE III.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

$\frac{x}{c}$	$\delta = 7.5 \quad M = 0.80$						$\alpha = 5.0 \quad P_{t,j}/P_{\infty} = 1.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90			
.01	-0.107	-0.748					+0.265	+0.304	
.03	-0.145	-0.776	-0.576	-0.587			+0.234	+0.266	
.05	-0.168	-0.793		-0.588	-0.593	-0.623	+0.215	+0.225	
.08	-0.143	-0.243		-0.587			+0.155	+0.214	
.10	-0.185						+0.187		
.15	-0.189	-0.166		-0.618	-0.603		+0.177	+0.140	
.20	-0.172			-0.628	-0.625		+0.162		
.25	-0.207	-0.195	-0.500	-0.576	-0.641	-0.611	+0.153	+0.098	
.30	-0.174	-0.206	-0.427	-0.521	-0.657	-0.617	+0.133	+0.077	
.35	-0.199	-0.227	-0.310	-0.461	-0.654	-0.628	+0.116	+0.077	
.40	-0.231	-0.223	-0.228	-0.425	-0.692	-0.621	+0.086	+0.052	
.45	-0.168	-0.207	-0.192	-0.372	-0.603	-0.608	+0.003	+0.039	
.50	-0.157	-0.211	-0.185	-0.335	-0.579	-0.594	+0.020	+0.006	
.55	-0.177	-0.206	-0.197	-0.293	-0.526		+0.028	+0.016	
.60	-0.177	-0.203	-0.169	-0.281	-0.466	-0.554	+0.016	+0.032	
.65	-0.150	-0.203	-0.192	-0.252	-0.404		+0.001	+0.052	
.68							+0.090	+0.131	
.70	-0.184	-0.203	-0.205	-0.320	-0.387	-0.504	+0.008	+0.001	
.73	-0.184	-0.206	-0.240		-0.254		+0.027	+0.077	
.77							+0.027	+0.077	
.79							+0.212		
.80	-0.213	-0.248					+0.167		
.82							+0.165		
.85	-0.270	-0.475					+0.305		
.87	-0.547	-0.145	-0.095				+0.256		
.88	-0.303						+0.111		
.89							+0.040		
.90	-0.275		-0.091		-0.058	-0.388	+0.255		
.91	-0.189		-0.058				+0.223		
.92	-0.194						+0.146		
.93	-0.122		-0.038				+0.080		
.94	-0.132						+0.030		
.95	-0.076						+0.030		
.96	-0.089								
.97									
.98	-0.056	-0.031					+0.097	+0.051	
							+0.064		
$\frac{x}{c}$	$\delta = 7.5 \quad M = 0.80$						$\alpha = 5.0 \quad P_{t,j}/P_{\infty} = 3.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90			
.01	-0.114	-0.754					+0.260	+0.304	
.03	-0.152	-0.785	-0.580	-0.649			+0.231	+0.268	
.05	-0.114	-0.782		-0.586	-0.598	-0.639	+0.214	+0.226	
.08	-0.152	-0.266	-0.592				+0.192	+0.214	
.10	-0.191	-0.164		-0.620	-0.607		+0.184	+0.136	
.15	-0.197	-0.210	-0.638		-0.632		+0.174	+0.187	
.20	-0.180						+0.163		
.25	-0.214	-0.197	-0.513	-0.595	-0.650	-0.625	+0.150	+0.099	
.30	-0.179	-0.209	-0.441	-0.538	-0.666	-0.634	+0.127	+0.011	
.35	-0.207	-0.228	-0.316	-0.483	-0.666	-0.641	+0.115	+0.035	
.40	-0.240	-0.225	-0.235	-0.442	-0.649	-0.631	+0.082	+0.050	
.45	-0.194	-0.209	-0.195	-0.382	-0.623	-0.618	+0.007	+0.036	
.50	-0.193	-0.213	-0.190	-0.348	-0.595	-0.604	+0.018	+0.007	
.55	-0.184	-0.211	-0.200	-0.299	-0.550		+0.027	+0.010	
.60	-0.183	-0.205	-0.172	-0.268	-0.490	-0.564	+0.011	+0.021	
.65	-0.158	-0.256	-0.197	-0.256	-0.423		+0.004	+0.036	
.68							+0.114	+0.135	
.70	-0.190	-0.205	-0.210	-0.323	-0.393	-0.516	+0.005	+0.025	
.73	-0.190	-0.209	-0.246		-0.268		+0.046	+0.115	
.77							+0.211		
.79							+0.165		
.80	-0.221	-0.251			-0.184	-0.457	+0.108	+0.195	
.82							+0.295		
.83							+0.241		
.85	-0.279	-0.477					+0.106		
.87	-0.493	-0.150	-0.104		-0.115		+0.183		
.88	-0.311						+0.231		
.89									
.90	-0.292		-0.093		-0.064	-0.061	-0.407	+0.246	
.91	-0.193						+0.209		
.92	-0.208						+0.170		
.93	-0.129	-0.349					+0.121		
.94	-0.148						+0.106		
.95	-0.082						+0.084		
.96	-0.104						+0.069		
.97	-0.071	-0.037					+0.052		

TABLE III. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

$\delta = 7.5$						$M = 0.80$	$\alpha = 5.0$	$P_{t,j}/P_{\infty} = 5.0$						
X/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	-0.115	-0.760	-0.583	-0.443	-0.598	-0.643	.260	.302						
.03	-0.153	-0.799	-0.583	-0.443	-0.598	-0.643	.229	.243	.196	.225	.229	.233		
.05	-0.116	-0.748	-0.527	-0.469	-0.588	-0.643	.210	.222						
.08	-0.158	-0.246	-0.592	-0.421	-0.621	-0.610	.191	.212	.132	.180	.193	.233		
.10	-0.195	-0.172	-0.634	-0.534	-0.634	-0.634	.163	.173		.145				
.15	-0.197	-0.212	-0.634	-0.534	-0.634	-0.634	.160	.183	.073	.123				
.20	-0.181													
.25	-0.215	-0.200	-0.506	-0.588	-0.651	-0.630	.147	.097	.077	.092	.119	.167		
.30	-0.180	-0.212	-0.432	-0.527	-0.669	-0.638	.125	.095	.073	.077	.094	.145	9	-0.045
.35	-0.209	-0.232	-0.312	-0.473	-0.645	-0.645	.113	.039	.046	.070	.085	.104	10	-0.023
.40	-0.241	-0.225	-0.232	-0.432	-0.664	-0.637	.059	.050	.035	.055	.070	.086	11	-0.133
.45	-0.195	-0.213	-0.197	-0.377	-0.617	-0.623	.005	.036	.022	.062	.049	.058	12	-0.109
.50	-0.197	-0.216	-0.193	-0.140	-0.587	-0.609	.018	.004	.008	.072	.041	.100	13	-0.100
.55	-0.185	-0.214	-0.203	-0.294	-0.537	-0.570	.028	.012	.023	.122	.091			
.60	-0.187	-0.210	-0.174	-0.285	-0.478	-0.570	.010	.021	.071	.111	.092	.019		
.65	-0.159	-0.210	-0.199	-0.258	-0.414	-0.405	.005	.032	.111	.133	.086			
.68											.068			
.70	-0.192	-0.208	-0.208	-0.326	-0.342	-0.519	.007	.033	.156	.168		.010		
.73											.057			
.75	-0.195	-0.214	-0.244		-0.259		.050	.112	.209		.164	.054		
.77					.345						.159			
.79					.259									
.80	-0.224	-0.255				-0.181	-0.460	.104	.188		.294	.039	-0.047	
.82						.937					.248	.104		
.83						.349	-0.164							
.85	-0.281	-0.482				-0.112		.197	.278					
.87	-0.477		-0.152	-0.100				.266	.182					
.88			.316					.232						
.89								.242						
.90	-0.295		.093		.067	-0.058	-0.411	.206	.133		.054	.016	-0.114	
.91	-0.195							.168						
.92	-0.212							.153						
.93	-0.134		.054					.114	.109					
.94	-0.151							.100						
.95	-0.085							.085						
.96	-0.103							.062			.046			
.97								.050			.090			
.98	-0.075		.041											
$\delta = 7.5$						$M = 0.80$	$\alpha = 10.0$	$P_{t,j}/P_{\infty} = 1.0$						
X/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	-0.428	-1.100					.383	.386						
.03	-0.489	-1.150	-0.937	-0.903	-0.859	-0.733	.398	.401	.282	.291	.252			
.05	-0.359	-1.242					.395	.366	.289	.275	.238			
.08	-0.403	-1.479	-0.945				.373	.361	.246					
.10	-0.452	-1.522		-0.914		-0.861	.363		.264					
.15	-0.418	-0.881	-0.943		-0.870		.337	.341	.217		.244			
.20	-0.376						.315							
.25	-0.398	-0.266	-0.934	-0.937	-0.867	-0.645	.296	.265	.209	.206	.227	.239		
.30	-0.359	-0.300	-1.003	-0.938	-0.891	-0.653	.282	.157	.197	.186	.200	.241	9	-0.029
.35	-0.361	-0.326	-1.054	-0.990	-0.905	-0.609	.253	.091	.164	.173	.193	.181	10	.009
.40	-0.378	-0.344	-1.054	-1.030	-0.909	-0.582	.186	.060	.145	.157	.174	.145	11	.081
.45	-0.308	-0.323	-0.993	-1.083	-0.921	-0.560	.108	.069	.127	.153	.164	.136	12	.096
.50	-0.309	-0.319	-0.868	-1.116	-0.944	-0.555	.123	.097	.098	.178	.163	.114	13	.083
.55	-0.296	-0.308	-0.725	-1.104	-0.962		.094	.074	.103	.199	.178			
.60	-0.284	-0.296	-0.514	-1.113	-0.980	-0.489	.088	.054	.134	.185	.167	.078		
.65	-0.252	-0.253	-0.370	-0.073	-1.047		.081	.022	.169	.203	.155			
.68											.138			
.70	-0.280	-0.269	-0.247	-1.116	-1.055	-0.453	.069	.068	.214	.229		.024		
.73											.124			
.75	-0.258	-0.258	-0.223		-0.957		.100	.136	.274					
.77						.645					.224			
.79						.347					.220			
.80	-0.274	-0.283				-0.916	-0.413	.144	.227		.361		.105	-0.040
.82											.308	.174		
.83											.244	.160		
.85	-0.298	-0.433				-0.876		.240	.319				.076	
.87											.308			
.88											.270			
.89											.290			
.90	-0.284		-0.057			-0.806	-0.393	.255	.210	.199		.034	-0.151	
.91			-0.207			.038					.134			
.92	-0.213			-0.007				.200		.155	.169			
.93	-0.140													
.94	-0.156													
.95	-0.090													
.96	-0.108													
.97														
.98	-0.071		-0.038			.033								

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TABLE III. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

S = 7.5 M = 0.80							$\alpha = 10^\circ 0$ $P_{t,j}/P_{\infty} = 3.1$							Nacelle	
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			
.01	-0.423	-1.087					+384	+390							
.03	-0.481	-1.137	-0.920	.035			+399	+407	.288	.302	.258				
.05	-0.352	-1.227		-0.897	-0.851	-0.733	+401	+373		+299	+283	.243			
.08	-0.398	-1.465	-0.933				+376	+365	.256						
.10	-0.448	-1.510		-0.905	-0.854		+366			.277					
.15	-0.405	-1.870	-0.940		-0.862		+339	+347	.224		.249				
.20	-0.367						+319								
.25	-0.386	-0.251	-0.928	-0.931	-0.859	-0.641	+301	+275	.218	.220	.239	.245			
.30	-0.351	-0.289	-1.001	-0.931	-0.887	-0.622	+285	+164	.204	.200	.212	.251	9	.003	
.35	-0.354	-0.317	-1.050	-0.978	-0.861	-0.605	+255	+160	.172	.188	.203	.188	10	.038	
.40	-0.310	-0.353	-1.050	-1.025	-0.965	-0.583	+203	+072	.155	.171	.181	.157	11	.059	
.45	-0.302	-0.312	-0.987	-1.083	-0.915	-0.557	+116	+078	.142	.172	.175	.143	12	.089	
.50	-0.305	-0.312	-0.972	-1.112	-0.943	-0.533	+130	+110	.117	.192	.169	.119	13	.102	
.55	-0.292	-0.300	-0.919	-1.099	-0.962		+104	+087	.125	.215	.185				
.60	-0.276	-0.268	-0.566	-1.112	-0.978	-0.487	+099	+073	.159	.197	.174	.082			
.65	-0.247	-0.278	-0.363	-1.070	-1.054		+092	+053	.196	.211	.160				
.68												.145			
.70	-0.279	-0.263	-0.243	-1.110	-1.087	-0.446	+094	+100	.234	.236	.130	.031			
.75	-0.250	-0.253	-0.218		-0.967		+123	+182	.274		.231				
.77											.222				
.79															
.80	-0.266	-0.273			-0.904	-0.408	+170	+245			.106	-0.034			
.82															
.83															
.85	-0.295	-0.476					+258	+321		+302	.181	.005			
.87	-0.481	-0.124	.010		-0.874		+306	+241	.158						
.88	-0.329						+266								
.89															
.90	-0.279						+295								
.91	-0.202	-0.055	.039		-0.810	-0.386	+258								
.92	-0.209						+204	+205	.196	.134	.037	-0.145			
.93															
.94	-0.181	-0.136	-0.005				+146	+155	.163						
.95	-0.188														
.96	-0.104						+100	+116		.110	.397				
.97							+077	+140							
.98	-0.068	-0.037													
S = 7.5 M = 0.80							$\alpha = 10^\circ 0$ $P_{t,j}/P_{\infty} = 4.8$							Nacelle	
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			
.01	-0.425	-1.090					+383	+392							
.03	-0.484	-1.142	-0.922	-0.898			+398	+406	.288	+298	+254				
.05	-0.356	-1.233		-0.907	-0.856	-0.734	+400	+374							
.08	-0.399	-1.470	-0.939				+374	+365	.259						
.10	-0.453	-1.515		-0.936	-0.859		+364			.275					
.15	-0.414	-0.871	-0.945		-0.868		+339	+345	.226		.248				
.20	-0.374						+321								
.25	-0.393	-0.253	-0.933	-0.936	-0.866	-0.645	+299	+275	.218	.219	.238	.240			
.30	-0.355	-0.294	-1.006	-0.936	-0.893	-0.624	+287	+168	.207	.198	.209	.245	9	.016	
.35	-0.362	-0.316	-1.066	-0.984	-0.905	-0.611	+256	+107	.178	.185	.201	.184	10	.053	
.40	-0.372	-0.334	-1.056	-1.029	-0.911	-0.583	+200	+079	.155	.169	.180	.150	11	.158	
.45	-0.307	-0.316	-0.992	-1.086	-0.922	-0.555	+113	+079	.144	.169	.173	.139	12	.142	
.50	-0.308	-0.313	-0.872	-1.121	-0.946	-0.536	+130	+112	.121	.188	.167	.114	13	.143	
.55	-0.297	-0.302	-0.725	-1.107	-0.967		+102	+092	.133	.211	.183				
.60	-0.282	-0.292	-0.510	-1.120	-0.982	-0.494	+100	+078	.161	.192	.170	.078			
.65	-0.249	-0.279	-0.364	-1.076	-1.061		+097	+056	.193	.204	.157				
.68												.139			
.70	-0.280	-0.266	-0.242	-1.120	-1.070	-0.451	+095	+110	.230	.224		.025			
.73															
.75	-0.252	-0.253	-0.219		-0.969		+126								
.77															
.79															
.80	-0.272	-0.276			-0.908	-0.419	+170	+245							
.82															
.83															
.85	-0.299	-0.477					+257	+321		+302	.174	.080			
.87	-0.475	-0.128	.004		-0.878		+306	+239	.157						
.88	-0.333						+272								
.89															
.90	-0.281		-0.061		-0.815	-0.392	+296								
.91	-0.205		.036				+255								
.92	-0.211		-0.138		-0.010		+209								
.93							+202								
.94	-0.154						+148								
.95	-0.092						+117								
.96	-0.104						+100								
.97	-0.070		-0.039		-0.036		+140								
.98							.081								

TABLE III. - Continued

TABLE III.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	+.106	-.013					.011	.048						
.03	-.025	-.025	-.139	-.080	-.082	-.083	-.140	.026	-.028	-.081	-.109	-.101		
.05	-.045	-.023					.048	.032			-.099	-.098	-.033	
.08	-.030	-.035	-.096				.035	.032	-.093					
.10	-.009	-.048					.022							
.15	-.001	-.070	-.106		-.037		.029	.021	-.146					
.20	-.019						.026							
.25	-.063	-.066	-.096	-.087	-.081	-.154	.019	-.118	-.079	-.087	-.074	-.026		
.30	-.033	-.091	-.090	-.092	-.129	-.169	-.003	-.222	-.065	-.094	-.078	-.050	9	-.107
.35	-.077	-.118	-.104	-.113	-.130	-.190	-.075	-.190	-.090	-.098	-.072	-.055	10	-.127
.40	-.111	-.123	-.118	-.132	-.134	-.210	-.106	-.189	-.104	-.107	-.068	-.073	11	.069
.45	-.098	-.113	-.129	-.132	-.131	-.228	-.142	-.163	-.125	-.083	-.053	-.075	12	.078
.50	-.098	-.123	-.135	-.144	-.158	-.253	-.182	-.095	-.148	-.023	-.026	-.076	13	.078
.55	-.095	-.127	-.144	-.139	-.153		-.129	-.112	-.122	-.029	.002			
.60	-.105	-.127	-.122	-.156	-.159	-.266	-.108	-.131	-.027	-.037	-.025	-.066		
.65	-.084	-.132	-.138	-.134	-.194		-.095	-.145	-.049	-.079	-.036			
.68														
.70	-.115	-.132	-.144	-.186	-.259	-.212	-.076	-.045	.115	-.133				
.73														
.75	-.124	-.129	-.160		-.315		-.014	.073	.180					
.77														
.79														
.80	-.153	-.152					-.068	.163						
.82														
.83														
.85	-.207	-.304												
.87														
.88														
.89														
.90	-.581		-.268		-.161		-.060	-.020						
.91														
.92	-.472		-.276		-.090									
.93														
.94	-.286													
.95														
.96	-.162													
.97														
.98	-.100	-.041												
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-.055	-.678					.279	.316						
.03	-.100	-.004	-.963	-.578	-.599	-.688	.250	.277	-.172	-.209	-.221			
.05	-.058	-.640					.229	.238		.186	.181	.232		
.08	-.100	-.156	-.569				.207	.224		.121				
.10	-.154	-.159		-.611	-.611		.204			.133				
.15	-.159	-.210		-.600	-.635		.195	.202		.056		.112		
.20	-.143						.181							
.25	-.200	-.187	-.477	-.553	-.644	-.715	.169	.096	-.065	-.373	-.107	-.166		
.30	-.172	-.212	-.422	-.489	-.652	-.733	.159	-.009	.069	-.056	-.084	-.050	9	-.100
.35	-.208	-.238	-.327	-.438	-.646	-.756	.131	-.053	-.037	-.042	-.078	-.114	10	-.078
.40	-.243	-.233	-.253	-.402	-.619	-.758	.059	-.073	.016	-.027	-.056	-.074	11	.065
.45	-.197	-.215	-.220	-.359	-.590	-.744	.014	-.053	-.005	-.057	-.070	-.065	12	.070
.50	-.197	-.223	-.216	-.332	-.562	-.729	.034	-.006	-.036	-.072	-.080	-.052	13	.068
.55	-.192	-.225	-.224	-.293	-.515		.036	-.021	-.027	.113	.103			
.60	-.194	-.225	-.191	-.285	-.477	-.674	.024	-.047	-.022	.111	.109	.028		
.65	-.171	-.227	-.211	-.258	-.458		.011	-.080	-.086	.144	.106			
.68														
.70	-.216	-.217	-.209	-.307	-.448	-.607	-.026	-.026	-.155	-.186		-.001		
.73														
.75	-.200	-.212	-.204		-.403		-.019	.080	-.229					
.77														
.79														
.80	-.218	-.224		-.743	-.476	-.318	-.519	-.069	-.191					
.82														
.83														
.85	-.254	-.390		-.734	-.391	-.195								
.87														
.88														
.89														
.90	-.557		-.257		-.101	-.451								
.91														
.92	-.389		-.444		-.098									
.93														
.94	-.242		-.227		-.073									
.95														
.96	-.147		-.125		-.027									
.97														
.98	-.086	-.045		.017	-.033									

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TABLE III - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

$\delta = 7.5$						$M = 0.90$	$\alpha = 5.0$	$P_{t,j}/P_{\infty} = 3.0$			Nacelle		
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan					Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74		
.01	-0.054	-0.687					.270	.311					
.03	-0.106	-0.712	-0.569	.051			.240	.272	.169	.207	.220		
.05	-0.064	-0.639					.222	.233		.168	.181	.229	
.08	-0.109	-0.165	-0.571				.200	.220	.123				
.10	-0.160	-0.167		-0.619	-0.619		.197			.129			
.15	-0.162	-0.217	-0.601		-0.642		.186	.197	.054		.113		
.20	-0.150						.172						
.25	-0.210	-0.194	-0.476	-0.552	-0.655	-0.731	.162	.093	.066	.075	.113	.166	
.30	-0.178	-0.220	-0.421	-0.486	-0.667	-0.754	.153	-.014	.067	.057	.087	.147	
.35	-0.216	-0.244	-0.392	-0.442	-0.657	-0.776	.120	-.059	.040	.046	.082	.109	10
.40	-0.249	-0.240	-0.281	-0.404	-0.631	-0.778	.056	-.072	.020	.035	.071	.059	11
.45	-0.205	-0.220	-0.230	-0.362	-0.596	-0.763	.021	-.055	.001	.046	.075	.062	12
.50	-0.207	-0.230	-0.222	-0.335	-0.565	-0.749	.036	-.001	.024	.080	.045	.045	13
.55	-0.199	-0.233	-0.231	-0.301	-0.524	-0.709	.039	-.021	.010	.121	.123		
.60	-0.202	-0.231	-0.198	-0.295	-0.481	-0.692	.024	-.042	.050	.115	.105	.025	
.65	-0.182	-0.233	-0.231	-0.266	-0.463	-0.611	.067		.118	.144	.101		
.68											.080		
.70	-0.222	-0.226	-0.217	-0.317	-0.455	-0.621	-.014	.006	.174	.183		-.008	
.75											.065		
.76	-0.210	-0.220	-0.212		-0.418		.039	.124	.229			.057	
.77											.178		
.79											.171		
.80	-0.227	-0.232					.117	.211				.038	-.043
.82													
.83													
.85	-0.262	-0.398					.218	.292				.024	
.87	-0.657	-0.734	-0.238				.268	.284	.191	.051			
.88	-0.678						.248						
.89													
.90	-0.572	-0.298		-0.120	-0.107	-0.464	.233						
.91	-0.481						.182						
.92	-0.411						.178						
.93	-0.252	-0.085					.118						
.94	-0.259												
.95	-0.141												
.96	-0.159												
.97	-0.097	-0.069											
.98													
$\delta = 7.5$						$M = 0.90$	$\alpha = 5.0$	$P_{t,j}/P_{\infty} = 5.1$			Nacelle		
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan					Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74		
.01	-0.055	-0.680					.277	.317					
.03	-0.099	-0.705	-0.560	.056			.248	.279	.177	.216	.226		
.05	-0.059	-0.649		-0.578	-0.598	-0.592	.220	.241		.176	.189	.233	
.08	-0.103	-0.164	-0.565				.210	.229	.132				
.10	-0.155	-0.157			-0.608	-0.609	.203			.142			
.15	-0.160	-0.210	-0.597				.194	.204	.062				
.20	-0.147						.178						
.25	-0.201	-0.183	-0.479	-0.562	-0.645	-0.722	.170	.104	.072	.084	.119	.171	
.30	-0.172	-0.211	-0.422	-0.502	-0.659	-0.743	.142	-.003	.073	.070	.066	.152	9
.35	-0.206	-0.236	-0.385	-0.456	-0.651	-0.745	.129	-.048	.045	.062	.090	.113	10
.40	-0.193	-0.215	-0.359	-0.425	-0.642	-0.746	.060	-.041	.029	.048	.076	.073	11
.45	-0.197	-0.213	-0.221	-0.377	-0.601	-0.751	.013	-.042	.013	.057	.080	.067	12
.50	-0.199	-0.223	-0.215	-0.334	-0.574	-0.737	.029	-.011	.011	.087	.084	.050	13
.55	-0.191	-0.226	-0.223	-0.295	-0.536	-0.737	.031	-.009	.010	.127	.106		
.60	-0.195	-0.224	-0.189	-0.286	-0.497	-0.678	.016	-.029	.065	.122	.110	.029	
.65	-0.173	-0.225	-0.211	-0.256	-0.475	-0.675	.002	-.045	.122	.147	.104		
.68											.084		
.70	-0.219	-0.218	-0.209	-0.306	-0.460	-0.610	.001	.030	.174	.187		-.004	
.75	-0.198	-0.211	-0.204		-0.416		.056	.131	.229			.069	
.77												.061	
.79													
.80	-0.221	-0.226					.126	.211				.044	-.040
.82													
.83													
.85	-0.257	-0.387		-0.732	-0.406	-0.205	.222	.299	.271	.111		.028	
.87	-0.671	-0.726	-0.229				.293	.198	.058				
.88	-0.669						.257						
.89													
.90	-0.567	-0.292			-0.110	-0.461	.273						
.91	-0.475						.238						
.92	-0.404						.180						
.93	-0.247	-0.078					.125						
.94	-0.255												
.95	-0.136												
.96	-0.153												
.97													
.98	-0.091	-0.061											

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TABLE III.- Continued

TABLE III.-- Continued

TABLE III. - Continued

TABLE III-- Continued

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TABLE III. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

X C	$\delta = 7.5$ $M = 1.00$						$\alpha = 5.0$ $P_{t,j}/P_{\infty} = 3.0$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-011	-565					+265	+370						
.03	-045	-572	-507	-503			+259	+331	+194	+225	+227			
.05	-002	-444		-503	-534	-611	+265	+297		+185	+182	+257		
.08	-032	-588	-502				+254	+283	+154					
.10	-077	-076		-520	-542		+247							
.15	-066	-118	-490	-520	-568		+250	+264	+073	+151	+098			
.20	-049						+242							
.25	-108	-166	-340	-474	-569	-643	+232	+148	+089	+089	+094	+197		
.30	-074	-177	-287	-401	-574	-666	+224	+014	+098	+062	+085	+175	9	-+255
.35	-147	-188	-530	-359	-558	-696	+178	-031	+070	+033	+095	+143	10	-+122
.40	-195	-162	-213	-315	-525	-716	+095	-057	+044	+001	+098	+101	11	+115
.45	-162	-162	-210	-288	-489	-726	+002	-028	+010	+011	+136	+094	12	+112
.50	-112	-173	-216	-273	-460	-741	+015	+036	-041	+097	+118	+077	13	+118
.55	-139	-180	-224	-254	-417		+004	+012	+084	+150	+143			
.60	-150	-184	-194	-252	-382	-739	+002	-027	+058	+156	+150	+056		
.65	-134	-199	-217	-230	-363		+004	-086	+158	+188	+149			
.70	-189	-205	-214	-261	-366	-723	+039	+066	+217	+232		+109		
.73							+019	+163	+276			+106		
.75	-192	-205	-200				+019					+095		
.77												+225		
.79												+219		
.80	-214	-211					+156	+257				+050	-+024	
.82												+370		
.83												+316		
.85	-227	-303	-578	-427			+262	+334				+137	+025	
.87							+326	+229	+059					
.88							+289							
.89							+315							
.90	-531	-534	-345	-345	-523		+281							
.91	-542		-333				+221	+161	+016	+006	+092			
.92	-485	-533	-498				+222							
.93	-452						+155	+154	+107					
.94	-452						+095							
.95	-501						+098							
.96	-418						+020	+051	+022					
.97	-431	-445					+054							
.98	-431	-413												
X C	$\delta = 7.5$ $M = 1.00$						$\alpha = 5.0$ $P_{t,j}/P_{\infty} = 5.1$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-003	-557					+274	+376						
.03	-037	-555	-502	-060			+267	+335	+193	+225	+229			
.05	-006	-553		-497	-531	-610	+270	+301	+186	+186	+256			
.08	-031	-082	-502				+261	+287	+156					
.10	-070	-067		-519	-541		+261							
.15	-061	-114	-483		-565		+254	+268	+076	+149	+102			
.20	-046						+247							
.25	-102	-161	-337	-457	-566	-640	+235	+153	+091	+090	+107	+199		
.30	-070	-173	-280	-393	-571	-665	+226	+017	+100	+265	+097	+177	9	-+139
.35	-144	-182	-232	-350	-552	-694	+183	-027	+071	+037	+104	+147	10	-+024
.40	-194	-179	-213	-308	-516	-714	+090	-054	+047	+011	+101	+103	11	+157
.45	-161	-158	-210	-284	-482	-727	+004	-024	+014	+039	+107	+095	12	+163
.50	-139	-169	-214	-272	-452	-738	+012	-045	-035	+110	+118	+077	13	+145
.55	-137	-174	-220	-249	-406		+000	+017	+066	+156	+142			
.60	-148	-180	-193	-250	-376	-740	+004	-023	+085	+158	+150	+058		
.65	-131	-192	-215	-229	-356		+009	-076	+160	+187	+148			
.70	-187	-199	-213	-258	-359	-727	+027	+006	+214	+229		+022		
.73							+049	+170	+273			+105		
.75	-189	-199	-197				+049					+095		
.77														
.79														
.80	-213	-206					+161	+252				+060	-+020	
.82														
.83														
.85	-225	-297					+262	+338				+026		
.87							+331	+227	+062					
.88							+299							
.89							+312							
.90	-528	-577		-343	-528		+283							
.91	-535		-330				+162							
.92	-482	-528	-493				+221	+225	+021					
.93							+152	+152	+120					
.94	-450	-493					+152	+105						
.95	-417						+095	+077	+011					
.96	-427	-400		-344			+054							

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TABLE III - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

 $\delta = 7.5$ $M = 1.00$ $\alpha = 5.0$ $P_{t,j}/P_{\infty} = 6.0$

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	-.003	-.555					.272	.373						
.03	-.039	-.566	-.504	-.497			.266	.332	.192	.226	.228			
.05	-.005	-.431		-.499	-.531	-.614	.269	.297		.187	.183	.255		
.08	-.036	-.077	-.500				.259	.284	.154					
.10	-.070	-.073			-.523	-.543	.260				.150			
.15	-.052	-.115	-.480		-.568		.252	.265	.073					
.20	-.046						.244							
.25	-.103	-.163	-.331	-.459	-.568	-.645	.233	.153	.087	.088	.114	.196		
.30	-.071	-.176		-.275	-.381	-.572	.227	.013	.059	.065	.059	.173	9	-.052
.35	-.145	-.185	-.228	-.345	-.553	-.699	.181	-.033	.071	.028	.163	.142	10	.014
.40	-.193	-.183	-.232	-.299	-.515	-.717	.099	-.058	.045	.037	.039	.100	11	.187
.45	-.163	-.163	-.212	-.270	-.480	-.732	.002	-.030	.012	.050	.105	.071	12	.207
.50	-.142	-.172	-.216	-.261	-.446	-.744	.015	.011	.023	.131	.123	.075	13	.170
.55	-.159	-.179	-.223	-.247	-.359		.004	.015	.048	.150	.147			
.60	-.148	-.183	-.197	-.247	-.365	-.745	.001	-.024	.053	.149	.147	.053		
.65	-.133	-.197	-.216	-.228	-.344		.006	-.073	.150	.184	.147			
.68												.125		
.70	-.187	-.203	-.212	-.260	-.349	-.732	-.023	.030	.201	.228		.015		
.73												.103		
.75	-.191	-.203	-.198				.661	.162	.267			.092		
.77												.223		
.79												.217		
.80	-.214	-.203					.156	.238				.057	-.029	
.82														
.83														
.85	-.225	-.302					.263	.334						
.87								.313	.138					
.88								.236	.056					
.89								.256						
.90	-.530						.311							
.91	-.540						.279							
.92	-.483						.220							
.93	-.529						.159							
.94	-.451						.150							
.95	-.500						.085							
.96	-.420													
.97	-.434													
.98	-.414													

 $\delta = 7.5$ $M = 1.00$ $\alpha = 10.0$ $P_{t,j}/P_{\infty} = 1.0$

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	-.143	-.772					.436	.497						
.03	-.221	-.787	-.693	.027			.471	.494	.322	.349	.297			
.05	-.104	-.871		-.697	-.713	-.747	.488	.455		.327	.299	.278		
.08	-.187	-.936	-.694				.469	.446	.283					
.10	-.261	-.985		-.707	-.717		.421							
.15	-.205	-.727	-.745		-.729		.426	.421	.246					
.20	-.166						.409							
.25	-.213	-.251	-.881	-.752	-.725	-.751	.385	.332	.241	.228	.240	.263		
.30	-.190	-.248	-.882	-.769	-.752	-.767	.363	.194	.232	.202	.217	.258	9	-.025
.35	-.164	-.280	-.887	-.813	-.764	-.782	.319	.119	.197	.181	.212	.210	10	.010
.40	-.155	-.222	-.867	-.852	-.773	-.795	.244	.086	.168	.164	.198	.176	11	.133
.45	-.275	-.262	-.849	-.879	-.792	-.803	.137	.059	.140	.170	.157	.168	12	.143
.50	-.258	-.243	-.797	-.804	-.819	-.810	.161	.148	.096	.202	.198	.151	13	.135
.55	-.246	-.269	-.685	-.891	-.849		.135	.115	.089	.233	.216			
.60	-.243	-.279	-.509	-.895	-.874	-.841	.122	.077	.144	.228	.208	.124		
.65	-.238	-.290	-.385	-.875	-.920		.103	.022	.202	.252	.194			
.68														
.70	-.302	-.298	-.282	-.881	-.934	-.849	.070	.060	.266	.283		.085		
.73														
.75	-.288	-.297	-.253		-.946		.106	.171	.326					
.77														
.79														
.80	-.315	-.304			-.961	-.734	.189	.284						
.82														
.83														
.85	-.324	-.363			-.565	-.524								
.87														
.88														
.89														
.90	-.579	-.556			-.951	-.618	.355							
.91	-.608						.323							
.92	-.528						.265							
.93	-.591	-.579					.261							
.94	-.506						.189							
.95	-.557						.132							
.96	-.478						.129							
.97	-.461	-.437			-.586			.050						

TABLE III. - Continued

		$\delta = 7.5$			$M = 1.00$			$\alpha = 10.0$		$p_{t,j}/p_{\infty} = 3/1$					
X/c	Y/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	Cp
		0.10	0.32	0.53	0.65	0.74	0.90	0.10	0.32	0.53	0.65	0.74	0.90		
.01	-.148	-.770						.4435	.493						
.03	-.228	-.797	-.702	-.699				.448	.489	.320	.347	.305			
.05	-.109	-.882		-.702	-.721	-.753		.4486	.450		.325	.307	.275		
.08	-.192	-.946	-.697					.4488	.438	.281					
.10	-.264	-.994						.4456			.299				
.13	-.210	-.736	-.785		-.735			.4423	.415	.247		.263			
.20	-.173							.4402							
.25	-.218	-.257	-.883	-.756	-.740	-.753		.3862	.325	.241	.235	.255	.264		
.30	-.196	-.255		-.927	-.769	-.759	-.769	.361	.193	.234	.212	.233	.229	9	.009
.35	-.280	-.289		-.890	-.814	-.769	-.785	.315	.116	.200	.195	.225	.221	10	.043
.40	-.320	-.301		-.871	-.853	-.778	-.799	.226	.085	.174	.177	.209	.178	11	.145
.45	-.280	-.269		-.853	-.883	-.796	-.805	.136	.078	.147	.185	.106	.170	12	.145
.50	-.283	-.272		-.799	-.902	-.824	-.815	.165	.142	.118	.215	.206	.150	13	.145
.55	-.251	-.276		-.684	-.895	-.857	-	.156	.115	.123	.247	.221			
.60	-.247	-.284		-.510	-.900	-.883	-.844	.05	.083	.179	.235	.209	.125		
.65	-.242	-.296		-.380	-.880	-.929		.109	.043	.236	.253	.193			
.68													.173		
.73	-.306	-.303	-.278	-.887	-.945	-.855		.092	.101	.282	.279		.085		
.75	-.294	-.301	-.250		-.955			.014C	.221	.322					
.77					-1.002						.267				
.79					-	-883					.253				
.80	-.319	-.309		-.600		-.970	-.728	.214	.297						
.82				-	-571	-.524	-.976								
.83				-	-			.314	.367						
.85	-.330	-.367		-.640	-.559	-.364	-.976		.359	.270	.387				
.87				-	-	-			.321						
.88				-	-652										
.89				-	-	-		.352							
.90	-.577		-.555			-.965	-.622	.317		.203					
.91		-.614		-	-258				.243		.052				
.92	-.527								.251						
.93		-.595	-.576						.180	.176	.151				
.94	-.503									.121	.116				
.95	-.479	-.562				-.905					.089	.022			
.96					-	-263									
.97					-	-584									
.98	-.462	-.470							.035						

TABLE III - Continued

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TABLE III.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

S = 7.5 M = 1.05						$\alpha = 0.0$	$P_{t,j}/P_{\infty} = 5.1$	Nacelle				
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan					Wing lower surface stations, fraction of semispan				Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.01	.096	.032	-135	-086			.096	.076				
.03	.066	.023	-135	-086	-093	-102	.057	.065	-087	-070	-166	
.05	.041	.026					.039	.075		-096	-171	-161
.08	.027	.014	-093				.027	.079	-137			
.10	.010	.005					.011			-135		
.15	.023	-019	-078				.036	.091	-049		-132	
.20	.016						.052					
.25	.009	-038	-097	-097	-134	-145	.068	.028	-177	-119	-104	-159
.30	.001	-063	-081	-091	-117	-155	.089	-185	-232	-091	-124	-175
.35	-.031	-089	-096	-107	-137	-167	.076	-179	-117	-104	-152	-214
.40	-.046	-112	-111	-129	-141	-175	.085	-212	-108	-134	-181	-221
.45	-.094	-105	-122	-132	-143	-186	.225	-268	-120	-135	-186	-232
.50	-.089	-107	-132	-145	-163	-202	.206	-118	-145	-136	-197	-252
.55	-.087	-112	-147	-145	-171		.238	-109	-166	-148	-205	
.60	-.091	-118	-124	-144	-167	-216	.163	-114	-194	-210	-225	-254
.65	-.076	-122	-140	-138	-176		.056	-155	-301	-242	-166	
.68												-053
.70	-114	-128	-140	-157	-190	-238	-116	-212	-337	.065		-077
.73												.021
.75	-120	-120	-127		-203		-169	-263	.130			.033
.77												
.79												
.80	-137	-126			-232	-235	-167	.079		.282		
.82										.258	.117	
.83										.240		
.85	-144	-167	-473	-382			.106	.240				-002
.87												
.88												
.89												
.90	-419	-463	-283	-266	-232		.221					
.91	-471						.209					
.92	-396						.195					
.93	-458	-421					.176					
.94	-374						.124					
.95	-428						.072					
.96	-354						.071					
.97												-034
.98	-382	-388										.014
												.309
S = 7.5 M = 1.05						$\alpha = 0.0$	$P_{t,j}/P_{\infty} = 6.7$	Nacelle				
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan					Wing lower surface stations, fraction of semispan				Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.01	.097	.032					.094	.068				
.03	.045	.023	-136	-091	-097	-103	.055	.064	-093	-076	-166	
.05	.040	.023					.035	.072		-099	-173	-161
.08	.020	.009	-096				.025	.076	-148			
.10	.008	-002		-113	-102		.008		-137			
.15	.027	-024	-082		-105		.038	.089	-056		-130	
.20	.015						.053					
.25	.007	-042	-102	-103	-109	-146	.064	.025	-182	-121	-104	-161
.30	-.004	-067	-084	-098	-124	-154	.086	-185	-238	-096	-125	-174
.35	-.032	-090	-100	-115	-142	-168	.072	-180	-125	-107	-154	-215
.40	-.046	-116	-117	-134	-145	-176	.094	-214	-114	-137	-182	-223
.45	-.097	-107	-129	-135	-148	-188	.229	-270	-124	-155	-186	-235
.50	-.092	-108	-137	-150	-164	-203	.207	-118	-156	-137	-197	-254
.55	-.087	-115	-150	-146	-175		.238	-112	-174	-151	-207	
.60	-.094	-121	-129	-146	-172	-219	.164	.115	-198	-211	-228	-241
.65	-.079	-125	-145	-142	-181		.091	.159	-295	-203	-092	
.68												.020
.70	-115	-130	-145	-158	-190	-238	-119	-215	.020	.046		-055
.73												.017
.75	-122	-122	-134		-205		-173	-056	.045			.030
.77												
.79												
.80	-143	-127			-236	-236	-120	.010		.269		
.82										.243	.099	
.83										.236		
.85	-148	-171			-259		.091	.236				-014
.87	-.491	-488	-327					.256	.176	.027		
.88	-.499							.236				
.89												
.90	-423	-472	-289	-268	-234		.194		.123		-002	-040
.91	-473											-082
.92	-402											
.93	-463	-425										
.94	-381											
.95	-356	-430			-271							
.97					-295							
.98	-380	-388	-368									

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TABLE III - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

						$\delta = 7.5$	$M = 1.05$	$\alpha = 4.9$	$P_{t,j}/P_{\infty} = 1.0$					
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	-0.042	-0.638					.218	.324						
.03	-0.087	-0.577	-0.541	.012			.200	.286	.143	.147	.197			
.05	-0.037	-0.288					.191	.259		.093	.156	.144		
.08	-0.085	-0.104	-0.503				.177	.256	.126					
.10	-0.147	-0.133					.179							
.15	-0.121	-0.141	-0.450				.199	.254	.055					
.20	-0.079						.206							
.25	-0.124	-0.171	-0.364	-0.472	-0.590	-0.620	.214	.168	-0.064	.070	.057	.080		
.30	-0.095	-0.196	-0.345	-0.402	-0.584	-0.637	.223	.048	.049	.054	.019	.115	9	-0.260
.35	-0.153	-0.233	-0.307	-0.359	-0.547	-0.652	.188	.066	.047	.021	.011	.117	10	-0.256
.40	-0.198	-0.254	-0.283	-0.327	-0.494	-0.659	.038	.142	.029	.010	.053	.086	11	-0.032
.45	-0.220	-0.224	-0.262	-0.305	-0.439	-0.657	.075	.195	.005	.036	.074	.083	12	-0.028
.50	-0.214	-0.218	-0.255	-0.294	-0.398	-0.674	.090	.018	.040	.037	.101	.075	13	-0.027
.55	-0.188	-0.223	-0.255	-0.280	-0.355		.155	.001	.078	.067	.114			
.60	-0.188	-0.225	-0.219	-0.281	-0.329	-0.678	.092	.036	.129	.144	.084	.069		
.65	-0.173	-0.233	-0.236	-0.264	-0.319		.002	.082	.239	.058	.147			
.68												.135		
.70	-0.224	-0.238	-0.235	-0.286	-0.331	-0.674	.038	.167	.050	.204		.041		
.73												.122		
.75	-0.221	-0.230	-0.226				.104	.124	.204			.115		
.77												.222		
.79												.243		
.80	-0.235	-0.233					.139	.121						
.82												.360		
.83												.324		
.85	-0.232	-0.293					.185	.280						
.87	-0.513	-0.550	-0.348	-0.357				.297	.262	.094				
.88	-0.541											.281		
.89												.285		
.90	-0.490							.268						
.91	-0.517	-0.547	-0.321	-0.346	-0.528			.231						
.92	-0.466													
.93	-0.513	-0.514												
.94	-0.438													
.95	-0.492													
.96	-0.405													
.97	-0.431	-0.451	-0.331					.133						
.98	-0.431	-0.421						.103						
								.067						
						$\delta = 7.5$	$M = 1.05$	$\alpha = 5.0$	$P_{t,j}/P_{\infty} = 3.1$					
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	-0.044	-0.632					.217	.326						
.03	-0.090	-0.580	-0.539	-0.531			.200	.289	.143	.146	.200			
.05	-0.044	-0.305	-0.533	-0.565	-0.615		.192	.261		.092	.156	.147		
.08	-0.094	-0.101	-0.515				.175	.257	.126					
.10	-0.148	-0.129	-0.552	-0.574			.183							
.12	-0.119	-0.140	-0.458	-0.602			.200	.258	.055	.071		.076		
.20	-0.060							.206						
.25	-0.125	-0.169	-0.373	-0.490	-0.596	-0.617	.216	.166	-0.063	.069	.056	.185		
.30	-0.104	-0.197	-0.349	-0.420	-0.592	-0.632	.227	.046	.056	.051	.020	.161	9	-0.256
.35	-0.152	-0.231	-0.310	-0.385	-0.560	-0.649	.186	.067	.044	.022	.014	.137	10	-0.258
.40	-0.167	-0.252	-0.282	-0.343	-0.508	-0.661	.072	.135	.033	.015	.055	.098	11	-0.100
.45	-0.218	-0.221	-0.261	-0.315	-0.459	-0.667	.073	.183	.010	.040	.074	.093	12	-0.118
.50	-0.211	-0.213	-0.254	-0.304	-0.419	-0.675	.085	.009	.039	.045	.084	.083	13	-0.109
.55	-0.186	-0.221	-0.254	-0.287	-0.374		.135	.002	.079	.070	.107			
.60	-0.187	-0.224	-0.219	-0.285	-0.346	-0.675	.067	.032	.138	.057	.156	.072		
.65	-0.172	-0.229	-0.238	-0.267	-0.331		.001	.078	.208	.177	.164			
.68												.145		
.70	-0.222	-0.234	-0.236	-0.289	-0.339	-0.670	.035	.166	.191	.244		.042		
.73												.124		
.75	-0.218	-0.227	-0.225	-0.348			.103	.027	.277			.116		
.77												.244		
.79												.245		
.80	-0.233	-0.228					.027	.229						
.82												.382		
.83												.334		
.85	-0.231	-0.288					.049	.318				.166		
.87												.326		
.88												.301		
.89												.312		
.90	-0.492	-0.552					.287							
.91	-0.515							.243						
.92	-0.469							.238						
.93	-0.510	-0.514						.184						
.94	-0.440	-0.489						.178						
.95	-0.408							.128						
.97	-0.436	-0.448						.127						
.98	-0.434	-0.434						.087						
								.061						

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TABLE III.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

X C	$\delta = 7.5$ $M = 1.05$						$\alpha = 5.0$	$P_{t_1}/P_{\infty} = 5.1$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.038	-0.523	-0.541	-0.528	-0.506	-0.608	+225	+327		
.03	-0.054	-0.574	-0.541	-0.528	-0.529	-0.563	+206	+290	+148	+207
.05	-0.039	-0.509					+198	+263		
.08	-0.095	-0.101	-0.506				+182	+257	+133	
.10	-0.139	-0.130					+190			
.15	-0.113	-0.121	-0.448				+205	+256	+060	+078
.20	-0.078						+213			
.25	-0.126	-0.172	-0.368	-0.480	-0.596	-0.609	+221	+165	-0.359	+071
.30	-0.093	-0.198	-0.345	-0.419	-0.591	-0.614	+232	-0.465	+060	+052
.35	-0.150	-0.233	-0.305	-0.375	-0.562	-0.641	+189	-0.667	+047	+022
.40	-0.194	-0.252	-0.276	-0.339	-0.510	-0.651	+079	-1.131	+035	-0.012
.45	-0.218	-0.223	-0.254	-0.313	-0.461	-0.659	+071	-1.185	+013	-0.038
.50	-0.209	-0.217	-0.248	-0.302	-0.423	-0.665	+082	-0.600	-0.012	-0.070
.55	-0.183	-0.222	-0.251	-0.285	-0.374		+127	-0.504	-0.077	-0.049
.60	-0.184	-0.226	-0.217	-0.285	-0.346	-0.665	+088	-0.332	-0.134	+012
.65	-0.170	-0.232	-0.235	-0.266	-0.330		+002	-0.077	+108	+182
.68										+150
.70	-0.221	-0.233	-0.231	-0.288	-0.338	-0.661	+034	-0.169	+194	+244
.75	-0.217	-0.227	-0.223		-0.347		+101	+069	+277	
.77										+245
.79										+245
.80	-0.229	-0.227					+070	+228		
.82										+387
.83										+339
.85	-0.230	-0.290					+250	+324		
.87	-0.540	-0.549	-0.349					+332	+258	+089
.88	-0.537							+303		
.89										+055
.90	-0.489		-0.547		-0.399	-0.503	+317			
.91	-0.516		-0.318				+292	+199	+031	-0.046
.92	-0.465		-0.512	-0.509			+244	+248	+049	
.93							+188	+193	+160	
.94	-0.435						+138	+147	+019	+031
.95	-0.405							+105		
.97	-0.429	-0.437	-0.444				+074			
.98										
X C	$\delta = 7.5$ $M = 1.05$						$\alpha = 10.0$	$P_{t_1}/P_{\infty} = 1.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.182	-0.754					+336	+478		
.03	-0.253	-0.767	-0.685	-0.686	-0.708	-0.726	+287	+467	+283	+326
.05	-0.195	-0.689					+431	-0.26	+303	+286
.08	-0.219	-0.891	-0.677				+118	+420	+241	
.10	-0.276	-0.911					+420		+276	
.15	-0.234	-0.716	-0.696	-0.708	-0.716		+388	+406	+205	+229
.20	-0.195						+377			
.25	-0.220	-0.254	-0.853	-0.742	-0.720	-0.721	+363	+306	+215	+206
.30	-0.196	-0.295	-0.869	-0.769	-0.735	-0.732	+347	+150	+217	+178
.35	-0.260	-0.312	-0.836	-0.798	-0.745	-0.751	+302	+060	+187	+156
.40	-0.303	-0.346	-0.810	-0.818	-0.758	-0.762	+204	+015	+159	+182
.45	-0.324	-0.314	-0.797	-0.832	-0.777	-0.767	+083	+057	+125	+155
.50	-0.304	-0.304	-0.764	-0.846	-0.800	-0.776	+115	+133	+078	+228
.55	-0.279	-0.300	-0.686	-0.839	-0.815		+115	+105	+043	+219
.60	-0.271	-0.303	-0.533	-0.842	-0.825	-0.795	+106	+063	+161	+235
.65	-0.257	-0.305	-0.413	-0.821	-0.852		+095	+008	+234	+287
.68										+224
.70	-0.301	-0.311	-0.316	-0.818	-0.874	-0.807	+051	-0.015	+301	+325
.73	-0.293	-0.305	-0.277		-0.873		+086	+204	+368	
.75										+200
.77										+189
.79										
.80	-0.314	-0.309					+219	+315		
.82										+152
.83										+107
.85	-0.319	-0.351								
.87	-0.376	-0.576	-0.516	-0.393			+344	+405		
.88	-0.602							+420	+227	
.89								+336	+145	
.90	-0.536		-0.509		-0.872	-0.585	+395			
.91	-0.575			-0.264			+366	+309	+273	+110
.92	-0.495						+312		+247	+219
.93	-0.566	-0.527					+246	+186		
.94	-0.482						+187	+187	+089	+105
.95	-0.536							+160		
.96	-0.451							+103		
.97	-0.477	-0.447								

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TABLE III.- Concluded
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 3

0 = 7.5 M = 1.05 $\alpha = 10.0 P_{t,j}/P_{\infty} = 3.0$							
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan					Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	
.01	-0.193	-0.762	-0.690	-0.688	-0.707	-0.731	
.03	-0.262	-0.775	-0.694	-0.690	-0.707	-0.731	
.05	-0.162	-0.854	-0.875	-0.770	-0.734	-0.739	
.08	-0.214	-0.896	-0.678	-0.745	-0.757	-0.707	
.10	-0.283	-0.917	-0.699	-0.706	-0.718	-0.718	
.15	-0.237	-0.726	-0.718	-0.718	-0.707	-0.704	
.20	-0.198					-0.736	
.25	-0.232	-0.253	-0.856	-0.744	-0.719	-0.726	
.30	-0.197	-0.293	-0.875	-0.770	-0.734	-0.739	
.35	-0.262	-0.310	-0.837	-0.792	-0.745	-0.757	
.40	-0.305	-0.341	-0.812	-0.820	-0.759	-0.767	
.45	-0.325	-0.309	-0.802	-0.835	-0.778	-0.775	
.50	-0.303	-0.302	-0.767	-0.848	-0.801	-0.783	
.55	-0.280	-0.296	-0.688	-0.841	-0.817	-0.774	
.60	-0.270	-0.300	-0.938	-0.845	-0.830	-0.802	
.65	-0.254	-0.304	-0.414	-0.825	-0.858	-0.700	
.68							
.70	-0.303	-0.311	-0.313	-0.821	-0.875	-0.813	
.73						-0.76	
.75	-0.294	-0.303	-0.266		-0.877		
.77					-0.149	-0.245	
.79					-0.360		
.80	-0.313	-0.309			-0.885	-0.736	
.82					-0.248	-0.329	
.83					-0.400		
.85	-0.320	-0.348	-0.529	-0.522			
.87					-0.347	-0.396	
.88					-0.393	-0.317	
.89					-0.355		
.90	-0.353	-0.506			-0.358	-0.253	
.91	-0.574	-0.263			-0.358	-0.098	
.92	-0.497				-0.286		
.93	-0.561	-0.526			-0.300		
.94	-0.475				-0.223	-0.199	
.95	-0.534				-0.232	-0.165	
.96	-0.450				-0.174	-0.079	
.97	-0.427	-0.530	-0.260		-0.090	-0.147	
.98	-0.427	-0.452					

0 = 7.5 M = 1.05 $\alpha = 10.0 P_{t,j}/P_{\infty} = 5.0$							
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan					Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74		
.01	-0.193	-0.765					
.03	-0.263	-0.779	-0.697	-0.694	-0.713	-0.733	
.05	-0.165	-0.855	-0.875	-0.794	-0.749	-0.760	
.08	-0.217	-0.901	-0.688				
.10	-0.285	-0.921	-0.703	-0.712			
.15	-0.238	-0.737	-0.725	-0.723			
.20	-0.197						
.25	-0.234	-0.256	-0.863	-0.746	-0.724	-0.733	
.30	-0.201	-0.296	-0.882	-0.771	-0.738	-0.746	
.35	-0.265	-0.315	-0.844	-0.792	-0.749	-0.760	
.40	-0.309	-0.346	-0.821	-0.823	-0.763	-0.771	
.45	-0.330	-0.311	-0.808	-0.841	-0.783	-0.778	
.50	-0.304	-0.304	-0.775	-0.856	-0.809	-0.787	
.55	-0.281	-0.300	-0.690	-0.849	-0.824		
.60	-0.272	-0.303	-0.544	-0.851	-0.835	-0.806	
.65	-0.258	-0.307	-0.418	-0.833	-0.864		
.68							
.70	-0.306	-0.314	-0.320	-0.829	-0.882	-0.816	
.73					-0.086		
.75	-0.296	-0.304	-0.271		-0.159	-0.251	
.77					-0.354		
.79					-0.219		
.80	-0.317	-0.313			-0.324	-0.245	
.82					-0.151	-0.217	
.83					-0.069	-0.190	
.85	-0.320	-0.348	-0.556	-0.535	-0.528		
.87	-0.496	-0.503	-0.522	-0.398			
.88	-0.475	-0.535			-0.347	-0.399	
.89	-0.449	-0.510			-0.397	-0.451	
.90	-0.535	-0.510	-0.268	-0.885	-0.594		
.91	-0.577				-0.361	-0.252	
.92	-0.496	-0.562	-0.528		-0.301	-0.190	
.93	-0.475	-0.535			-0.225	-0.216	
.94	-0.449	-0.535	-0.267	-0.840			
.95	-0.424	-0.452	-0.535		-0.181	-0.180	
.97					-0.167	-0.079	
.98					-0.114		

TABLE IV

TABLE IV-- Continued

TABLE IV.- Continued

TABLE IV.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X c	δ = 0.0 M = 0.80						α = 5°0	P _{t,j} /P _{∞0} = 5±1	Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-112	-758	-776	-694	-704	-845	-526	.255 .317 .282 .203 .200 .212 .162 .176 .214	1	-143
.03	-147	-785	-	-	-	-	-	.193 .253 .127 .118	2	-170
.05	-111	-744	-	-	-	-	-	.193 .253 .127 .118	3	-086
.08	-238	-221	-796	-	-	-	-	.186 .268 .127 .118	4	-020
.10	-317	-263	-	-725	-610	-	-	.175 .165 .077 .096	5	-099
.15	-175	-247	-539	-	-635	-	-	.156 .165 .077 .096	6	-155
.20	-158	-	-	-	-	-	-	.162 .026 .061 .057 .090 .144	7	-120
.25	-221	-182	-180	-530	-618	-505	-	.135 .021 .041 .039 .057 .122	8	-004
.30	-189	-204	-178	-406	-617	-509	-	.105 .021 .041 .039 .057 .122	9	-110
.35	-215	-228	-187	-335	-596	-517	-	.069 .050 .007 .024 .043 .077	10	-107
.40	-238	-225	-196	-268	-541	-513	-	.034 .060 .027 .003 .052 .032	11	-025
.45	-184	-203	-197	-212	-483	-504	-	.034 .062 .026 .010 .007 .014	12	-045
.50	-186	-200	-195	-195	-427	-498	-	.034 .062 .026 .010 .007 .014	13	-052
.55	-174	-193	-187	-164	-345	-	-	.062 .049 .050 .050 .024 .005	-	-
.60	-167	-180	-148	-157	-277	-477	-	.052 .051 .008 .020 .021 .005	-	-
.65	-125	-167	-144	-117	-216	-	-	.046 .109 .028 .017 .017 .002	-	-
.68	-	-	-	-	-	-	-	-	-	-
.70	-149	-147	-127	-128	-161	-446	-	.050 .057 .039 .005 .001	-	-
.73	-	-	-	-	-	-	-	-	-	-
.75	-128	-122	-107	-	-109	-	-	.024 .022 .037 .003	-	-
.77	-	-	-	-	-	-	-	-	-	-
.79	-	-	-	-	-	-	-	-	-	-
.80	-116	-105	-	-	-	-	-	-	-	-
.82	-	-	-	-	-	-	-	-	-	-
.83	-	-	-	-	-	-	-	-	-	-
.85	-0.088	-0.097	-	-	-	-	-	-	-	-
.87	-	-	-	-	-	-	-	-	-	-
.88	-0.071	-0.052	-0.033	-	-	-	-	-	-	-
.90	-	-	-	-	-	-	-	-	-	-
.91	-0.050	-0.007	-	-0.05	-	-355	-	-	-	-
.93	-0.040	-0.013	-	-	-	-	-	-	-	-
.95	-0.049	-0.029	-	-	-	-	-	-	-	-
.96	-	-	-	-	-	-	-	-	-	-
.97	-	-	-	-	-	-	-	-	-	-
.98	-	-	-	-	-	-	-	-	-	-
								.004		
X c	δ = 0.0 M = 0.80						α = 5°0	P _{t,j} /P _{∞0} = 5±6	Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-108	-750	-767	-684	-622	-525	-	.263 .327 .293 .205 .219 .217	1	-119
.03	-142	-779	-	-	-	-	-	.239 .293 .265 .184 .182 .217	2	-130
.05	-105	-740	-	-	-	-	-	.202 .265 .131 .122	3	-037
.08	-230	-224	-797	-	-	-	-	.197 .274 .122 .118	4	-079
.10	-310	-251	-	-717	-593	-	-	.185 .172 .084 .104	5	-080
.15	-167	-235	-538	-	-585	-	-	.164 .172 .084 .104	6	-129
.20	-150	-	-	-	-	-	-	-	7	-081
.25	-215	-171	-170	-539	-548	-506	-	.149 .034 .045 .062 .097 .150	8	-058
.30	-182	-193	-171	-424	-570	-510	-	.105 .046 .044 .065 .120 .90	9	-077
.35	-205	-215	-183	-353	-559	-517	-	.080 .038 .017 .031 .048 .081	10	-065
.40	-230	-213	-194	-277	-516	-512	-	.026 .040 .003 .012 .023 .041	11	-088
.45	-177	-191	-193	-377	-501	-	-	.023 .048 .017 .009 .015 .036	12	-155
.50	-179	-188	-192	-403	-425	-497	-	.024 .050 .040 .016 .032 .020	13	-115
.55	-167	-179	-183	-158	-358	-	-	.051 .054 .032 .050 .028 .000	-	-
.60	-156	-187	-145	-153	-299	-476	-	.049 .042 .010 .016 .025 .000	-	-
.65	-117	-156	-140	-113	-250	-	-	.031 .084 .027 .015 .022 .007	-	-
.70	-139	-135	-122	-126	-193	-443	-	.034 .027 .011 .002 .014	-	-
.73	-	-	-	-	-	-	-	-	-	-
.75	-119	-110	-102	-	-146	-	-	.010 .036 .019 .042 .009	-	-
.77	-	-	-	-	-	-	-	-	-	-
.79	-	-	-	-	-	-	-	-	-	-
.80	-107	-0.095	-	-	-	-	-	.006 .008 .003	11	-037
.82	-	-	-	-	-	-	-	-	-	-
.83	-	-	-	-	-	-	-	-	-	-
.85	-0.080	-0.084	-	-	-	-	-	.008 .010 .009	12	-012
.87	-	-	-	-	-	-	-	-	-	-
.88	-	-	-	-	-	-	-	-	-	-
.90	-	-	-	-	-	-	-	-	-	-
.91	-0.040	-0.027	-	-	-	-	-	.019 .021 .019	14	-083
.93	-0.030	-0.008	-	-	-	-	-	.017 .021 .019	-	-
.95	-0.043	-0.019	-	-	-	-	-	.015 .008 .028	15	-014
.96	-	-	-	-	-	-	-	-	-	-
.97	-	-	-	-	-	-	-	-	-	-
.98	-	-	-	-	-	-	-	.000	-	-

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CONFIDENTIAL

TABLE IV. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

~~CONTINUED~~

TABLE IV.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

 $\delta = 0.0 \quad M = 0.80 \quad \alpha = 10.0 \quad p_{t,j}/p_{\infty} = 5.1$

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-0.437	-1.108	-1.104	-0.953			.385	.408					1	-0.052
.03	-0.495	-1.159	-1.104	-0.953			.413	.427	.325	.266	.239		2	-0.090
.05	-0.474	-1.263		-0.960	-0.830	-0.531	.385	.423	.275	.261	.245		3	-0.013
.08	-0.446	-1.502	-1.218				.379	.434	.280				4	-0.080
.10	-0.432	-1.556		-0.973	-0.838		.386			.249			5	-0.008
.15	-0.329	-0.907	-1.227		-0.847		.372	.345	.224		.220		6	-0.057
.20	-0.385						.326						7	-0.047
.25	-0.367	-0.227	-0.808	-1.080	-0.849	-0.481	.285	.165	.199	.188	.203	.220	8	.087
.30	-0.297	-0.367	-0.684	-1.099	-0.869	-0.472	.245	.113	.176	.166	.170	.208	9	-0.038
.35	-0.291	-0.416	-0.572	-1.104	-0.894	-0.468	.196	.069	.134	.144	.155	.156	10	-0.026
.40	-0.308	-0.398	-0.512	-1.115	-0.909	-0.456	.146	.082	.111	.124	.122	.118	11	-0.098
.45	-0.357	-0.340	-0.459	-1.044	-0.933	-0.439	.086	.054	.090	.114	.113	.107	12	.098
.50	-0.332	-0.319	-0.415	-0.947	-0.951	-0.433	.083	.042	.061	.118	.099	.083	13	.109
.55	-0.299	-0.286	-0.371	-0.798	-0.977		.068	.029	.060	.144	.109			
.60	-0.280	-0.257	-0.307	-0.662	-0.979	-0.404	.054	.012	.092	.107	.094	.050		
.65	-0.235	-0.224	-0.277	-0.480	-0.958		.042	-0.025	.116	.091	.078			
.66												.050		
.70	-0.218	-0.187	-0.239	-0.395	-0.923	-0.378	.033	.021	.121	.074		.006		
.73	-0.178	-0.148	-0.205		-0.861		.043	.037	.108			.045		
.75												.063		
.77												.030		
.80	-0.153	-0.121					.065	.109					12	-0.053
.82														
.83														
.85	-0.112	-0.106	-0.142	-0.152			.059	.067						
.87	-0.081	-0.114	-0.149	-0.195			.056	.061						
.88	-0.071						.056							
.90							.052	.040						
.91	-0.054						.021	.033						
.93	-0.041	-0.059					.018	.014						
.95	-0.059	-0.027					.029	.003						
.96							.032							
.97														
.98	-0.001													

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-0.429	-1.102					.387	.414					1	-0.029
.03	-0.493	-1.157	-1.084	-0.941			.408	.434	.328	.272	.243		2	-0.056
.05	-0.444	-1.243		-0.945	-0.825	-0.517	.385	.426		.279	.265	.250	3	.034
.08	-0.441	-1.489	-1.202				.379	.444	.286				4	.138
.10	-0.426	-1.546		-0.957	-0.834		.364			.252			5	.011
.15	-0.327	-0.884	-1.210		-0.841		.375	.354	.231		.224		6	-0.042
.20	-0.380						.327						7	-0.014
.25	-0.363	-0.217	-0.820	-1.059	-0.845	-0.468	.287	.170	.204	.192	.206	.224	8	.115
.30	-0.293	-0.354	-0.694	-1.077	-0.857	-0.456	.247	.119	.178	.167	.174	.214	9	-0.011
.35	-0.287	-0.405	-0.582	-1.084	-0.880	-0.454	.197	.081	.141	.148	.158	.160	10	.005
.40	-0.302	-0.388	-0.524	-1.097	-0.900	-0.443	.150	.068	.117	.126	.127	.123	11	.149
.45	-0.351	-0.333	-0.464	-1.037	-0.919	-0.428	.090	.059	.097	.116	.117	.112	12	.197
.50	-0.329	-0.306	-0.420	-0.955	-0.936	-0.419	.085	.051	.088	.116	.102	.088	13	.165
.55	-0.262	-0.275	-0.373	-0.813	-0.958		.076	.038	.072	.143	.113			
.60	-0.275	-0.246	-0.312	-0.864	-0.959	-0.390	.065	.023	.040	.104	.097	.057		
.65	-0.231	-0.221	-0.278	-0.501	-0.944		.053	-0.007	.111	.088	.080			
.68														
.70	-0.215	-0.176	-0.236	-0.414	-0.912	-0.365	.041	.037	.100	.070		.009		
.73							.047	.094	.098					
.75	-0.176	-0.136	-0.202		-0.854									
.77														
.79														
.80	-0.147	-0.114			-0.781	-0.340	.067	.078						
.82														
.85	-0.109	-0.095	-0.139	-0.158			.056	.076						
.87	-0.079	-0.109	-0.103		-0.686			.054						
.88	-0.064							.048						
.90		-0.081						.049						
.91	-0.046							.054						
.93	-0.031	-0.054						.042						
.95	-0.056	-0.021			-0.455		.023	.027						
.96		-0.024			-0.017			.024						
.98	-0.002							.019						

~~CONTINUED~~

TABLE IV. -- Continued

TABLE IV. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

		$\delta = C_0$			$M = 0.90$			$\alpha = C_0$		$P_{t,1}/P_{CO} = 5.0$			Nacelle	
$\frac{c}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	0.80		
.01	.114	-.012	-.003	-.113	-.093	-.086	.015	.063	-.051	-.065	-.103	-.117	1	-.243
.03	.076	-.003	-.005	-.080	-.086	-.060	.077	.051	-.061	-.101	-.107	-.113	2	-.244
.05	.052	-.005	-.004	-.076	-.082	-.060	.051	.061	-.070	-.084	-.101	-.113	3	-.148
.08	.044	-.018	-.076	-.079	-.082	-.065	.043	.070	-.084	-.111	-.117	-.136	4	-.241
.10	.017	-.035	-.076	-.079	-.082	-.065	.028	.084	-.111	-.136	-.142	-.156	5	-.202
.15	.034	-.061	-.076	-.079	-.085	-.065	.041	-.051	-.112	-.136	-.142	-.156	6	-.279
.20	.003	-.086	-.076	-.079	-.085	-.065	.035	-.061	-.112	-.136	-.142	-.156	7	-.196
.25	-.052	-.052	-.077	-.087	-.072	-.087	.002	-.099	-.094	-.121	-.119	-.121	8	-.041
.30	-.035	-.080	-.080	-.084	-.100	-.097	.056	-.147	-.114	-.128	-.127	-.096	9	-.232
.35	-.039	-.104	-.094	-.104	-.113	-.108	.104	-.144	-.130	-.142	-.127	-.090	10	-.225
.40	-.116	-.107	-.108	-.124	-.120	-.111	.096	-.153	-.139	-.170	-.129	-.130	11	-.026
.45	-.083	-.099	-.113	-.115	-.105	-.114	.149	-.155	-.161	-.167	-.123	-.115	12	-.032
.50	-.086	-.105	-.118	-.125	-.122	-.120	.140	-.159	-.198	-.122	-.111	-.116	13	-.018
.55	-.083	-.106	-.117	-.128	-.110	-.110	.170	-.156	-.220	-.058	-.087	-.070	14	-.041
.60	-.087	-.101	-.091	-.110	-.058	-.094	.149	-.174	-.113	-.072	-.070	-.096	15	-.041
.65	-.059	-.100	-.092	-.080	-.110	-.110	.131	-.213	-.038	-.064	-.054	-.071	16	-.041
.68	-.091	-.087	-.083	-.092	-.089	-.063	-.137	-.139	-.023	-.063	-.063	-.065	17	-.041
.70	-.080	-.068	-.072	-.072	-.065	-.065	-.090	-.014	-.011	-.011	-.051	-.062	18	-.041
.73	-.077	-.071	-.071	-.072	-.065	-.065	-.090	-.014	-.011	-.011	-.047	-.050	19	-.041
.75	-.079	-.052	-.052	-.060	-.053	-.053	-.091	-.031	-.016	-.031	-.031	-.032	20	-.041
.80	-.076	-.059	-.053	-.040	-.023	-.023	-.076	-.037	-.027	-.030	-.030	-.032	21	-.041
.82	-.073	-.040	-.034	-.016	-.016	-.016	-.007	-.006	-.028	-.005	-.015	-.014	22	-.041
.85	-.057	-.065	-.049	-.019	-.005	-.016	-.007	-.006	-.028	-.001	-.015	-.014	23	-.041
.87	-.035	-.001	-.016	-.006	-.019	-.019	-.018	-.005	-.022	-.001	-.002	-.010	24	-.041
.90	-.017	-.007	-.016	-.040	-.030	-.030	-.055	-.003	-.022	-.001	-.002	-.007	25	-.041
.93	-.003	-.001	-.014	-.038	-.030	-.030	-.007	-.028	-.032	-.007	-.007	-.007	26	-.041
.95	-.013	-.001	-.014	-.038	-.040	-.040	-.007	-.028	-.032	-.007	-.007	-.007	27	-.041
.96	-.001	-.001	-.014	-.038	-.040	-.040	-.007	-.028	-.032	-.007	-.007	-.007	28	-.041
.97	-.001	-.001	-.014	-.038	-.040	-.040	-.007	-.028	-.032	-.007	-.007	-.007	29	-.041
.98	-.001	-.001	-.014	-.038	-.040	-.040	-.007	-.028	-.032	-.007	-.007	-.007	30	-.041

TABLE IV.- Continued

x c	Wing upper surface stations, fraction of semispan					Wing lower surface stations, fraction of semispan					Nacelle	
											Orifice number	Cp
	0.19	0.32	0.53	0.65	0.74	0.19	0.32	0.53	0.65	0.74	0.90	
.01	-.065	-.699				.268	.333				1	-.172
.03	-.107	-.716	-.737	-.684		.248	.299	.202	.199	.204	2	-.228
.05	-.065	-.623		-.690	-.673	.207	.275		.159	.162	3	-.130
.08	-.189	-.148	-.757			.204	.292	.126			4	-.019
.10	-.287	-.244		-.714	-.661	.194			.115		5	-.121
.15	-.141	-.274		-.578	-.563	.174	.173	.074		.084	6	-.250
.20	-.131					.186					7	-.167
.25	-.241	-.170	-.198	-.587	-.655	.155	.022	.058	.047	.079	8	-.039
.30	-.192	-.209	-.199	-.456	-.658	.118	-.031	.039	.027	.048	9	-.157
.35	-.204	-.246	-.216	-.384	-.663	.075	-.058	.005	.008	.037	10	-.154
.40	-.252	-.242	-.229	-.308	-.597	.023	-.056	-.015	-.015	.017	11	-.020
.45	-.204	-.221	-.236	-.252	-.546	.039	-.065	-.041	-.015	.013	12	-.026
.50	-.208	-.230	-.237	-.235	-.493	.037	-.072	-.081	.004	.013	13	-.011
.55	-.204	-.229	-.231	-.204	-.415	.067	-.077	-.097	.052	.030		
.60	-.200	-.218	-.191	-.197	-.333	-.530	-.062	-.099	-.030	-.029	-.029	-.001
.65	-.158	-.212	-.185	-.151	-.256	-.059	-.147	-.041	-.028	-.027		
.68												
.70	-.196	-.187	-.160	-.154	-.186		-.078	-.103	.070	.014		-.017
.73												.006
.75	-.166	-.158	-.131		-.117		-.043	.030	.056			.010
.77					-.109					.050		
.79					-.086					.009		
.80	-.147	-.128			-.057	-.428		.017	.078			
.82					-.091					.022		
.83					-.073	-.058				.020		
.85	-.105	-.108				-.013		.032	.037			
.87					-.091	-.052			.025			
.88					-.078	-.022			.022			
.90						-.023		-.021	-.375			
.91										.002		
.93										-.016		
.95	-.051	-.027				-.005				-.019		-.013
.96										-.010		
.97						-.032				-.010		
.98						-.024				-.030		
						-.007				-.030		

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TABLE IV. - Continued

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TABLE IV - Continued

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TABLE IV. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$$\delta = 0.0 \quad M = 0.90 \quad \alpha = 10.0 \quad p_{t,j}/p_{\infty} = 5.1$$

X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-297	-239					+410	+453					1	-0.52
.03	-280	-281	-1.044	-2.956			+435	+457	+331	+274	+239		2	-0.097
.05	-375	-1.085	-1.044	-2.978	-0.948	-0.724	+404	+443	+275	+253	+232		3	-0.004
.08	-355	-1.223	-1.166				+359	+461	+281				4	-1.01
.10	-354	-1.251		-1.004	-0.963		+386			+243			5	-0.06
.15	-254	-1.007	-1.199		-0.977		+397	+362	+231		+214		6	-0.079
.20	-310						+349						7	-0.050
.25	+341	-1.147	-0.840	-1.094	-0.980	-0.624	+306	+170	+201	+181	+200	+226	8	-0.085
.30	-473	-318	-0.683	-1.03	-0.909	-0.602	+258	+112	+177	+162	+166	+217	9	-0.041
.35	-212	-401	-0.583	-1.038	-0.937	-0.589	+200	+075	+135	+136	+155	+167	10	-0.026
.40	-249	-426	-0.543	-1.112	-1.045	-0.564	+152	+066	+111	+117	+126	+132	11	-0.126
.45	-367	-395	-0.508	-1.092	-1.040	-0.535	+039	+060	+086	+110	+119	+123	12	-0.114
.50	-388	-359	-0.488	-1.055	-1.039	-0.522	+050	+049	+054	+120	+106	+105	13	+0.128
.55	-368	-350	-0.461	-0.963	-1.035		+078	+037	+046	+149	+117			
.60	-380	-377	-0.406	-0.864	-1.022	-0.477	+060	+015	+094	+115	+102	+075		
.65	-360	-366	-0.370	-0.640	-1.020		+045	+029	+129	+100	+098			
.68											+069			
.70	-365	-328	-0.316	-0.509	-1.013	-0.443	+029	+013	+137	+081		+034		
.73												+057		
.75	-303	-255	-0.253		-0.988		+045	+101	+120		+070			
.77											+057			
.79														
.80	-237	-181		-0.174		-0.936	-0.429	+080	+126			+046	-0.016	
.82														
.83														
.85	-146	-139		-0.155	-0.144		+069	+070						
.87														
.88	-088			-0.110	-0.123	-0.078	+056							
.90														
.91	-067			-0.028		-0.717	-0.421	+067						
.93	-045			-0.061			+025							
.95	-055	-025				-0.541	+019	+012				+004		
.96						+010								
.97				-0.031				+014						
.98	-001													

X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-299	-939					+411	+450					1	-0.036
.03	-382	-983	-1.047	-0.958			+436	+457	+326	+271	+233		2	-0.050
.05	-379	-1.084		-0.983	-0.961	-0.756	+406	+442		+271	+251	+222	3	-0.055
.08	-359	-1.219	-1.185				+400	+462	+278				4	-1.01
.10	-351	-1.248		-1.012	-0.976		+385			+244			5	-0.005
.15	-262	-1.011	-1.215		-0.990		+348	+363	+226		+209		6	-0.054
.20	-311						+349						7	-0.069
.25	-238	-144	-0.841	-1.102	-0.995	-0.649	+311	+168	+199	+183	+156	+219	8	-1.042
.30	-475	-323	-0.687	-1.111	-1.022	-0.623	+262	+109	+173	+159	+162	+212	9	-0.077
.35	-218	-400	-0.592	-1.118	-1.047	-0.507	+204	+074	+133	+136	+147	+162	10	-0.015
.40	-243	-423	-0.551	-1.125	-1.058	-0.580	+157	+065	+170	+119	+129	+126	11	-0.173
.45	-364	-397	-0.518	-1.103	-1.056	-0.549	+089	+061	+089	+131	+110	+119	12	-0.245
.50	-288	-401	-0.500	-1.067	-1.055	-0.536	+092	+056	+058	+118	+095	+097	13	+0.204
.55	-372	-397	-0.474	-0.980	-1.051		+082	+037	+062	+128	+107			
.60	-380	-381	-0.424	-0.870	-1.038	-0.491	+068	+020	+103	+103	+092	+071		
.65	-362	-371	-0.392	-0.661	-1.032		+053	-0.017	+118	+084	+083			
.68														
.70	-373	-340	-0.334	-0.534	-1.017	-0.457	+040	+040	+081	+065		+028		
.73							+054	+110	+088		+051			
.75	-315	-271	-0.269		-0.999					+058	+052			
.77														
.79														
.80	-252	-195		-0.189		-0.947	-0.443	+071	+037		+060		043	-0.022
.82														
.83														
.85	-154	-144		-0.168	-0.162	-0.861	+057	+097		+043	+043		025	
.87														
.88	-094			-0.115	-0.132	-0.889		+042						
.90	-066			-0.102		-0.739	-0.433		+031		+019		000	-0.105
.91									+046	+051				
.93	-051	-073		-0.035				+027	+050					
.95	-055	-030		-0.003		-0.563								
.96				-0.045										
.98	-006								+030		+024			

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TABLE IV. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X C	$\delta = 0.0$ $M = 1.00$						$\alpha = 0.0$ $P_{t,j}/P_{\infty} = 1.0$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.100	.085					.097	.126					1	-.205
.03	.078	.072	-.057	-.045			.092	.120	-.003	-.075	-.069		2	-.331
.05	.070	.076		-.037	-.047	-.084	.065	.137		-.066	-.073	-.150	3	-.327
.08	.074	.067	-.015				.070	.154	-.030				4	-.331
.10	.079	.048					.076						5	-.143
.15	.101	-.050	-.028				.056	.010	-.061	-.044			6	-.227
.20	.072						.121						7	-.411
.25	.044	.018	-.030	-.053	-.052	-.126	.087	-.014	-.044	-.062	-.080	-.164	8	-.404
.30	.027	-.015	-.029	-.052	-.086	-.149	.024	-.083	-.048	-.064	-.104	-.185	9	-.342
.35	.030	-.059	-.050	-.075	-.112	-.169	.014	-.130	-.068	-.082	-.138	-.226	10	-.387
.40	-.036	-.052	-.070	-.099	-.114	-.173	.025	-.125	-.078	-.119	-.177	-.233	11	-.309
.45	-.032	-.040	-.087	-.102	-.116	-.185	.023	-.123	-.104	-.092	-.146	-.195	12	-.291
.50	-.028	-.063	-.096	-.120	-.138	-.198	.081	-.123	-.137	-.162	-.215	-.274	13	-.305
.55	-.032	-.072	-.106	-.120	-.148		.121	-.086	-.181	-.171	-.228			
.60	-.048	-.076	-.094	-.127	-.141	-.209	.097	-.112	-.246	-.245	-.250	-.303		
.65	-.030	-.091	-.110	-.118	-.164		.076	-.165	-.367	-.283	-.283			
.68												-.276		
.70	-.079	-.091	-.115	-.148	-.176	-.232	.106	-.276	-.242	-.322	-.308			
.73												-.311		
.75	-.085	-.086	-.115				.175	-.379	-.160			-.269		
.77														
.79														
.80	-.101	-.095					.164	-.219	.282	-.160				
.82														
.83														
.85	-.102	-.138	-.133	-.147			.122	-.004						
.87														
.88														
.90														
.91	-.097						.115	-.149	-.209					
.93	-.094	-.101												
.95	-.105	-.080					.123		-.072	-.047				
.96														
.97														
.98														
X C	$\delta = 0.0$ $M = 1.00$						$\alpha = 0.0$ $P_{t,j}/P_{\infty} = 3.0$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.115	.081					.117	.135					1	-.208
.03	.052	.076	-.067	-.058	-.067	-.093	.053	.124	-.007	-.074	-.086		2	-.323
.05	.078	.074					.078	.135		-.069	-.077	-.144	3	-.307
.08	.083	.062	-.024				.054	.156	-.035				4	-.250
.10	.096	.043					.087			-.045	-.096		5	-.146
.15	.057	-.049	-.044				.095	.011	-.067				6	-.230
.20	.073						.119						7	-.409
.25	.045	.011	-.039	-.066	-.060	-.137	.082	-.020	-.042	-.071	-.086	-.158	8	-.344
.30	.022	-.023	-.037	-.064	-.092	-.143	.020	-.089	-.047	-.073	-.111	-.183	9	-.334
.35	.025	-.070	-.057	-.086	-.119	-.162	.020	-.136	-.093	-.095	-.143	-.223	10	-.384
.40	-.037	-.064	-.078	-.111	-.122	-.167	.033	-.132	-.091	-.126	-.182	-.228	11	-.244
.45	-.043	-.049	-.095	-.112	-.123	-.179	.033	-.105	-.102	-.152	-.199	-.247	12	-.279
.50	-.038	-.074	-.107	-.129	-.146	-.192	.090	-.133	-.142	-.166	-.226	-.270	13	-.267
.55	-.041	-.081	-.114	-.129	-.154		.126	-.107	-.184	-.178	-.234			
.60	-.063	-.086	-.103	-.135	-.148	-.204	.105	-.127	-.248	-.255	-.255	-.296		
.65	-.041	-.103	-.119	-.124	-.171		.090	-.177	-.366	-.287	-.269			
.68												-.211		
.70	-.093	-.102	-.122	-.155	-.184	-.226	.118	-.282	-.077	-.138	-.127	-.234		
.73	-.097	-.097	-.123		-.179		.188	-.259	-.040					
.77														
.79														
.80	-.112	-.103					.226	-.004						
.82														
.83														
.85	-.113	-.147					.028	-.016						
.87														
.88														
.90														
.91														
.93														
.95	-.110	-.090					.133		-.114					
.96														
.97														
.98														

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TABLE IV - Continued

TABLE IV.- Continued

		$\delta = 0.0$			$M = 1.0$			$\alpha = 4.9$		$P_{t,j}/P_{\infty} = 3.1$				
x c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp
.01	-e006	-e553					.267	.393					1	-e146
.03	-e038	-e564	-e644	-e568			.267	.358	.235	.231	.221		2	-e270
.05	-e003	-e449		-e576	-e597	-e634	.252	.339		.189	.178	.182	3	-e277
.08	-e078	-e011	-e685				.257	.362	.162				4	-e092
.10	-e176	-e112		-e589	-e576		.254			.145			5	-e082
.15	-e029	-e256	-e461		-e585		.239	.234	.111		.088		6	-e175
.20	-e024						.260						7	-e376
.25	-e121	-e101	-e128	-e497	-e568	-e647	.224	.076	.098	.072	.062	.138	8	-e177
.30	-e169	-e118	-e137	-e425	-e570	-e668	.179	.011	.072	.049	.012	.112	9	-e303
.35	-e175	-e159	-e168	-e38C	-e555	-e687	.150	.020	.044	.017	-eC23	.076	10	-e327
.40	-e146	-e178	-e192	-e342	-e520	-e698	.072	-e015	.021	-e026	-e070	.027	11	-e100
.45	-e132	-e158	-e209	-e301	-e482	-e702	.004	-e017	-e009	-eC65	-e089	.023	12	-e109
.50	-e138	-e177	-e225	-e290	-e457	-e710	.015	-e034	-eC56	-e389	.012	.014	13	-e093
.55	-e141	-e192	-e237	-e266	-e412		.022	-eC31	.114	-e102	.036			
.60	-e155	-e201	-e220	-e266	-e385	-e704	.019	-e059	-e197	-e330	.035	-e003		
.65	-e133	-e213	-e238	-e242	-e370		.020	-e116	.024	-e052	.034			
.68												.008		
.70	-e202	-e217	-e242	-e288	-e352	-e665	-e060	-e245	-e097	-e30		-e025		
.73												-e301		
.75	-e203	-e214	-e241		-e332		-e135	-e004	.083			-e303		
.77				-e285								.056		
.79				-e272								.007		
.80	-e224	-e224		-e264		-e311	-e576	-e031	.098			.001	-eC63	
.82				-e250	-e266							.031		
.83												.025	-e003	
.85	-e219	-e249			-e295		.048	.057				-e007		
.87	-e241	-e246	-e252					.043	.010	-e008				
.88	-e234							.034						
.90			-e234		-e278	-e466								
.91		-e218		-e228					.007			-e024	-e117	
.93		-e215	-e213						.015	-e020				
.95	-e212	-e195			-e197		-e017	-e040				-e024		
.96				-e194								-e028		
.97			-e142									-e028		
.98		-e145							.058					

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TABLE IV - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-.019	-.544					+.267	+.393					1	-.136
.03	-.059	-.553	-.655	-.534	-.563	-.615	+.273	+.350	+.239	+.237	+.224	+.194	2	-.249
.05	-.067	-.478					+.248	+.336		+.196	+.183	+.194	3	-.189
.08	-.064	-.089	-.680				+.258	+.359	+.165				4	-.007
.10	-.061	-.114					+.257			+.145			5	-.073
.15	-.048	-.256	-.509				+.286	+.232	+.114		+.093		6	-.163
.20							+.260						7	-.360
.25	-.092	-.146	-.093	-.529	-.597	-.635	+.220	+.074	+.105	+.077	+.064	+.124	8	-.034
.30	-.213	-.108	-.119	-.474	-.612	-.650	+.167	+.010	+.076	+.051	+.012	+.107	9	-.287
.35	-.183	-.153	-.159	-.417	-.607	-.678	+.114	-.016	-.051	-.019	-.018	-.080	10	-.315
.40	-.079	-.174	-.190	-.368	-.578	-.691	+.069	-.015	-.027	-.020	-.059	-.038	11	-.019
.45	-.135	-.151	-.205	-.429	-.547	-.697	+.000	-.014	-.003	-.056	-.024	-.033	12	-.026
.50	-.149	-.173	-.221	-.471	-.521	-.706	+.015	-.030	-.051	-.079	-.022	-.022	13	-.030
.55	-.142	-.183	-.232	-.420	-.470		-.010	-.025	-.106	-.041	-.036			
.60	-.174	-.196	-.214	-.424	-.430	-.696	-.016	-.055	-.188	-.057	-.034	-.014		
.65	-.167	-.203	-.232	-.219	-.392		-.021	-.110	-.060	-.046	-.031			
.68												-.005		
.70	-.261	-.212	-.240	-.264	-.363	-.653	-.053	-.235	-.093	-.022		-.016		
.73	-.203	-.206	-.232				-.018	-.046	-.081			-.004		
.75												-.011		
.77														
.79														
.80	-.224	-.217					-.002	-.096						
.82														
.83														
.85	-.219	-.249					+.051	-.036						
.87	-.236	-.241	-.246					-.016	-.051	-.002				
.88	-.229							-.041						
.90														
.91	-.222							-.059						
.93	-.218	-.210						-.020						
.95	-.219	-.194					-.012	-.060						
.96														
.97														
.98	-.155							-.080						
								-.096						

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-.017	-.542					+.268	+.393					1	-.119
.03	-.056	-.553	-.655	-.536	-.561	-.620	+.272	+.356	+.236	+.233	+.223	+.193	2	-.173
.05	-.065	-.481					+.249	+.337		+.190	+.180	+.193	3	-.027
.08	-.085	-.085	-.684				+.257	+.364	+.164				4	-.102
.10	-.080	-.115					+.257						5	-.072
.15	-.026	-.280	-.509	-.570	-.575		+.286	+.233	+.109	+.091			6	-.157
.20							+.259						7	-.227
.25	-.097	-.146	-.096	-.533	-.655	-.640	+.221	-.072	-.101	-.076	-.066	-.125	8	-.095
.30	-.212	-.108	-.123	-.477	-.620	-.653	+.172	-.009	-.071	-.048	-.026	-.112	9	-.253
.35	-.184	-.154	-.164	-.422	-.609	-.682	+.111	-.021	-.046	-.016	-.041	-.083	10	-.113
.40	-.082	-.173	-.194	-.369	-.577	-.694	+.049	-.014	-.024	-.023	-.033	-.042	11	-.116
.45	-.136	-.151	-.209	-.300	-.548	-.702	-.001	-.018	-.004	-.054	-.023	-.036	12	-.236
.50	-.146	-.168	-.225	-.277	-.514	-.712	-.017	-.036	-.052	-.022	-.067	-.021	13	-.141
.55	-.144	-.185	-.237	-.239	-.468		-.010	-.027	-.115	-.075	-.009			
.60	-.171	-.194	-.215	-.234	-.429	-.709	-.015	-.051	-.003	-.031	-.018	-.010		
.65	-.168	-.203	-.237	-.216	-.391		-.019	-.108	-.091	-.010	-.036			
.68														
.70	-.202	-.211	-.243	-.263	-.365	-.678	-.047	-.139	-.022	-.001		-.028		
.73	-.203	-.204	-.235				-.054	-.090	-.065					
.75														
.77														
.79														
.80	-.226	-.217					-.037	-.045		-.093				
.82										-.067				
.83														
.85	-.217	-.248					-.020	-.121						
.87	-.235	-.243	-.250					-.113	-.007	-.046		-.030		
.88	-.228							-.085						
.91	-.220							-.020	-.102	-.092	-.061	-.129		
.93	-.213	-.209						-.141	-.177					
.95	-.216	-.190					-.100	-.212						
.96														
.97								-.097		-.085				
.98	-.140	-.133												

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TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X c	$\delta = 0.0$ $M = 1.00$						$\alpha = 10.0$ $P_{t,j}/P_{\infty} = 1.0$						Nacelle Orifice number	C_p	
	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan								
	0.19	0.32	0.53	0.65	0.74	0.90		0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.152	-0.748	-0.945	-0.802	-0.618	-0.785	-0.790	.426	.524	.371	.313	.272		1	=.040
.03	-0.227	-0.778	-0.945	-0.802	-0.618	-0.785	-0.790	.476	.527	.371	.307	.273	.246	2	=.040
.05	-0.237	-0.808	-0.946	-0.803	-0.618	-0.785	-0.790	.446	.527	.371	.307	.273		3	=.152
.08	-0.225	-0.964	-1.007					.446	.526	.316				4	=.044
.10	-0.119	-0.972		-0.843	-0.802			.458	.526					5	=.006
.15	-0.146	-0.816	-1.096		-0.825			.470	.420	.266	.271			6	=.096
.20	-0.183							.420						7	=.306
.25	-0.188	-0.135	-0.674	-0.950	-0.839	-0.793		.375	.220	.237	.197	.194	.233	8	=.036
.30	-0.303	-0.203	-0.466	-0.959	-0.866	-0.802		.317	.156	.213	.164	.163	.223	9	=.238
.35	-0.379	-0.284	-0.394	-0.944	-0.901	-0.820		.253	.116	.171	.129	.154	.179	10	=.242
.40	-0.147	-0.315	-0.378	-0.972	-0.919	-0.828		.195	.103	.137	.093	.134	.147	11	=.063
.45	-0.233	-0.302	-0.369	-0.953	-0.928	-0.834		.126	.100	.103	.074	.131	.138	12	=.071
.50	-0.280	-0.296	-0.379	-0.923	-0.933	-0.840		.131	.095	.045	.107	.124	.123	13	=.064
.55	-0.277	-0.304	-0.381	-0.842	-0.945			.124	.074	.015	.149	.137			
.60	-0.289	-0.309	-0.361	-0.791	-0.949	-0.851		.102	.034	.011	.131	.125	.105		
.65	-0.289	-0.305	-0.377	-0.609	-0.956			.069	-0.039	.116	.127	.115			
.68															
.70	-0.318	-0.314	-0.375	-0.555	-0.950	-0.829		.423	-0.178	.156	.118		.069		
.73															
.75	-0.307	-0.306	-0.377		-0.949			.064	.065	.180		.089			
.77															
.79															
.80	-0.332	-0.322		-0.372		-0.953	-0.712	.069	.139			.099	.081	.026	
.82															
.83															
.85	-0.324	-0.340						.097	.156						
.87															
.88															
.90															
.91															
.93	-0.323	-0.307	-0.297		-0.299	-0.911	-0.622								
.95	-0.288	-0.200				-0.865		.059	.039						
.96															
.97															
.98															
X c	$\delta = 0.0$ $M = 1.00$						$\alpha = 10.0$ $P_{t,j}/P_{\infty} = 3.1$						Nacelle Orifice number	C_p	
	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan								
	0.19	0.32	0.53	0.65	0.74	0.90		0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.148	-0.732	-0.940	-0.805	-0.622	-0.780	-0.790	.432	.527	.373	.315	.276		1	=.044
.03	-0.224	-0.772	-0.940	-0.805	-0.622	-0.780	-0.790	.476	.527	.373	.305	.280	.247	2	=.34
.05	-0.231	-0.851						.473	.509					3	=.018
.08	-0.218	-0.954	-1.006					.473	.532	.316				4	=.084
.10	-0.213	-0.970						.464						5	=.014
.15	-0.127	-0.893	-1.098	-0.845	-0.801			.474	.423	.264	.273			6	=.086
.20	-0.174							.420						7	=.148
.25	-0.184	-0.123	-0.672	-0.955	-0.827	-0.794		.376	.222	.237	.203	.215	.230	8	=.072
.30	-0.298	-0.194	-0.467	-0.955	-0.867	-0.806		.323	.165	.208	.175	.185	.220	9	=.126
.35	-0.372	-0.276	-0.394	-0.962	-0.897	-0.825		.257	.123	.169	.146	.174	.178	10	=.048
.40	-0.139	-0.328	-0.380	-0.974	-0.917	-0.832		.198	.112	.160	.121	.152	.142	11	=.085
.45	-0.227	-0.295	-0.373	-0.955	-0.927	-0.836		.137	.114	.110	.118	.145	.137	12	=.081
.50	-0.279	-0.287	-0.381	-0.921	-0.934	-0.841		.139	.297	.057	.144	.133	.121	13	=.097
.55	-0.271	-0.299	-0.378	-0.842	-0.942			.133	.081	.021	.176	.142			
.60	-0.281	-0.301	-0.356	-0.746	-0.949	-0.855		.109	.046	.059	.151	.125	.101		
.65	-0.280	-0.297	-0.375	-0.607	-0.954			.084	-0.013	.170	.135	.111			
.68															
.70	-0.310	-0.305	-0.373	-0.560	-0.952	-0.932		.446	-0.037	.196	.113		.064		
.73															
.75	-0.300	-0.294	-0.375		-0.953			.050	.146	.163		.081			
.77															
.79															
.80	-0.323	-0.311		-0.370		-0.949	-0.725	.127	.185		.105				
.82															
.83															
.85	-0.314	-0.330		-0.373	-0.932	-0.936		.115	.131		.097	.071	.057		
.87															
.88															
.90															
.91															
.93															
.95	-0.281	-0.274			-0.263	-0.865		.051	.015			.004	.031	-0.050	
.96															
.97															
.99															

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TABLE IV. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

TABLE IV.- Continued

TABLE IV.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X C	$\delta = C.O.$ $M = 1.05$						$\alpha = 0^{\circ}0$ $P_{t,j}/P_{\infty} = 5:1$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.084	.026					.079	.072					1	-.245
.03	.052	.025	-.128	-.107	-.110	-.125	.043	.072	-.050	-.103	-.117		2	-.332
.05	.022	.033					.018	.104		-.118	-.121	-.162	3	-.338
.06	.019	.039	-.086				.010	.142	-.094				4	-.179
.10	.014	.039					.002						5	-.181
.15	.022	-.054	-.090				.025	.020	-.114		-.163		6	-.250
.20	.026						.097						7	-.448
.25	.034	-.053	-.117	-.103	-.108	-.160	.081	-.077	-.100	-.162	-.113	-.167	8	-.244
.30	-.028	-.066	-.093	-.102	-.133	-.176	.012	.128	-.110	-.127	-.134	-.186	9	-.347
.35	-.049	-.114	-.107	-.129	-.156	-.192	.135	-.165	-.191	-.130	-.165	-.221	10	-.387
.40	-.086	-.137	-.122	-.148	-.159	-.198	.070	-.197	-.139	-.162	-.196	-.232	11	-.136
.45	-.106	-.115	-.135	-.151	-.157	-.207	.181	-.170	-.147	-.183	-.205	-.247	12	-.155
.50	-.101	-.125	-.146	-.169	-.177	-.219	.161	-.172	-.185	-.183	-.215	-.265	13	-.147
.55	-.100	-.132	-.157	-.168	-.185		.199	-.177	-.206	-.181	-.225			
.60	-.108	-.140	-.138	-.167	-.183	-.233	.174	-.173	-.249	-.244	-.245	-.290		
.65	-.086	-.142	-.155	-.161	-.198		.143	-.202	-.350	-.273	-.269			
.68														
.70	-.134	-.147	-.158	-.189	-.216	-.257	.152	-.276	-.092	-.180		-.291		
.73														
.75	-.135	-.139	-.154				.209		-.191	-.192	-.171			
.77														
.79														
.80	-.150	-.143					.201	-.245		.250	-.148			
.82														
.83														
.85	-.147	-.177					.193		.108	-.018		-.137		
.87	-.171	-.169	-.177						.030	-.119	-.129			
.88									.048					
.90														
.91	-.150						.156	-.185	-.234		-.127	-.159	-.150	-.140
.93	-.150	-.156					.161			.172	-.145			-.227
.95	-.159						.185		.167	-.157			-.139	
.96							.162							
.97							.144							
.98							.131							

X C	$\delta = C.O.$ $M = 1.05$						$\alpha = 0^{\circ}0$ $P_{t,j}/P_{\infty} = 1:1$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.093	.039					.091	.078					1	-.223
.03	.063	.036	-.118	-.103	-.103	-.129	.053	.081	-.046	-.097	-.111		2	-.279
.05	.031	.040					.027	.110		-.112	-.119	-.159	3	-.184
.06	.027	.046	-.080				.019	.150	-.039				4	-.066
.10	.026	.046					.007			-.112			5	-.174
.15	.030	-.046	-.085				.033	.026	-.111		-.161		6	-.241
.20	.038						.106						7	-.382
.25	.043	-.043	-.111	-.100	-.103	-.154	.089	-.072	-.097	-.158	-.110	-.163	8	-.072
.30	-.019	-.059	-.089	-.099	-.128	-.170	.020	-.118	-.107	-.122	-.130	-.182	9	-.338
.35	-.039	-.104	-.101	-.123	-.150	-.184	.125	-.156	-.187	-.124	-.162	-.216	10	-.367
.40	-.077	-.128	-.116	-.146	-.152	-.192	.062	-.187	-.135	-.157	-.190	-.227	11	-.050
.45	-.098	-.108	-.130	-.148	-.152	-.201	.170	-.162	-.143	-.174	-.201	-.241	12	-.020
.50	-.093	-.117	-.143	-.163	-.173	-.212	.150	-.164	-.180	-.176	-.210	-.260	13	-.008
.55	-.091	-.124	-.152	-.162	-.179		.190	-.169	-.201	-.174	-.220			
.60	-.099	-.131	-.133	-.161	-.176	-.228	.160	-.165	-.237	-.232	-.234	-.280		
.65	-.080	-.134	-.149	-.155	-.192		.135	-.195	-.161	-.140	-.150			
.68														
.70	-.123	-.137	-.152	-.182	-.210	-.250	.141	-.258	-.041	-.116		-.201		
.73														
.75	-.126	-.130	-.150				.202		.177	-.032	-.155			
.77														
.79														
.80	-.141	-.135					.194	-.238	.127	-.102				
.82														
.83														
.85	-.138	-.167					.174		.086	-.309				
.87							.163	-.177		.185	-.241			
.88							.163	-.171		.136	-.081	-.226		
.90							.160	-.178	-.229		.045	-.092	-.228	-.205
.91							.151	-.154			.044	-.126		
.93							.141	-.151			.095	-.111		
.95	-.134	-.131					.156	-.178	.153	-.134		-.120		-.229
.96														
.97														
.98														

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TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.18	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-0.051	-0.592					+0.221	+0.360					1	-0.132
.03	-0.049	-0.556	-0.636	-0.608			+0.217	+0.329	+0.195	+0.180	+0.200		2	-0.245
.05	-0.117	-0.559		-0.612	-0.591	-0.608	+0.170	+0.319	+0.143	+0.157	+0.144		3	-0.411
.08	-0.122	-0.083		-0.679			+0.175	+0.355	+0.121				4	-0.276
.10	-0.179	-0.112		-0.624	-0.595		+0.181		+0.099				5	-0.072
.15	-0.071	-0.201		-0.524	-0.616		+0.251	+0.227		+0.048	+0.081		6	-0.148
.20	-0.070						+0.243						7	-0.331
.25	-0.066	-0.193	-0.173	-0.519	-0.611	-0.610	+0.217	+0.053	+0.060	+0.057	+0.060		8	-0.341
.30	-0.157	-0.194	-0.180	-0.426	-0.621		+0.151	-0.021	+0.024	+0.038	+0.023		9	-0.268
.35	-0.237	-0.189	-0.215	-0.354	-0.607	-0.634	+0.084	-0.049	+0.018	+0.012	-0.013		10	-0.290
.40	-0.157	-0.217	-0.233	-0.300	-0.572	-0.644	+0.046	-0.045	+0.025	+0.014	-0.057		11	-0.244
.45	-0.179	-0.198	-0.237	-0.266	-0.538	-0.650	+0.082	-0.059	+0.011	-0.049	-0.073		12	-0.239
.50	-0.191	-0.206	-0.251	-0.257	-0.502	-0.658	+0.041	-0.066	+0.053	-0.061	-0.100		13	-0.249
.55	-0.186	-0.219	-0.258	-0.260	-0.446		+0.057	-0.044	+0.097	-0.080	-0.118			
.60	-0.207	-0.229	-0.238	-0.263	-0.402	-0.663	+0.037	-0.054	+0.164	-0.153	-0.142			
.65	-0.200	-0.225	-0.233	-0.255	-0.365		+0.023	-0.099	+0.271	-0.184	-0.167			
.68											-0.160			
.70	-0.226	-0.220	-0.248	-0.286	-0.338	-0.666	+0.050	-0.201	-0.136	-0.168				-0.062
.73									-0.100	-0.322	-0.061			-0.108
.75	-0.222	-0.221	-0.248		-0.317						-0.054			
.77					-0.281						-0.092			
.79					-0.275						-0.020			
.80	-0.240	-0.227			-0.290	-0.588	+0.229	-0.154		+0.021				-0.013
.82				-0.266						+0.026	-0.001			
.83			-0.254	-0.270						+0.050	-0.029			
.85	-0.228	-0.248			-0.272					+0.047				-0.001
.87	-0.212	-0.253	-0.267											
.88	-0.233													
.90	-0.243				-0.265	-0.518								-0.075
.91	-0.229		-0.248							+0.035	-0.002			
.93	-0.225	-0.239								+0.015	-0.009			
.95	-0.219	-0.213			-0.261					+0.001	-0.003			
.96					-0.251						-0.012			
.97			-0.211								-0.009			
.98	-0.179										-0.022			
<hr/>														
Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle		
X c	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-0.049	-0.587					+0.225	+0.363					1	-0.132
.03	-0.097	-0.556	-0.639	-0.608			+0.219	+0.329	+0.199	+0.190	+0.203		2	-0.242
.05	-0.110	-0.360			-0.613	-0.592	-0.608	+0.174	+0.320	+0.154	+0.160	+0.146	3	-0.364
.08	-0.116	-0.081	-0.680				+0.182	+0.357	+0.126				4	-0.199
.10	-0.124	-0.111		-0.626	-0.593		+0.186		+0.055		+0.080		5	-0.070
.15	-0.067	-0.207		-0.524	-0.618		+0.251	+0.228	+0.050				6	-0.148
.20	-0.063	-0.193	-0.166	-0.518	-0.612	-0.611	+0.246						7	-0.330
.25	-0.063	-0.193	-0.166	-0.514	-0.612	-0.611	+0.219	+0.049	+0.074	+0.065	+0.061	+0.036	8	-0.313
.30	-0.156	-0.195	-0.174	-0.424	-0.620	-0.619	+0.152	-0.023	+0.027	+0.046	+0.020	+0.019	9	-0.266
.35	-0.245	-0.189	-0.209	-0.346	-0.608	-0.635	+0.088	-0.077	+0.020	+0.014	-0.012	-0.022	10	-0.284
.40	-0.150	-0.216	-0.224	-0.296	-0.573	-0.643	+0.047	-0.088	+0.011	+0.016	-0.057	-0.062	11	-0.202
.45	-0.177	-0.199	-0.235	-0.262	-0.540	-0.650	+0.074	-0.052	-0.009	-0.047	-0.073	-0.034	12	-0.217
.50	-0.186	-0.205	-0.246	-0.254	-0.503	-0.658	+0.032	-0.060	-0.053	-0.058	-0.100	-0.001	13	-0.211
.55	-0.181	-0.217	-0.254	-0.255	-0.448		+0.039	-0.048	-0.093	-0.074	-0.118			
.60	-0.203	-0.225	-0.232	-0.261	-0.406	-0.664	+0.029	-0.059	-0.162	-0.131		+0.006		
.65	-0.197	-0.223	-0.247	-0.250	-0.369		+0.023	-0.099	-0.269	-0.128	-0.003			-0.001
.68														
.70	-0.220	-0.229	-0.247	-0.280	-0.342	-0.664	+0.049	-0.200	+0.040	-0.004				-0.012
.73					-0.318				-0.100	-0.230	-0.052			
.75	-0.216	-0.218	-0.242						-0.055					
.77				-0.275										
.79				-0.269										
.80	-0.233	-0.225			-0.295	-0.580	+0.208	+0.062		+0.005				-0.035
.82			-0.262						+0.006					
.83			-0.249	-0.266					+0.014	-0.004				
.85	-0.223	-0.247			-0.275		+0.116	+0.337		+0.027				-0.001
.87	-0.238	-0.250	-0.263						+0.004	-0.012				
.88	-0.232								+0.029					
.90			-0.238		-0.264	-0.513			-0.002		-0.010	-0.088		
.91	-0.226			-0.244					+0.006		-0.011			
.93	-0.225	-0.234			-0.263		+0.002	+0.112		-0.007		-0.010		
.95	-0.211	-0.212			-0.245		+0.025					-0.021		
.96				-0.208						-0.023				
.97			-0.185							-0.043				

CONFIDENTIAL

TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X c	Wing upper surface stations, fraction of semispan						$\alpha = 5.0$	$P_{t_1}/P_{\infty} = 5.1$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.051	-0.592					.220	.360		
.03	-0.097	-0.563	-0.643	-0.610			.215	.329	.199	.132
.05	-0.116	-0.372					.175	.320	.191	.241
.08	-0.121	-0.086	-0.679				.181	.354	.153	.280
.10	-0.129	-0.117					.186		.160	.144
.15	-0.070	-0.211	-0.590				.251	.223	.103	.082
.20	-0.070						.243			.073
.25	-0.068	-0.197	-0.166	-0.522	-0.612		.210	.047	.073	.031
.30	-0.159	-0.197	-0.177	-0.428	-0.623		.147	.025	.028	.196
.35	-0.247	-0.193	-0.210	-0.357	-0.612		.084	.073	.021	.17
.40	-0.154	-0.220	-0.228	-0.298	-0.580		.041	.090	.011	.266
.45	-0.180	-0.202	-0.235	-0.264	-0.542		.080	.047	.011	.289
.50	-0.189	-0.208	-0.246	-0.260	-0.507		.037	.060	.052	.020
.55	-0.185	-0.221	-0.253	-0.255	-0.453		.042	.052	.060	.014
.60	-0.207	-0.228	-0.232	-0.263	-0.411		.035	.060	.103	.007
.65	-0.199	-0.226	-0.247	-0.254	-0.376		.026	.099	.202	.002
.68										.002
.70	-0.221	-0.231	-0.249	-0.284	-0.346	-0.665	-0.053	-0.207	.013	-0.017
.73										.002
.75	-0.220	-0.223	-0.243				-0.105	-0.059	.051	-0.003
.77									.011	
.79									.003	
.80	-0.235	-0.228					-0.171	.003		
.82									.001	
.83									.023	
.85	-0.225	-0.249					.012	.056		
.87								.001	.031	
.88								.042		
.90								.001	.057	
.91	-0.227	-0.238	-0.246	-0.264	-0.518			.004	.020	-0.043
.93								.002	.012	-0.106
.95	-0.222	-0.215					-0.025	.001		-0.043
.96									.013	
.97									.016	
.98	-0.192							-0.037		

X c	Wing upper surface stations, fraction of semispan						$\alpha = 5.0$	$P_{t_1}/P_{\infty} = 7.1$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.049	-0.584					.223	.361		.126
.03	-0.093	-0.561	-0.639	-0.606			.217	.335	.201	.203
.05	-0.114	-0.361		-0.611	-0.592	-0.609	.175	.324	.193	.202
.08	-0.118	-0.033	-0.678				.181	.357	.157	.142
.10	-0.127	-0.109		-0.626	-0.591		.188		.161	.048
.15	-0.068	-0.200	-0.528		-0.613		.255	.230	.052	.070
.20	-0.071						.248			.147
.25	-0.065	-0.191	-0.164	-0.521	-0.608	-0.608	.216	.055	.078	.307
.30	-0.163	-0.192	-0.172	-0.421	-0.619	-0.617	.147	.015	.029	.022
.35	-0.247	-0.188	-0.208	-0.356	-0.608	-0.633	.086	.071	.022	.265
.40	-0.152	-0.211	-0.227	-0.292	-0.573	-0.641	.043	.084	.013	.272
.45	-0.177	-0.197	-0.232	-0.261	-0.535	-0.647	.076	.041	.006	.147
.50	-0.188	-0.205	-0.245	-0.252	-0.502	-0.654	.033	.059	.049	.121
.55	-0.180	-0.214	-0.252	-0.255	-0.449	-0.654	.039	.046	.091	.074
.60	-0.203	-0.221	-0.230	-0.261	-0.404	-0.660	.029	.035	.155	.044
.65	-0.196	-0.220	-0.244	-0.250	-0.366		.024	.096	.070	.033
.68										
.70	-0.220	-0.225	-0.247	-0.282	-0.341	-0.651	-0.047	-0.184	.042	-0.069
.73									.063	
.75	-0.215	-0.217	-0.241		-0.315		-0.098	.070	.091	-0.093
.77									.069	
.79									.089	
.80	-0.234	-0.222					-0.012	-0.037		-0.045
.82									.013	
.83									.003	
.85	-0.222	-0.243					-0.018	.026	.074	
.87								.079	.052	
.88								.087		
.90									.000	
.91	-0.224									
.93	-0.222	-0.233								
.95	-0.222	-0.210								
.96										
.97										
.98	-0.189									

CONT'D ON PAGE 2

TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$\frac{x}{c}$	$\delta = 0^\circ C$						$M = 1.05$						$\alpha = 10^\circ 0$		$P_{t,j}/P_{\infty} = 1.0$		Nacelle Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65			
.01	-190	-747	-766	-949	-809	-	-331	-511	-	-	-	-	1	-043	-	-	-	-	
.03	-261	-766	-949	-809	-823	-792	-765	-504	-339	-297	-270	-	2	-170	-	-	-	-	
.05	-266	-886	-	-	-	-	-411	-692	-288	-268	-247	-	3	-303	-	-	-	-	
.08	-239	-922	-992	-	-	-	-419	-512	-280	-	-	-	4	-056	-	-	-	-	
.10	-258	-908	-	-043	-801	-	-414	-	-263	-	-	-	5	-023	-	-	-	-	
.15	-188	-729	-1.048	-	-821	-	-446	-400	-228	-206	-	-	6	-066	-	-	-	-	
.20	-194	-	-	-	-	-	-499	-	-	-	-	-	7	-271	-	-	-	-	
.25	-186	-143	-739	-929	-837	-766	-357	-183	-217	-191	-172	-241	8	-194	-	-	-	-	
.30	-234	-294	-479	-937	-866	-773	-292	-109	-195	-160	-121	-231	9	-211	-	-	-	-	
.35	-400	-508	-382	-945	-592	-790	-421	-61	-160	-128	-93	-198	10	-220	-	-	-	-	
.40	-230	-341	-365	-957	-910	-798	-323	-64	-132	-87	-67	-170	11	-033	-	-	-	-	
.45	-265	-336	-360	-959	-916	-801	-086	-677	-135	-649	-552	-169	12	-026	-	-	-	-	
.50	-288	-330	-371	-936	-921	-806	-100	-690	-254	-621	-102	-158	13	-032	-	-	-	-	
.55	-310	-326	-373	-863	-931	-	-109	-772	-003	-003	-144	-	-	-	-	-	-	-	
.60	-311	-327	-353	-762	-939	-816	-690	-646	-082	-C74	-142	-148	-	-	-	-	-	-	
.65	-300	-319	-369	-618	-943	-	-074	-017	-128	-137	-138	-	-	-	-	-	-	-	
.68	-	-	-	-	-	-	-	-	-	-	-	-125	-	-	-	-	-	-	
.70	-322	-320	-367	-535	-933	-813	-044	-147	-137	-138	-	-119	-	-	-	-	-	-	
.75	-	-	-	-	-	-	-	-	-	-	-	-123	-	-	-	-	-	-	
.77	-312	-313	-363	-	-	-937	-048	-065	-183	-	-	-128	-	-	-	-	-	-	
.79	-	-	-	-	-	-427	-	-	-	-	-	-135	-	-	-	-	-	-	
.80	-328	-321	-	-	-	-400	-	-938	-714	-124	-116	-	-132	-	-126	-	-081	-	
.82	-	-	-	-	-	-570	-	-	-	-	-	-177	-	-	-	-	-	-	
.85	-321	-332	-363	-347	-	-	-097	-145	-173	-	-	-24	-	-	-	-	-	-	
.87	-324	-367	-320	-	-	-932	-067	-165	-112	-	-	-115	-	-	-	-	-	-	
.88	-	-	-319	-	-	-	-167	-	-	-	-	-	-	-	-	-	-	-	
.90	-	-	-357	-	-	-297	-914	-638	-	-140	-096	-593	-015	-	-	-	-	-	-
.91	-	-	-311	-	-	-297	-	-	-	-153	-	-	-	-	-	-	-	-	-
.93	-	-	-313	-	-	-326	-	-	-	-120	-116	-	-	-	-	-	-	-	-
.95	-279	-292	-	-	-	-297	-	-886	-	-110	-092	-	-	-	-	-	-	-	-
.96	-	-	-	-	-	-230	-	-	-	-	-	-078	-	-	-	-	-	-	-
.97	-	-	-	-	-	-249	-	-	-	-	-	-045	-	-	-	-	-	-	-
.98	-	-	-	-	-	-	-	-	-	-	-	-	-	-	-	-	-	-	-
$\delta = 0^\circ 0$																			
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						$M = 1.05$						$\alpha = 10^\circ 0$		$P_{t,j}/P_{\infty} = 3+1$		Nacelle Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65			
.01	-184	-732	-	-	-	-	-336	-514	-	-	-	-	1	-039	-	-	-	-	-
.03	-257	-754	-940	-801	-811	-779	-765	-506	-342	-303	-272	-	2	-149	-	-	-	-	-
.05	-261	-862	-	-	-	-	-412	-495	-293	-269	-264	-	3	-100	-	-	-	-	-
.08	-253	-916	-988	-	-	-	-423	-517	-285	-	-	-	4	-048	-	-	-	-	-
.10	-253	-906	-	-838	-786	-	-422	-	-269	-	-	-	5	-026	-	-	-	-	-
.15	-178	-708	-1.040	-	-806	-	-450	-406	-235	-208	-	-	6	-065	-	-	-	-	-
.20	-186	-	-	-	-	-	-406	-	-	-	-	-	7	-259	-	-	-	-	-
.25	-181	-135	-727	-923	-823	-765	-367	-186	-219	-197	-177	-250	8	-011	-	-	-	-	-
.30	-247	-288	-467	-930	-853	-775	-302	-119	-195	-166	-134	-237	9	-202	-	-	-	-	-
.35	-403	-298	-374	-941	-881	-792	-229	-003	-163	-135	-128	-202	10	-251	-	-	-	-	-
.40	-216	-337	-355	-949	-898	-798	-161	-078	-137	-095	-137	-172	11	-051	-	-	-	-	-
.45	-255	-324	-350	-951	-905	-801	-098	-093	-108	-059	-150	-170	12	-047	-	-	-	-	-
.50	-281	-320	-364	-926	-912	-808	-114	-085	-059	-054	-149	-158	13	-051	-	-	-	-	-
.55	-303	-317	-368	-852	-918	-	-118	-078	-001	-166	-160	-	-	-	-	-	-	-	-
.60	-302	-317	-343	-750	-923	-818	-099	-050	-074	-171	-148	-145	-	-	-	-	-	-	-
.65	-289	-309	-362	-601	-927	-	-077	-007	-184	-163	-137	-	-	-	-	-	-	-	-
.70	-310	-312	-362	-521	-919	-817	-049	-128	-217	-141	-	-116	-	-	-	-	-	-	-
.73	-	-	-	-	-	-	-	-	-	-	-	-	-	-	-	-	-	-	-
.75	-303	-305	-354	-	-	-920	-	-036	-123	-198	-	-120	-	-	-	-	-	-	-
.77	-	-	-	-	-	-410	-	-	-	-	-	-133	-	-	-	-	-	-	-
.79	-	-	-	-	-	-380	-	-	-	-	-	-119	-	-	-	-	-	-	-
.80	-322	-311	-	-	-	-927	-	-693	-	-111	-198	-	-	-	-	-	-	-	-
.82	-	-	-	-	-	-359	-	-	-	-	-	-141	-	-	-	-	-	-	-
.83	-	-	-	-	-	-350	-	-329	-	-	-	-136	-	-	-	-	-	-	-
.85	-311	-324	-	-	-	-919	-	-	-	-128	-156	-	-	-	-	-	-	-	-
.87	-317	-354	-304	-	-	-	-	-	-	-142	-127	-101	-	-	-	-	-	-	-
.88	-310	-	-	-	-	-	-	-	-	-136	-	-	-	-	-	-	-	-	-
.90	-	-	-	-	-	-342	-	-281	-	-606	-	-122	-	-109	-	-085	-	-013	-
.91	-	-	-	-	-	-303	-	-	-	-	-	-122	-	-	-	-	-	-	-
.93	-	-	-	-	-	-305	-	-309	-	-	-	-055	-	-094	-	-085	-	-	-
.95	-268	-284	-	-	-	-870	-	-	-	-099	-071	-	-	-	-	-	-	-	-
.96	-	-	-	-	-	-220	-	-282	-	-	-	-064	-	-	-	-	-	-	-
.97	-	-	-	-	-	-241	-	-	-	-	-	-032	-	-	-	-	-	-	-

TABLE IV.-- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-0.181	-0.730					.338	.517					1	-0.032
.03	-0.253	-0.552	-0.943	-0.794			.406	.511	.351	.308	.274		2	-0.126
.05	-0.253	-0.857					.416	.500		.297	.274	.267	3	-0.026
.08	-0.251	-0.912	-0.934				.430	.519	.294				4	.113
.10	-0.249	-0.903					.424				.271		5	.031
.15	-0.176	-0.723	-1.040				.457	.415	.244		.213		6	-0.058
.20	-0.184						.408						7	-0.238
.25	-0.175	-0.127	-0.725	-0.917	-0.818	-0.762	.369	.196	.228	.201	.188	.251	8	.089
.30	-0.243	-0.278	-0.470	-0.923	-0.832	-0.771	.305	.129	.200	.174	.159	.241	9	-0.194
.35	-0.396	-0.286	-0.373	-0.934	-0.878	-0.789	.230	.089	.166	.139	.165	.205	10	-0.186
.40	-0.208	-0.336	-0.351	-0.942	-0.895	-0.797	.197	.082	.143	.102	.158	.176	11	.131
.45	-0.250	-0.322	-0.350	-0.945	-0.905	-0.799	.106	.059	.115	.072	.160	.171	12	.148
.50	-0.277	-0.317	-0.357	-0.921	-0.909	-0.806	.123	.090	.065	.115	.153	.159	13	.139
.55	-0.297	-0.314	-0.363	-0.849	-0.915		.125	.079	.010	.192	.163			
.60	-0.294	-0.310	-0.339	-0.747	-0.920	-0.816	.105	.056	.029	.175	.150	.144		
.65	-0.286	-0.304	-0.358	-0.602	-0.925		.086	-0.003	.203	.159	.140			
.68														
.70	-0.307	-0.309	-0.356	-0.520	-0.919	-0.815	.058	-0.116	.214	.137		.113		
.73	-0.297	-0.301	-0.350				.014	.160	.191					
.75														
.77														
.79														
.80	-0.317	-0.308					.139	.206						
.82														
.83														
.85	-0.307	-0.317					.129	.125						
.87	-0.310	-0.348	-0.304				.129	.150						
.88	-0.303						.150							
.90														
.91	-0.298						.145	.097						
.93	-0.297	-0.297					.085	.038						
.95	-0.266	-0.276					.094	.020						
.96														
.97														
.98	-0.239													

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.80	0.19	0.32	0.53	0.65	0.74	0.80	Orifice number	C _p
.01	-0.187	-0.732					.336	.510					1	-0.021
.03	-0.256	-0.124	-0.955	-0.804			.404	.505	.345	.305	.271		2	-0.055
.05	-0.253	-0.588					.417	.490		.293	.270	.261	3	.081
.08	-0.255	-0.528	-0.993				.425	.512	.290				4	.167
.10	-0.254	-0.521					.421				.267		5	.031
.15	-0.186	-0.146	-1.044				.449	.402	.238		.212		6	.054
.20	-0.190						.404						7	.118
.25	-0.133	-0.731	-0.929	-0.829	-0.772		.365	.185	.221	.196	.203	.246	8	.169
.30	-0.254	-0.288	-0.472	-0.938	-0.860	-0.781	.297	.118	.193	.170	.183	.231	9	-0.136
.35	-0.406	-0.298	-0.375	-0.948	-0.887	-0.800	.227	.077	.161	.132	.179	.194	10	.011
.40	-0.220	-0.342	-0.389	-0.984	-0.905	-0.805	.159	.070	.137	.105	.158	.181	11	.182
.45	-0.252	-0.332	-0.353	-0.959	-0.912	-0.811	.100	.095	.110	.112	.152	.155	12	.283
.50	-0.286	-0.327	-0.364	-0.930	-0.918	-0.816	.115	.078	.059	.156	.143	.139	13	.223
.55	-0.303	-0.324	-0.371	-0.857	-0.926		.118	.073	.010	.183	.155			
.60	-0.305	-0.326	-0.345	-0.752	-0.931	-0.825	.103	.040	.147	.152	.143	.122		
.65	-0.294	-0.319	-0.363	-0.604	-0.938		.078	-0.009	.192	.142	.131			
.68														
.70	-0.316	-0.319	-0.363	-0.523	-0.929	-0.819	.055	-0.025	.137	.135		.089		
.73	-0.306	-0.313	-0.354				.040	.169	.173					
.75														
.77														
.79														
.80	-0.322	-0.321					.131	.097						
.82														
.83														
.85	-0.316	-0.332					.109	.199						
.87	-0.314	-0.351	-0.334				.115	.113	.058					
.88	-0.315						.115	.026	.016					
.90	-0.342						.000							
.91	-0.309						.000							
.93	-0.304	-0.297					.103	-0.114						
.95	-0.274	-0.281					.003	-0.064						
.96														
.97														
.98	-0.232	-0.199					.038	.096						

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TABLE IV.- Continued						
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4						
$\alpha = 0.0 \quad M = 1.09$						
Wing upper surface stations, fraction of semispan				$\alpha = 5.0 \quad p_{t,j}/p_{\infty} = 1.0$		
X c	0.19	0.32	0.53	0.65	0.74	0.90
.01	-.070	-.572				
.03	-.111	-.515	-.508	-.535		
.05	-.120	-.311				
.08	-.120	-.120	-.627			
.10	-.127	-.100				
.15	-.111	-.154	-.649	-.563	-.527	
.20	-.077					
.25	-.046	-.163	-.093	-.480	-.350	-.522
.30	-.109	-.171	-.103	-.395	-.563	-.531
.35	-.211	-.177	-.130	-.320	-.557	-.544
.40	-.151	-.186	-.135	-.246	-.532	-.553
.45	-.151	-.142	-.130	-.184	-.503	-.559
.50	-.154	-.091	-.138	-.168	-.469	-.568
.55	-.102	-.059	-.151	-.173	-.420	
.60	-.027	-.084	-.147	-.187	-.372	-.576
.65	-.063	-.120	-.175	-.182	-.330	
.68						
.70	-.093	-.134	-.183	-.224	-.303	-.585
.73						
.75	-.132	-.140	-.187		-.271	
.77						
.79						
.80	-.158	-.163				
.82						
.83						
.85	-.171	-.194	-.209	-.219		
.87						
.88						
.90						
.91						
.93						
.95	-.196	-.194	-.205			
.96						
.97						
.98						

$\alpha = 0.0 \quad M = 1.09$						
$\alpha = 5.0 \quad p_{t,j}/p_{\infty} = 5.0$						
Wing upper surface stations, fraction of semispan				Wing lower surface stations, fraction of semispan		
X c	0.19	0.32	0.53	0.65	0.74	0.90
.01	-.056	-.558				
.03	-.099	-.507	-.568	-.527		
.05	-.137	-.297				
.08	-.138	-.096	-.611			
.10	-.116	-.082				
.15	-.052	-.142	-.430	-.529		
.20	-.058					
.25	-.025	-.150	-.087	-.455	-.533	-.519
.30	-.101	-.161	-.293	-.358	-.546	-.529
.35	-.196	-.152	-.124	-.279	-.540	-.546
.40	-.126	-.162	-.129	-.214	-.512	-.595
.45	-.137	-.117	-.132	-.175	-.480	-.558
.50	-.131	-.077	-.148	-.166	-.447	-.567
.55	-.074	-.073	-.164	-.175	-.398	
.60	-.043	-.106	-.156	-.188	-.355	-.573
.65	-.085	-.128	-.177	-.185	-.312	
.70	-.104	-.142	-.182	-.223	-.289	-.579
.73						
.75	-.137	-.146	-.186		-.256	
.77						
.79						
.80	-.159	-.163				
.82						
.83						
.85	-.172	-.192				
.87						
.88						
.90						
.91						
.93						
.95	-.182	-.181				
.96						
.97						
.99						

TABLE IV. - Continued

TABLE IV.- Continued

TABLE IV.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

 $\delta = -7.5 \quad M = 0.80 \quad \alpha = -0.1 \quad P_{t,i}/P_{\infty} = 5.1$

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.093	-.012					.093	.019						
.03	.057	-.031	-.108	-.048			.057	.009	-.077	-.127	-.175			
.05	.032	-.024					.036	.021		-.115	-.144			
.08	.017	-.037	-.065	-.043	-.020	.054	.016	.025	-.094	-.144	-.220			
.10	.004	-.055					-.003			-.120				
.15	.017	-.079	-.070	-.055	-.032		-.010	-.069	-.119		-.154			
.20	-.020						.004							
.25	-.030	-.055	-.065	-.042	-.023		.015	-.115	-.127	-.124	-.118	-.127		
.30	-.040	-.065	-.060	-.032	-.026		.063	-.150	-.141	-.139	-.144	-.153	9	-.234
.35	-.057	-.092	-.060	-.037	-.029		.080	-.172	-.145	-.146	-.160	-.174	10	-.259
.40	-.037	-.092	-.059	-.048	-.021		.080	-.168	-.157	-.170	-.184	-.210	11	-.082
.45	-.065	-.074	-.051	-.030	-.006		.130	-.175	-.176	-.171	-.184	-.191	12	-.041
.50	-.061	-.074	-.048	-.025	-.009		.121	-.174	-.204	-.157	-.184	-.178	13	-.043
.55	-.049	-.067	-.034	-.007	.010		.173	-.181	-.212	-.114	-.172			
.60	-.054	-.050	-.017	-.027	-.022		.165	-.206	-.158	-.153	-.176	-.146		
.65	-.031	-.026	-.039	-.083	-.015		.157	-.241	-.119	-.169	-.188			
.68														
.70	-.012	.011	.087	.127	.017	-.038	-.155	-.199	-.131	-.232		-.112		
.73														
.75	.013	.058	.149				.012		-.139	-.121	-.161			
.77														
.79														
.80	.067	.136					.008	-.012	-.150	-.150				
.82														
.83														
.85	.154	.265					.008		-.223	-.374	-.917			
.87														
.88														
.89														
.90	.196	.165	.125	.056	.014	.006	.042		-.466					
.91														
.92	.142													
.93														
.94	.109													
.95														
.96	.074													
.97														
.98	.034	.046												

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.085	-.016					.085	.011						
.03	.054	-.034	-.107	-.052	-.026	.051	.050	-.004	-.086	-.142	-.173			
.05	.026	-.053					.029	-.006		-.126	-.148			
.08	.012	-.048	-.071				.011	.016	-.102					
.10	-.003	-.063	-.062	-.036	-.039		-.007			-.130				
.15	.011	-.084	-.077				.016	-.078	-.124		-.159			
.20	-.029						.001							
.25	-.039	-.067	-.073	-.051	-.029		.019	-.121	-.133	-.133	-.122	-.132		
.30	-.049	-.074	-.068	-.039	-.032		.069	-.156	-.148	-.145	-.151	-.159	9	-.221
.33	-.067	-.095	-.067	-.045	-.036		.084	-.185	-.152	-.154	-.166	-.178	10	-.237
.40	-.044	-.102	-.065	-.059	-.027		.045	-.175	-.164	-.180	-.190	-.211	11	-.039
.45	-.072	-.087	-.059	-.039	-.012		.135	-.182	-.181	-.179	-.192	-.194	12	-.032
.50	-.069	-.084	-.055	-.034	-.017		.132	-.181	-.269	-.166	-.194	-.184	13	-.028
.55	-.057	-.077	-.039	-.001	.000		.168	-.190	-.207	-.126	-.184			
.60	-.062	-.059	.011	.018	.014		.166	-.208	-.155	-.172	-.187	-.148		
.65	-.037	-.036	.032	.073	.006		.158	-.243	-.129	-.193	-.197			
.68														
.70	-.016	-.002	.082	.116	.008	-.041	-.157	-.189	-.188	-.255		-.109		
.73														
.75	.012	.052	.164		.005		.0141	-.130	-.221					
.77														
.79														
.80	.062	.120		.287		.000	-.019	-.164	-.263					
.82														
.83														
.85	.150	.256		.229	.071	.001	-.246	-.367						
.87														
.88														
.89														
.90	.189		.117		.046	.001	-.417	-.134						
.91														
.92	.137	.157					-.296	-.347						
.93														
.94	.104	.115		.088			-.225	-.191						
.95	-.071	.073		.035			-.147	-.102						
.96														
.97														
.98	.026	.041		.064			-.059	-.015						

TABLE IV.- Continued

TABLE IV.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

TABLE IV. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	-e400	-e1086					e387	e413						
.03	-e346	-e1448	-e1088	-e897			e398	e432	e328	e271	e253			
.05	-e486	-e1227			-e964	-e715	-e351	e385	e422	e276	e265	e249		
.08	-e436	-e1475	-e1199				e382	e434	e284					
.10	-e417	-e1543		-e910	-e720		e373			e244				
.15	-e339	-e888	-e1194		-e720		e364	e348	e223		e210			
.20	-e470						e340							
.25	-e337	-e209	-e728	-e1010	-e703	-e330	e286	e166	e197	e178	e183	e200		
.30	-e273	-e353	-e598	-e1024	-e704	-e329	e221	e109	e167	e146	e142	e183	9	-e076
.35	-e245	-e464	-e488	-e1011	-e731	-e329	e221	e082	e126	e123	e121	e129	10	-e067
.40	-e361	-e310	-e434	-e997	-e744	-e324	e158	e02	e197	e193	e180	e191	11	-e075
.45	-e321	-e314	-e371	-e894	-e760	-e316	e081	e040	e031	e077	e064	e081	12	-e076
.50	-e297	-e278	-e348	-e775	-e786	-e312	e073	e035	e032	e070	e039	e057	13	-e093
.55	-e253	-e235	-e271	-e614	-e818		e059	e012	e021	e089	e038			
.60	-e226	-e188	-e198	-e464	-e809	-e294	e043	e-014	e047	e027	e009	e033		
.65	-e172	-e144	-e151	-e296	-e774		e024	e-059	e054	-e010	-e024			
.68												-e048		
.70	-e134	-e091	-e088	-e216	-e742	-e277	-e006	-e031	e016	-e106				-e005
.73														
.75	-e075	-e026	-e025		-e674		-e010	e013	-e028					-e053
.77					-e108									
.79					-e114									
.80	-e017	e048			-e587	-e257	-e026	-e045						-e048
.82					-e062									
.83					-e046	-e080								
.85	e056	e181			-e481		-e119	-e275						
.87		e199	e036	-e652				-e735	-e178	-e183				-e053
.88		e170						-e453						
.89														
.90	e118	e130	e035	-e023	-e378	-e241	-e361							
.91														
.92	e091	e103	e035				-e237							
.94	e074						-e171	-e121	-e053					
.95	e061	e079			-e001	-e265	-e105							
.97	e033	e055	e035											
.99									-e024					

TABLE IV.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

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TABLE IV.- Continued

x c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
	-.01	-.011	-.012	-.125	-.076	-.069	-.055	-.032	-.076	-.029	-.075	-.122	-.150	
.03	.071	-.012	-.125	-.076	-.069	-.055	-.032	.046	.041	-.0117	-.117	-.136	-.211	
.05	.046	-.011	-.125	-.076	-.069	-.055	-.032	.034	.054	-.097	-.127	-.159		
.08	.029	-.028	-.080	-.080	-.080	-.060	-.057	.019	-.069	-.125	-.159	-.194		
.10	.013	-.055	-.087	-.087	-.087	-.060	-.057	.030	-.007	-.114	-.128	-.136	-.120	
.15	.027	-.086	-.087	-.087	-.087	-.057	-.057	.061	-.165	-.153	-.141	-.163	-.149	9
.20	-.019	-.034	-.065	-.082	-.064	-.038	-.024	.083	-.182	-.143	-.157	-.180	-.165	10
.25	-.053	-.076	-.077	-.053	-.049	-.033	-.033	.088	-.173	-.153	-.197	-.177	-.245	11
.30	-.072	-.103	-.092	-.062	-.053	-.038	-.038	.139	-.175	-.175	-.216	-.167	-.246	12
.35	-.049	-.114	-.084	-.074	-.038	-.038	-.058	.130	-.187	-.219	-.290	-.161	-.254	13
.40	-.081	-.093	-.080	-.055	-.021	-.069	-.069	.177	-.184	-.266	-.093	-.151	-.018	
.45	-.082	-.098	-.072	-.046	-.020	-.080	-.080	.173	-.209	-.220	-.116	-.143	-.248	
.50	-.073	-.093	-.053	-.011	-.002	-.075	-.075	.165	-.260	-.068	-.121	-.148	-.194	
.60	-.079	-.076	-.004	.015	.020	-.075	-.075	.180	-.236	-.083	-.155	-.276	-.225	
.65	-.054	-.053	.028	.078	.015	-.075	-.075	.152	-.079	-.072	-.628	-.579	-.313	
.68	-.029	-.008	.085	.130	.011	-.063	-.063	.111	-.090	-.762	-.360	-.123		
.73	-.002	.052	.178	-.001	-.001	-.001	-.001	.154	-.239	-.743	-.555	-.334		
.77	-.021	-.044	-.044	.215	-.044	-.044	-.044	.174	-.740	-.729	-.416	-.721		
.80	.064	.129	-.025	.069	-.013	-.023	-.023	.150	-.587	-.587	-.385	-.154	-.035	
.82	-.057	-.025	-.025	.211	-.025	-.025	-.025	.151	-.451	-.158	-.260	-.149	-.152	
.83	-.263	-.017	-.017	.032	-.009	-.009	-.009	.152	-.451	-.158	-.211	-.149	-.152	
.85	-.243	-.016	-.016	.032	-.009	-.009	-.009	.153	-.451	-.158	-.211	-.149	-.152	
.87	-.016	-.029	-.029	.051	-.015	-.015	-.015	.154	-.451	-.158	-.211	-.149	-.152	
.88	-.016	-.029	-.029	.051	-.015	-.015	-.015	.155	-.451	-.158	-.211	-.149	-.152	
.89	-.016	-.029	-.029	.051	-.015	-.015	-.015	.156	-.451	-.158	-.211	-.149	-.152	
.90	-.0205	-.0175	-.0175	.033	-.001	-.001	-.001	.157	-.451	-.158	-.211	-.149	-.152	
.91	-.0175	-.0175	-.0175	.033	-.001	-.001	-.001	.158	-.451	-.158	-.211	-.149	-.152	
.92	-.0147	-.0125	-.0125	.085	-.015	-.015	-.015	.159	-.451	-.158	-.211	-.149	-.152	
.93	-.0109	-.0084	-.0084	.032	-.015	-.015	-.015	.160	-.451	-.158	-.211	-.149	-.152	
.94	-.0073	-.0029	-.0029	.051	-.015	-.015	-.015	.161	-.451	-.158	-.211	-.149	-.152	
.95	-.0016	-.0029	-.0029	.051	-.015	-.015	-.015	.162	-.451	-.158	-.211	-.149	-.152	
.96	-.0016	-.0029	-.0029	.051	-.015	-.015	-.015	.163	-.451	-.158	-.211	-.149	-.152	
.97	-.0016	-.0029	-.0029	.051	-.015	-.015	-.015	.164	-.451	-.158	-.211	-.149	-.152	
.98	-.0016	-.0029	-.0029	.051	-.015	-.015	-.015	.165	-.451	-.158	-.211	-.149	-.152	

TABLE IV. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$\delta = -7.5$						$M = 0.90$	$\alpha = 4.9$	$P_{t,i}/P_{\infty} = 1.0$						
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	-0.065	-0.684	-0.681	-0.716	-0.647	-0.600	-0.379	-0.275	-0.323	-0.195	-0.187	-0.189		
.03	-0.114	-0.681	-0.659	-0.651	-0.651	-0.600	-0.379	-0.254	-0.287	-0.195	-0.143	-0.147	-0.179	
.05	-0.127	-0.598	-0.612	-0.612	-0.612	-0.600	-0.379	-0.210	-0.260	-0.143	-0.147	-0.179		
.08	-0.131	-0.171	-0.172	-0.742	-0.651	-0.600	-0.379	-0.203	-0.285	-0.117				
.10	-0.139	-0.245	-0.245	-0.681	-0.681	-0.600	-0.379	-0.191						
.15	-0.101	-0.265	-0.559	-0.614	-0.614	-0.600	-0.379	-0.159	-0.161	-0.064	-0.093	-0.048		
.20	-0.169	-0.265	-0.559	-0.614	-0.614	-0.600	-0.379	-0.184	-0.223	-0.043	-0.017	-0.033	-0.084	
.25	-0.189	-0.171	-0.154	-0.533	-0.595	-0.396	-0.379	-0.104	-0.036	-0.024	-0.007	-0.055	-0.055	9 -0.387
.30	-0.251	-0.196	-0.172	-0.418	-0.587	-0.380	-0.379	-0.061	-0.068	-0.013	-0.038	-0.025	-0.018	10 -0.408
.35	-0.186	-0.224	-0.197	-0.319	-0.595	-0.364	-0.379	-0.022	-0.069	-0.041	-0.083	-0.032	-0.038	11 -0.350
.40	-0.152	-0.224	-0.202	-0.240	-0.493	-0.362	-0.379	-0.036	-0.083	-0.075	-0.102	-0.056	-0.043	12 -0.142
.45	-0.251	-0.204	-0.192	-0.167	-0.421	-0.356	-0.379	-0.037	-0.081	-0.132	-0.093	-0.061	-0.061	13 -0.153
.50	-0.192	-0.208	-0.169	-0.129	-0.340	-0.348	-0.379	-0.077	-0.099	-0.194	-0.043	-0.054		
.55	-0.179	-0.194	-0.134	-0.074	-0.237	-0.335	-0.379	-0.086	-0.134	-0.219	-0.065	-0.065	-0.064	
.60	-0.179	-0.197	-0.064	-0.034	-0.129	-0.335	-0.379	-0.092	-0.212	-0.056	-0.056	-0.056	-0.056	-0.141
.65	-0.136	-0.118	-0.020	-0.037	-0.072	-0.379	-0.379	-0.137	-0.339	-0.056	-0.118			-0.064
.68								-0.172	-0.136	-0.061				-0.207
.70	-0.100	-0.052	0.046	0.088	-0.025	-0.312	-0.379	-0.172	-0.136	-0.061				-0.215
.73								-0.127	-0.136	-0.061				-0.150
.75	-0.046	0.024	0.137					-0.123	-0.074					-0.065
.77								-0.123	-0.074					-0.744
.79								-0.123	-0.074					-0.702
.80	0.026	0.101						-0.123	-0.074					-0.301
.82								-0.123	-0.074					-0.076
.85	0.128	0.220	0.207	0.084	0.027			-0.142	-0.236	-0.720	-0.304	-0.129		
.87								-0.142	-0.236	-0.720	-0.304	-0.129		
.88								-0.142	-0.236	-0.720	-0.304	-0.129		
.89								-0.142	-0.236	-0.720	-0.304	-0.129		
.90	0.193							-0.142	-0.236	-0.720	-0.304	-0.129		
.91								-0.142	-0.236	-0.720	-0.304	-0.129		
.92	0.143							-0.142	-0.236	-0.720	-0.304	-0.129		
.93								-0.142	-0.236	-0.720	-0.304	-0.129		
.94	0.110							-0.142	-0.236	-0.720	-0.304	-0.129		
.95	0.088							-0.142	-0.236	-0.720	-0.304	-0.129		
.96	0.082							-0.142	-0.236	-0.720	-0.304	-0.129		
.97								-0.142	-0.236	-0.720	-0.304	-0.129		
.98	0.039	0.062		0.078	0.061			-0.142	-0.236	-0.720	-0.304	-0.129		
								-0.142	-0.236	-0.720	-0.304	-0.129		
$\delta = -7.5$						$M = 0.90$	$\alpha = 4.9$	$P_{t,i}/P_{\infty} = 3.0$				Nacelle		
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.065	-0.674	-0.671	-0.730	-0.651	-0.610	-0.399	-0.274	-0.323	-0.195	-0.189	-0.189		
.03	-0.116	-0.657	-0.654	-0.651	-0.651	-0.610	-0.399	-0.251	-0.290	-0.195	-0.189	-0.189		
.05	-0.130	-0.618	-0.615	-0.654	-0.654	-0.610	-0.399	-0.202	-0.262	-0.147	-0.141	-0.141	-0.185	
.08	-0.134	-0.174	-0.174	-0.744	-0.654	-0.610	-0.399	-0.200	-0.284	-0.118				
.10	-0.143	-0.247	-0.247	-0.683	-0.683	-0.611	-0.399	-0.188						
.15	-0.104	-0.282	-0.567	-0.624	-0.624	-0.604	-0.399	-0.195	-0.165	-0.064	-0.064	-0.064	-0.062	
.20	-0.169	-0.265	-0.163	-0.557	-0.507	-0.373	-0.399	-0.182						
.25	-0.196	-0.177	-0.163	-0.557	-0.507	-0.373	-0.399	-0.136	-0.008	-0.043	-0.027	-0.048	-0.093	
.30	-0.210	-0.202	-0.176	-0.443	-0.602	-0.375	-0.399	-0.101	-0.044	-0.022	-0.04	-0.016	-0.056	9 -0.227
.35	-0.191	-0.232	-0.203	-0.333	-0.574	-0.381	-0.399	-0.060	-0.073	-0.009	-0.024	-0.001	-0.014	10 -0.227
.40	-0.160	-0.235	-0.211	-0.254	-0.512	-0.378	-0.399	-0.014	-0.069	-0.038	-0.054	-0.027	-0.042	11 -0.076
.45	-0.207	-0.215	-0.200	-0.173	-0.444	-0.375	-0.399	-0.044	-0.078	-0.064	-0.060	-0.033	-0.052	12 -0.083
.50	-0.200	-0.218	-0.180	-0.132	-0.369	-0.370	-0.399	-0.038	-0.089	-0.118	-0.041	-0.044	-0.068	13 -0.064
.55	-0.189	-0.209	-0.142	-0.075	-0.263	-0.364	-0.399	-0.075	-0.099	-0.155	-0.001	-0.030		
.60	-0.191	-0.176	-0.071	-0.036	-0.163	-0.364	-0.399	-0.083	-0.128	-0.085	-0.332	-0.357	-0.084	
.65	-0.147	-0.130	-0.028	-0.035	-0.036	-0.399	-0.399	-0.087	-0.193	-0.015	-0.047	-0.091		
.68								-0.138						
.70	-0.108	-0.073	0.040	0.086	-0.035	-0.345	-0.399	-0.119	-0.186	-0.008	-0.107			-0.093
.73								-0.119	-0.186	-0.008	-0.107			-0.221
.75	-0.053	0.003	0.133		-0.009			-0.103	-0.039	-0.044				-0.244
.77								-0.103	-0.039	-0.044				
.79								-0.103	-0.039	-0.044				
.80	0.019	0.094						-0.103	-0.039	-0.044				
.82								-0.103	-0.039	-0.044				
.85	0.118	0.204	0.201	0.057	0.081			-0.111	-0.232	-0.718	-0.436			
.87								-0.111	-0.232	-0.718	-0.436			
.88								-0.111	-0.232	-0.718	-0.436			
.89								-0.111	-0.232	-0.718	-0.436			
.90	0.183		0.106	0.053	0.049	-0.287	-0.399	-0.561	-0.615	-0.661	-0.236	-0.117	-0.065	-0.096
.91								-0.561	-0.615	-0.661	-0.236	-0.117	-0.065	-0.096
.92	0.136							-0.533	-0.615	-0.661	-0.236	-0.117	-0.065	-0.096
.93								-0.533	-0.615	-0.661	-0.236	-0.117	-0.065	-0.096
.94	0.106							-0.533	-0.615	-0.661	-0.236	-0.117	-0.065	-0.096
.95	0.075							-0.533	-0.615	-0.661	-0.236	-0.117	-0.065	-0.096
.96								-0.533	-0.615	-0.661	-0.236	-0.117	-0.065	-0.096
.97								-0.533	-0.615	-0.661	-0.236	-0.117	-0.065	-0.096
.98	0.028	0.045						-0.533	-0.615	-0.661	-0.236	-0.117	-0.065	-0.096

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TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$\frac{x}{c}$	$\delta = -7.5 \quad M = 0.90 \quad \alpha = 4^\circ \quad p_{t,j}/p_{\infty} = 5.0$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.060	-0.663						
.03	-0.108	-0.679	-0.712	-0.639				
.05	-0.121	-0.597						
.08	-0.123	-0.169	-0.730					
.10	-0.133	-0.238						
.15	-0.090	-0.265						
.20	-0.166							
.25	-0.184	-0.171	-0.148	-0.537	-0.599	-0.369		
.30	-0.193	-0.187	-0.166	-0.423	-0.594	-0.372		
.35	-0.179	-0.217	-0.191	-0.317	-0.541	-0.378		
.40	-0.150	-0.227	-0.196	-0.243	-0.502	-0.375		
.45	-0.194	-0.199	-0.188	-0.167	-0.433	-0.373		
.50	-0.187	-0.207	-0.170	-0.125	-0.354	-0.370		
.55	-0.177	-0.197	-0.134	-0.076	-0.251			
.60	-0.176	-0.146	-0.065	-0.036	-0.150	-0.361		
.65	-0.136	-0.121	-0.018	-0.034	-0.078			
.68								
.70	-0.093	-0.057	-0.047	-0.087	-0.028	-0.345		
.73								
.75	-0.043	-0.010	-0.137		-0.005			
.77								
.80	-0.023	-0.095						
.82								
.83								
.85	-0.127	-0.211	-0.204	-0.058	-0.034			
.87								
.88								
.89								
.90	-0.192		-0.110		-0.081	-0.286		
.91								
.92	-0.140	-0.153		-0.053				
.93								
.94	-0.106	-0.115	-0.087					
.95								
.96	-0.078							
.97								
.98	-0.030	-0.043						

$\frac{x}{c}$	$\delta = -7.5 \quad M = 0.90 \quad \alpha = 5.0 \quad p_{t,j}/p_{\infty} = 6.0$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.052	-0.669						
.03	-0.053	-0.689	-0.722	-0.652				
.05	-0.144	-0.650		-0.696	-0.607	-0.392		
.08	-0.134	-0.183	-0.752					
.10	-0.131	-0.244		-0.687	-0.604			
.15	-0.112	-0.288		-0.574	-0.615			
.20	-0.215							
.25	-0.191	-0.140	-0.157	-0.562	-0.607	-0.363		
.30	-0.188	-0.203	-0.172	-0.447	-0.601	-0.365		
.35	-0.160	-0.223	-0.196	-0.339	-0.586	-0.366		
.40	-0.250	-0.226	-0.203	-0.254	-0.522	-0.365		
.45	-0.197	-0.203	-0.192	-0.174	-0.456	-0.359		
.50	-0.192	-0.207	-0.174	-0.151	-0.380	-0.355		
.55	-0.177	-0.192	-0.137	-0.075	-0.275			
.60	-0.185	-0.160	-0.067	-0.033	-0.165	-0.346		
.65	-0.142	-0.116	-0.021	-0.037	-0.077			
.68								
.70	-0.096	-0.056	-0.044	-0.089	-0.021	-0.336		
.73								
.75	-0.045	-0.017	-0.133		-0.003			
.77								
.79								
.80	-0.025	-0.103						
.82								
.83								
.85	-0.127	-0.220	-0.204	-0.070				
.87								
.88								
.90	-0.195		-0.113		-0.055	-0.287		
.91								
.92	-0.146		-0.124		-0.090			
.93								
.94	-0.111		-0.092		-0.069			
.95	-0.085		-0.074		-0.043			
.97	-0.034	-0.060						

TABLE IV.- Continued

TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X c	Wing upper surface stations, fraction of semispan						$\alpha = 10^\circ$	$P_{t,i}/P_{\infty} = 5.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.293	-0.954	-1.058	-0.970			.408	.445		
.03	-0.239	-0.958	-1.052	-0.988	-0.932	-0.416	.431	.454	.318	.266
.05	-0.248	-1.073					.400	.438	.265	.237
.08	-0.360	-1.235	-1.173				.356	.454	.275	.232
.10	-0.353	-1.269		-1.019	-0.932		.386			
.15	-0.287	-1.032	-1.204		-0.932		.377	.355	.220	.199
.20	-0.405						.352			
.25	-0.365	-0.166	-0.837	-1.098	-0.895	-0.392	.298	.160	.189	.161
.30	-0.442	-0.329	-0.680	-1.066	-0.901	-0.392	.227	.102	.159	.136
.35	-0.228	-0.405	-0.577	-1.086	-0.925		.223	.062	.118	.145
.40	-0.345	-0.431	-0.526	-1.073	-0.942	-0.385	.157	.046	.087	.088
.45	-0.354	-0.404	-0.461	-1.013	-0.915		.176	.043	.085	.079
.50	-0.383	-0.409	-0.403	-0.522	-0.926	-0.368	.075	.020	.020	.060
.55	-0.369	-0.396	-0.319	-0.756	-0.868		.059	.012	.007	.051
.60	-0.368	-0.327	-0.231	-0.566	-0.998	-0.349	.043	.016	.022	.059
.65	-0.310	-0.241	-0.164	-0.317	-1.021		.022	.067	.085	.011
.68							.022	.027	.027	.013
.70	-0.222	-0.147	-0.096	-0.191	-0.962	-0.330	-0.007	-0.041	.075	-0.018
.73										
.75	-0.117	-0.062	-0.027		-0.860		-0.002	.048	.040	
.77										
.79										
.80	-0.040	0.025								
.82										
.83										
.85	-0.040	0.160	0.035	0.028						
.87										
.88										
.89										
.90	-0.119		0.021	0.007	-0.381	-0.308	-0.595			
.91										
.92	-0.095						-0.509			
.93										
.94	-0.078									
.95	-0.076									
.96	-0.064									
.97										
.98	-0.033	0.053								
X c	Wing upper surface stations, fraction of semispan						$\alpha = 10^\circ$	$P_{t,i}/P_{\infty} = 6.2$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.200	-0.959	-1.068	-0.991			.400	.437		
.03	-0.247	-1.003	-1.068	-1.010	-0.952	-0.433	.424	.447	.315	.257
.05	-0.335	-0.02					.396	.430	.256	.242
.08	-0.346	-1.241	-1.185				.390	.448	.267	.224
.10	-0.367	-1.294		-1.035	-0.952		.378		.229	
.15	-0.294	-1.033		-0.218	-0.952		.373	.347	.212	.191
.20	-0.407						.347			
.25	-0.355	-0.174	-0.845	-1.118	-0.916	-0.408	.280	.157	.184	.160
.30	-0.456	-0.334	-0.691	-1.116	-0.922	-0.380	.222	.096	.156	.139
.35	-0.236	-0.407	-0.586	-1.102	-0.938	-0.406	.218	.058	.112	.107
.40	-0.355	-0.436	-0.537	-1.093	-0.933	-0.399	.149	.045	.083	.083
.45	-0.362	-0.410	-0.479	-1.037	-0.931	-0.390	.068	.036	.060	.066
.50	-0.391	-0.414	-0.417	-0.943	-0.945	-0.384	.067	.023	.017	.072
.55	-0.378	-0.392	-0.340	-0.782	-0.978		.051	.007	.004	.037
.60	-0.378	-0.340	-0.241	-0.582	-1.018	-0.362	.040	.015	.050	.034
.65	-0.320	-0.251	-0.173	-0.333	-1.039		.021	.063	.067	.007
.68										
.70	-0.237	-0.152	-0.104	-0.201	-0.983	-0.343	-0.010	-0.025	.013	-0.027
.73										
.75	-0.130	-0.073	-0.037		-0.885		-0.006	.040	-0.004	
.77										
.79										
.80	-0.049	0.016								
.82										
.83										
.85	-0.032	0.152	0.024	0.040						
.87										
.88										
.89										
.90	-0.107		0.009		-0.399	-0.322				
.91										
.92	-0.085									
.93										
.94	-0.067									
.95										
.96	-0.052									
.97										
.98	-0.019	0.043								

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TABLE IV. - Continued

TABLE IV. - Continued

TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

δ = -7.5 M = 1.00						α = 5.0 Pt _{t,j} /P _∞ = 1.0						Nacelle		
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.001	-0.532					+0.278	+0.400						
.03	-0.011	-0.542	-0.634	-0.556			+0.284	+0.365	+0.240	+0.240	+0.228			
.05	-0.059	-0.493					+0.259	+0.345		+0.195	+0.185	+0.141		
.08	-0.056	-0.106	-0.664		-0.573	-0.619	+0.273	+0.366	+0.168					
.10	-0.054	-0.109					+0.270			+0.159				
.15	-0.021	-0.258	-0.525		-0.580	-0.578	+0.286	+0.241	+0.112		+0.095			
.20	-0.079						+0.279							
.25	-0.080	-0.151	-0.104	-0.530	-0.613	-0.608	+0.223	+0.081	+0.109	+0.078	+0.071	+0.045		
.30	-0.196	-0.117	-0.125	-0.467	-0.627	-0.616	+0.161	+0.021	+0.086	+0.052	+0.017	+0.022	9	-0.297
.35	-0.141	-0.158	-0.168	-0.417	-0.637	-0.626	+0.155	+0.014	+0.050	+0.021	+0.015	+0.031	10	-0.326
.40	-0.137	-0.171	-0.193	-0.370	-0.610	-0.630	+0.164	+0.014	+0.023	+0.014	+0.061	+0.022	11	-0.246
.45	-0.130	-0.150	-0.211	-0.314	-0.587	-0.631	+0.110	+0.022	+0.023	+0.056	+0.032	+0.025	12	-0.243
.50	-0.134	-0.170	-0.232	-0.295	-0.566	-0.631	+0.119	+0.013	+0.053	+0.085	+0.122	+0.441	13	-0.239
.55	-0.134	-0.188	-0.241	-0.261	-0.509		+0.004	+0.015	+0.110	+0.102	+0.139			
.60	-0.164	-0.199	-0.223	-0.258	-0.425	-0.617	+0.006	+0.044	+0.197	+0.181	+0.141	+0.059		
.65	-0.170	-0.205	-0.244	-0.229	-0.322		+0.008	+0.107	+0.303	+0.176	+0.046			
.68												+0.052		
.70	-0.185	-0.210	-0.200	-0.221	-0.243	-0.589	+0.151	+0.238	+0.137	+0.089		+0.101		
.73												+0.113		
.75	-0.199	-0.187	+0.054			+0.186	+0.134	+0.068	+0.027			+0.142		
.77						+0.159						+0.391		
.79						+0.094						+0.391		
.80	-0.137	+0.003				+0.144	-0.539	+0.263	+0.122			+0.181	+0.148	
.82						+0.231						+0.538		
.83						+0.198	+0.038					+0.518	+0.361	
.85	+0.059	+0.185				+0.111	+0.117					+0.508		
.87						+0.250	+0.126					+0.522	+0.499	+0.318
.88												+0.506		
.89												+0.424		
.90	+0.164					+0.077	+0.032	+0.085	+0.481	+0.462	+0.400	+0.313	+0.210	+0.201
.91												+0.430	+0.381	+0.294
.92	+0.122													
.93						+0.099	+0.035					+0.394	+0.359	+0.219
.94	+0.085											+0.358		
.95						+0.052		+0.058					+0.215	+0.295
.97						+0.002						+0.286		
.98	-0.019	+0.005												
δ = -7.5 M = 1.00						α = 4.9 Pt _{t,j} /P _∞ = 3.0						Nacelle		
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	+0.001	-0.529					+0.278	+0.398						
.03	-0.010	-0.536	-0.626	-0.554			+0.283	+0.363	+0.240	+0.236	+0.225			
.05	-0.055	-0.468					+0.256	+0.343		+0.193	+0.181	+0.179		
.08	-0.054	-0.079	-0.653				+0.268	+0.365	+0.166					
.10	-0.053	-0.106			-0.575	-0.576	+0.284	+0.238	+0.114		+0.096			
.15	-0.020	-0.251	-0.513				+0.277							
.20	-0.077						+0.222	+0.083	+0.106	+0.075	+0.070	+0.127		
.25	-0.080	-0.096	-0.121	-0.511	-0.609	-0.603	+0.160	+0.024	+0.077	+0.053	+0.020	+0.111	9	-0.291
.30	-0.191	-0.123	-0.126	-0.446	-0.623	-0.615	+0.155	+0.013	+0.048	+0.020	+0.014	+0.072	10	-0.317
.35	-0.133	-0.151	-0.164	-0.394	-0.610	-0.627	+0.062	+0.010	+0.022	+0.017	+0.011	+0.133	11	-0.127
.40	-0.134	-0.165	-0.193	-0.350	-0.600	-0.633	+0.012	+0.016	+0.002	+0.055	+0.089	+0.000	12	-0.136
.45	-0.127	-0.147	-0.209	-0.298	-0.573	-0.633	+0.021	+0.024	+0.056	+0.084	+0.022	+0.023	13	-0.113
.50	-0.132	-0.166	-0.230	-0.281	-0.548	-0.633	+0.003	+0.024	+0.107	+0.102	+0.031			
.55	-0.133	-0.180	-0.239	-0.259	-0.492		+0.006	+0.049	+0.193	+0.097	+0.024	+0.047		
.60	-0.160	-0.194	-0.220	-0.259	-0.398	-0.618	+0.011	+0.105	+0.086	+0.041	+0.010			
.65	-0.167	-0.201	-0.240	-0.225	-0.285		+0.053	+0.232	+0.081	+0.001		+0.029		
.68							+0.153	+0.183	+0.164					
.70	-0.182	-0.206	-0.198	-0.218	-0.239	-0.587	+0.128	+0.040	+0.055			+0.089		
.73							+0.156					+0.112		
.75	-0.195	-0.183	+0.059			+0.164	+0.128					+0.142		
.77						+0.094						+0.377		
.79						+0.124						+0.377		
.80	-0.134	+0.067				+0.125	-0.528	+0.081	+0.069			+0.181	+0.140	
.82						+0.234						+0.538		
.83						+0.200	+0.044	+0.102	+0.089	+0.052	+0.360			
.85	+0.060	+0.185				+0.084	+0.470	+0.401	+0.448	+0.441	+0.209	+0.192		
.87						+0.220	+0.132	+0.009		+0.440	+0.318			
.88						+0.191				+0.427	+0.343			
.89						+0.166	+0.081	+0.084	+0.470	+0.448	+0.399			
.91						+0.146	+0.025			+0.440	+0.318			
.92						+0.124	+0.100	+0.033		+0.427	+0.343			
.93						+0.087				+0.399				
.94						+0.052				+0.424		+0.219		
.95						+0.048				+0.373		+0.322		
.97						+0.023	+0.006	+0.011		+0.359				

CONTINUATION

TABLE IV.-- Continued

TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

		$\delta = -7.5$						$M = 1.00$						$\alpha = 10.0$								
x	c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle						Orifice number	C_p	
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			
.01	-149	-1746						+16	+24													
.03	-123	-1779	-949	-806				+475	+526	+371	+312	+274										
.05	-267	-879						+469	+507	+308	+273	+237										
.08	-236	-971	-1.006					+470	+523	+315												
.10	-222	-989						+462														
.15	-159	-842	-1.086					+455	+420	+265												
.20	-236							+431														
.25	-189	-111	-657	-554	-847	-721		+368	+218	+239	+196	+173	+204									
.30	-292	-212	-458	-360	-873	-726		+285	+157	+210	+162	+133	+185	9	-244							
.35	-302	-290	-386	-367	-908	-731		+279	+111	+164	+126	+120	+134	10	-261							
.40	-228	-315	-379	-376	-924	-728		+214	+194	+133	+104	+103	+090	11	-031							
.45	-237	-297	-369	-360	-929	-725		+123	+092	+099	+044	+099	+078	12	-020							
.50	-248	-305	-377	-376	-922	-935	-721	+125	+093	+042	+026	+090	+053	13	-027							
.55	-270	-309	-381	-384	-926			+117	+070	+022	+109	+098										
.60	-283	-315	-357	-744	-852	-711		+095	+030	+113	+094	+077	+023									
.65	-295	-310	-363	-570	-836			+065	+039	+079	+089	+041										
.68																						
.70	-315	-318	-291	-413	-875	-656		+015	+187	+106	+045											
.73																						
.75	-305	-304	-155					+004	+089		+080	+008	+116									
.77								+027														
.80	-269	-207						+354	+610		+019	+065										
.82								+009														
.83								+022	+060													
.85	-107	+066						+775														
.86	+136							+105														
.88	+076																					
.89																						
.90	+014	+034						+675	+563		+398	+437										
.91								+127			+377											
.92	+001							+069			+393											
.93																						
.94	-020																					
.95	-027																					
.96	-040																					
.97	-081																					
.98	-081	-047																				
		$\delta = -7.5$						$M = 1.00$						$\alpha = 10.0$								
x	c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle						Orifice number	C_p	
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			
.01	-147	-1749						+41C	+527													
.03	-127	-1784	-941	-802	-816	-784	-754	+48C	+530	+376	+321	+287										
.05	-268	-878						+472	+512		+311	+289	+245									
.08	-239	-972	-1.002					+473	+528	+324												
.10	-222	-984						+465			+280											
.15	-158	-846	-1.091					+459	+427	+268												
.20	-228							+436														
.25	-188	-108	-662	-549	-836	-716		+372	+226	+242	+207	+212	+214									
.30	-207	-212	-469	-394	-862	-720		+298	+165	+217	+180	+181	+197	9	-192							
.35	-303	-283	-390	-395	-899	-727		+282	+122	+170	+145	+168	+143	10	-091							
.40	-218	-308	-381	-397	-914	-726		+219	+110	+140	+115	+145	+198	11	-068							
.45	-228	-289	-370	-548	-921	-723		+130	+108	+113	+109	+136	+085	12	-066							
.50	-258	-295	-375	-914	-925	-721		+133	+058	+057	+131	+118	+059	13	-084							
.55	-259	-298	-378	-839	-918			+125	+078	+007	+170	+121										
.60	-270	-303	-353	-740	-855	-716		+102	+045	+080	+136	+092	+028									
.65	-281	-298	-353	-565	-834			+077	+020	+156	+117	+049										
.68																						
.70	-301	-303	-283	-414	-876	-577		+035	+103	+176	+058											
.73									+006	+121	+131											
.75	-293	-289	-156					+024	+892													
.77								+046														
.79								+072	+780													
.80	-254	-193						+056	+616		+089	+145										
.82								+014														
.83								+050	+072		+061	+064	+310									
.85	-099	+077						+047	+114		+061	+500	+489	+310								
.87								+124			+050	+499	+489	+377								
.88								+086														
.89																						
.90	+016							+066	+137		+084	+548	+407	+453								
.91	+039							+002	+087		+0412	+384	+292									
.92	-002							+022			+0368	+397	+325	+337								
.93								+022	+106		+0156	+574	+325	+337								
.94	-022							+027			+0325	+397	+325	+337								
.95								+046			+0106	+156	+325	+337								
.96	-046							+027			+0325	+397	+325	+337								
.97								+094	+059		+0365	+325	+325	+337								
.98																						

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TABLE IV. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C _p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.80			
.01	-147	-746	-948	-807	-822	-791	-767	-426	-528	-374	-318	-288			
.03	-121	-778	-948	-807	-822	-791	-767	-478	-530	-314	-291	-244			
.05	-265	-880	-948	-807	-822	-791	-767	-474	-511	-314	-291	-244			
.08	-233	-974	-1.004	-852	-806	-834	-737	-474	-527	-321	-281	-235			
.10	-219	-992	-1.004	-852	-806	-834	-737	-469	-422	-271	-235				
.15	-157	-852	-1.091	-834	-834	-834	-737	-459	-422	-271	-235				
.20	-235	-	-	-	-	-	-737	-473	-422	-243	-211	-211			
.25	-188	-107	-666	-955	-847	-728	-737	-490	-422	-123	-184	-196	9	-0.050	
.30	-294	-220	-472	-961	-873	-733	-737	-490	-512	-123	-190	-196	10	-0.021	
.35	-302	-288	-490	-968	-907	-739	-737	-485	-512	-167	-156	-176	11	-0.136	
.40	-231	-315	-385	-976	-924	-736	-737	-421	-512	-107	-148	-130	12	-0.138	
.45	-235	-298	-374	-957	-929	-759	-737	-421	-512	-120	-127	-139	13	-0.057	
.50	-266	-307	-381	-924	-934	-734	-737	-434	-502	-67	-144	-122			
.55	-272	-311	-383	-847	-925	-732	-737	-425	-502	-109	-142	-124			
.60	-283	-314	-360	-751	-863	-732	-737	-400	-466	-109	-142	-094			
.65	-295	-310	-359	-581	-885	-732	-737	-378	-411	-166	-123	-048			
.70	-314	-317	-295	-425	-382	-703	-737	-404	-409	-176	-162	-001			
.73	-	-	-	-	-	-	-737	-	-	-	-	-283			
.75	-304	-303	-162	-	-	-901	-737	-404	-135	-143	-	-106			
.77	-	-	-	-	-	-	-737	-	-	-	-	-344			
.79	-	-	-	-	-	-	-737	-	-	-	-	-344			
.80	-267	-205	-	-	-	-868	-629	-099	-157	-	-	-136	-061		
.82	-	-	-	-017	-	-	-	-	-	-518	-	-			
.83	-	-	-	-033	-	-068	-	-	-	-509	-	-313			
.85	-107	.063	-	-	-	-790	-	-071	-	-055	-	-147			
.87	-	.103	-	-058	-	-114	-	-	-	-504	-	-398			
.88	-	.073	-	-	-	-	-	-	-	-520	-	-			
.89	-	-	-	-	-	-	-	-	-	-393	-	-			
.90	.010	-	-071	-	-141	-700	-552	-436	-	-549	-	-174	-124		
.91	-	.029	-	-	-	-	-	-	-	-542	-	-434			
.92	-	-005	-	-007	-	-093	-	-	-	-414	-	-492	-313		
.93	-	-027	-	-	-	-	-	-	-	-433	-	-363	-204		
.95	-	-039	-	-	-	-591	-	-	-	-453	-	-382			
.96	-	-052	-	-114	-	-	-	-	-	-453	-	-348			
.97	-	-099	-	-074	-	-	-	-	-	-453	-	-348			
 $\delta = -7.5 \quad M = 1.00 \quad \alpha = 10.0 \quad P_{t_1}/P_{\infty} = 6.8$															
X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C _p	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.80			
.01	-141	-741	-947	-811	-824	-792	-771	-435	-528	-373	-321	-286			
.03	-142	-774	-947	-811	-824	-792	-771	-477	-531	-373	-311	-289	-236		
.05	-228	-674	-	-	-	-	-	-474	-512	-322	-311	-289			
.08	-235	-670	-1.007	-	-	-	-	-474	-524	-322	-279	-			
.10	-209	-690	-	-	-851	-812	-	-466	-422	-269	-	-233			
.15	-157	-846	-1.093	-	-	-834	-	-457	-422	-269	-	-			
.20	-232	-	-	-	-	-	-	-434	-	-	-	-			
.25	-185	-104	-666	-954	-848	-730	-737	-472	-422	-240	-210	-217	205		
.30	-289	-421	-446	-952	-877	-733	-737	-491	-511	-213	-179	-184	188	9	-0.022
.35	-299	-428	-398	-969	-909	-737	-737	-484	-512	-120	-176	-154	168	10	-0.018
.40	-232	-313	-386	-976	-924	-735	-737	-421	-512	-106	-146	-139	137	11	-173
.45	-236	-299	-376	-958	-931	-732	-737	-431	-512	-104	-121	-131	119	12	-238
.50	-265	-308	-382	-925	-939	-729	-737	-435	-509	-70	-140	-140	096	13	-219
.55	-270	-311	-384	-849	-931	-721	-737	-425	-75	-125	-175	-158	096		
.60	-281	-316	-361	-751	-865	-721	-737	-402	-47	-126	-103	-103	070	-026	
.65	-293	-314	-361	-577	-540	-	-	-080	-005	-157	-063	-036			
.70	-314	-319	-295	-428	-884	-678	-	-050	-039	-076	-018	-	-013		
.73	-	-	-	-	-	-	-	-061	-135	-046	-	-086			
.75	-304	-304	-164	-	-	-900	-	-	-	-352	-	-108			
.77	-	-	-	-	-	-	-	-	-	-352	-	-			
.79	-	-	-	-	-	-	-	-	-	-352	-	-			
.80	-267	-204	-	-018	-	-665	-619	-074	-	-031	-	-480		-131	-054
.82	-	-	-	-033	-	-064	-	-	-	-438	-	-300			
.83	-	-	-	-061	-	-789	-	-030	-	-089	-	-133			
.85	-105	-	-	-054	-	-115	-	-	-	-409	-	-267	-342		
.87	-	.071	-	-	-	-	-	-	-	-276	-	-			
.88	-	-	-	-	-	-	-	-	-	-430	-	-			
.90	.011	-	-070	-	-141	-692	-555	-424	-	-292	-	-143	-112		
.91	-	.025	-	-	-	-	-	-	-	-263	-	-293			
.92	-	-005	-	-012	-	-093	-	-	-	-286	-	-363	-332		
.93	-	-	-	-012	-	-093	-	-	-	-265	-	-419	-		
.94	-	-027	-	-043	-	-164	-	-	-	-244	-	-278	-156		
.95	-	-055	-	-124	-	-	-	-	-	-421	-	-395			
.97	-	-109	-	-680	-	-	-	-	-	-	-	-			

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TABLE IV.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X c	δ = -7.5 M = 1.05						α = -C ₁	P _{t,j} /P _∞ = 1.0	Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90				
Wing upper surface stations, fraction of semispan										
.01	.095	.061					.085	.061		
.03	.064	.036	-120	-0.92			.057	.068	-0.98	
.05	.033	.014					.026	.098	-119	-141
.08	.022	.049	-0.85		-100	-121	.013	.142	-0.99	
.10	.006	.046			-111	-105	.015		-125	
.15	.026	-0.96	-0.90			-108	.027	.020	-123	-162
.20	.034						.102			
.25	.035	-0.62	-115	-106	-110	-160	.081	-0.76	-134	-146
.30	.033	-0.71	-105	-102	-128	-184	.018	-125	-149	-126
.35	.064	-0.85	-114	-121	-154	-198	.105	-158	-180	-129
.40	.036	-124	-124	-144	-155	-207	.046	-198	-144	-156
.45	.086	-105	-137	-149	-153	-216	.157	-183	-142	-174
.50	.092	-114	-150	-165	-174	-226	.143	-286	-180	-176
.55	.087	-124	-159	-165	-187		.190	-175	-205	-175
.60	.110	-135	-134	-161	-181	-187	.166	-164	-239	-232
.65	.106	-135	-151	-153	-111		.147	.200	-356	-266
.68										-257
.70	-116	-141	-152	.059	.016	-105	-141	-280	-375	-311
.73							-181	-430	-266	-303
.75	-128	-128	-130							-310
.77										-610
.79										-537
.80	-0.89	.045					-292	-298		-337
.82										-297
.83										-531
.85	.118	.262	.297	.121			-352	-213	-515	-503
.87										-379
.88										-538
.89										-528
.90	.218		.162							-494
.91										-481
.92	.181	.220		.029						-495
.93										-376
.94										-314
.95										-422
.96	.108	.121								-437
.97										-419
.98										-417
										-408
										-375
										-405
										-346
										-357
δ = -7.5 M = 1.05										
X c	Wing upper surface stations, fraction of semispan						α = -C ₁	P _{t,j} /P _∞ = 3.0	Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90				
Wing lower surface stations, fraction of semispan										
.01	.085	.040					.079	.022		
.03	.057	.033	-124	-0.90	-0.98	-114	.053	.064	-0.96	-130
.05	.024	.014					.015	.097		
.08	.015	.056	-0.83				.008	.138	-0.99	
.10	.001	.046			-111	-104	.009		-117	
.15	.019	-0.99	-0.87			-106	.035	.020	-117	-160
.20	.031						.100			
.25	.030	-0.63	-112	-103	-113	-160	.077	-0.83	-126	-148
.30	.045	-0.77	-100	-0.99	-121	-181	.013	-127	-138	-126
.35	.063	-0.85	-111	-117	-153	-193	.107	-161	-161	-157
.40	.037	-125	-121	-141	-153	-201	.044	-194	-141	-154
.45	.088	-102	-132	-147	-151	-207	.163	-176	-140	-172
.50	.094	-115	-148	-161	-171	-214	.149	-178	-175	-173
.55	.088	-125	-157	-161	-183		.183	-175	-200	-174
.60	.112	-134	-131	-163	-179	-173	.161	-172	-240	-232
.65	.107	-135	-149	-150	-084		.145	-204	-356	-266
.68										-256
.70	-115	-143	-148	.075	.023	-065	-143	-280	-204	-306
.73							-178	-393	-379	-300
.75	-127	-127	-142							-437
.77										-410
.79										-249
.80	-0.85	.052					-290	-079		-297
.82										-515
.83										-507
.85	.120	.266	.336							-402
.87										-256
.88										-508
.89										-387
.90	.225		.163	.032	-0.93	-0.61				-417
.91										-504
.92	.180		.114							-389
.93										-458
.94	.151		.120							-438
.95										-410
.96	.110									-442
.97										-373
.98	.032	.059								-427

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TABLE IV. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$\frac{x}{c}$	$\delta = -7.5 \quad M = 1.05 \quad \alpha = -0.1 \quad p_{t,y}/p_{\infty} = 5:1$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.087	.050					.084	.066
.03	.060	.039	-.117	-.090			.056	.059
.05	.037	.043					.019	.100
.08	.02C	.051	-.080				.010	.139
.10	.001	.042					-.005	
.15	.025	-.064	-.086				.247	.019
.20	.034						.103	
.25	.033	-.063	-.109	-.102	-.114	-.161	.078	-.076
.30	-.038	.076	-.098	-.097	-.131	-.179	.013	-.126
.35	-.059	-.083	-.109	-.116	-.151	-.192	-.100	-.155
.40	-.033	-.122	-.120	-.139	-.153	-.198	-.048	-.262
.45	-.085	-.105	-.131	-.145	-.153	-.204	-.149	-.174
.50	-.089	-.114	-.145	-.145	-.161	-.170	-.211	-.139
.55	-.083	-.126	-.155	-.161	-.181		-.171	-.173
.60	-.108	-.133	-.129	-.160	-.179	-.168	-.156	-.167
.65	-.102	-.134	-.146	-.147	-.075		-.141	-.203
.68								-.349
.70	-.111	-.141	-.146	.081	.026	-.053	-.141	-.278
.73								-.074
.75	-.125	-.127	-.146				-.173	-.183
.77								-.157
.79								-.404
.80	-.079	.060						-.440
.82								
.83								
.85	-.125	.269						
.87								
.88								
.89								
.90	.225							
.91								
.92	.165							
.93								
.94	.149							
.96								
.96	.119							
.97	.107							
.98	.024	.059						
$\frac{x}{c}$	$\delta = -7.5 \quad M = 1.05 \quad \alpha = -0.1 \quad p_{t,y}/p_{\infty} = 7:1$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.088	.048					.088	.070
.03	.065	.039	-.117	-.093			.064	.076
.05	.038	.048					.024	.138
.08	.019	.056	-.083	-.102	-.120		.014	.142
.10	.004	.048					.004	.143
.15	.024	-.057	-.085	-.110	-.105		.038	.020
.20	.040						.107	
.25	.032	-.056	-.110	-.103	-.117	-.164	.082	-.069
.30	-.039	.068	-.096	-.098	-.132	-.181	.017	-.115
.35	-.054	-.079	-.119	-.115	-.155	-.192	-.043	-.147
.40	-.031	-.117	-.120	-.135	-.156	-.201	-.041	-.186
.45	-.055	-.102	-.131	-.135	-.155	-.206	-.149	-.143
.50	-.086	-.107	-.146	-.163	-.172	-.213	-.135	-.166
.55	-.083	-.117	-.155	-.163	-.186		-.167	-.166
.60	-.105	-.126	-.131	-.160	-.182	-.170	-.151	-.161
.65	-.099	-.126	-.148	-.150	-.077		-.136	-.191
.68								-.180
.70	-.108	-.132	-.146	.078	.024	-.055	-.137	-.257
.73								-.032
.75	-.123	-.120	-.144				-.168	-.033
.77								-.144
.79								-.391
.80	-.082	.066					-.138	-.144
.82								-.434
.83								
.85	.130	.270						
.87								
.88								
.89								
.90	.225							
.91								
.92	.181							
.93								
.94	.146							
.95								
.96	.108							
.97								
.98	.027	.056						

TABLE IV. - Continued

TABLE IV. - Continued

X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-.031	-.593					+.231	+.362						
.03	-.040	-.558	-.648	-.601			+.229	+.330	+.196	+.192	+.201			
.05	-.114	-.360		-.607	-.459	-.622	+.184	+.320		+.154	+.159	+.150		
.08	-.110	-.052	-.692				+.191	+.353	+.124					
.10	-.119	-.114		-.622	-.595		+.194			+.109				
.15	-.071	-.203	-.497		-.616		+.245	+.225	+.052		+.080			
.20	-.096						+.261							
.25	-.065	-.189	-.180	-.511	-.617	-.623	+.211	+.051	+.075	+.061	+.062	+.039		
.30	-.146	-.181	-.190	-.422	-.622	-.631	+.130	-.016	+.036	+.044	+.022	+.018		
.35	-.202	-.214	-.228	-.339	-.616	-.642	+.121	-.072	+.012	+.018	-.007	+.006	10	-.261
.40	-.233	-.228	-.233	-.307	-.612	-.648	+.035	-.089	+.006	+.012	-.052	+.020	11	-.288
.45	-.181	-.206	-.241	-.274	-.535	-.648	-.072	-.065	-.008	-.039	-.070	+.006	12	-.045
.50	-.186	-.214	-.252	-.275	-.504	-.648	-.042	-.070	-.050	-.063	-.096	-.019	13	-.045
.55	-.180	-.222	-.258	-.266	-.447		-.053	-.052	-.090	-.071	-.110			
.60	-.201	-.230	-.237	-.271	-.408	-.639	-.037	-.057	-.156	-.142	-.046	-.023		
.65	-.203	-.229	-.255	-.260	-.345		-.026	-.098	-.196	-.016	-.012			
.68												-.028		
.70	-.215	-.234	-.255	-.247	-.253	-.620	-.052	-.197	+.016	-.031				
.73												-.070		
.75	-.221	-.226	-.107		-.187		-.109	-.069	+.024			-.090		
.77												-.302		
.79												-.302		
.80	-.211	-.155					-.146	-.569	-.177	-.028				
.82												-.429		
.83												-.424		
.85	-.028	-.132	-.173	-.008			-.029	-.047						
.87												-.413		
.88												-.440		
.89												-.260		
.90	+.137		+.082		-.034		-.112	-.519	-.331					
.91			+.131						-.364					
.92			+.108						-.345					
.93			+.094		+.038				-.478					
.94			+.080						-.366					
.95			+.052						-.411					
.96			+.051		-.061				-.386					
.97			-.014		-.004				-.292					
.98									-.350					

TABLE IV.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

		$\delta = -7.5 \quad M = 1.05 \quad \alpha = 10.0 \quad P_{t,j}/P_{\infty} = 1.0$													
X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle		
	0.10	0.32	0.53	0.65	0.74	0.90	0.10	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p	
.01	-.196	-.750					.327	.566							
.02	-.147	-.571	-.967	-.822			.385	.505	.338	.301	.276				
.05	-.298	-.679					.414	.588		.288	.272				
.08	-.272	-.634	-.995				.423	.509	.283						
.10	-.268	-.625					.423								
.20	-.227	-.760	-1.041	-.853	-.810		.431	.460	.232	.264	.207				
.25	-.195	-.330	-.732	-.533	-.847	-.748	.354	.177	.225	.190	.173	.208			
.30	-.264	-.263	-.471	-.544	-.875	-.749	.255	.114	.201	.164	.126	.192	9	-.208	
.35	-.355	-.337	-.383	-.549	-.907	-.751	.237	.068	.162	.129	.094	.144	10	-.222	
.40	-.301	-.337	-.366	-.595	-.919	-.746	.193	.066	.134	.092	.049	.107	11	-.133	
.45	-.264	-.342	-.362	-.562	-.923	-.741	.085	.074	.107	.053	.020	.099	12	-.126	
.50	-.296	-.336	-.372	-.540	-.929	-.737	.102	.085	.055	.020	.004	.076	13	-.128	
.55	-.297	-.332	-.378	-.871	-.938		.110	.076	.002	.003	.066				
.60	-.308	-.336	-.354	-.773	-.938	-.730	.091	.060	.076	-.053	.095	.054			
.65	-.339	-.328	-.371	-.625	-.943		.075	-.019	-.178	.072	.076				
.68												.037			
.70	-.322	-.330	-.364	-.531	-.862	-.681	.044		.145	.024	.072		.020		
.73	-.314	-.321	-.221				.040		.201	.121			.042		
.77							.001						.067		
.79							.010						.284		
.90	-.297	-.296					.001						.284		
.82							.018						.095	-.016	
.83							.017								
.85	-.156	.024					.013								
.87		.100	-.040	-.066			.013								
.99		.064					.013								
.89							.013								
.90	-.015						.013								
.91		.022					.013								
.92		.020					.013								
.73							.013								
.94	-.032						.013								
.95	-.037						.013								
.96	-.046						.013								
.97	-.078						.013								
.98	-.051						.013								
		$\delta = -7.5 \quad M = 1.05 \quad \alpha = 10.0 \quad P_{t,j}/P_{\infty} = 3.0$						Wing lower surface stations, fraction of semispan						Nacelle	
X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C _p	
	0.10	0.32	0.53	0.65	0.74	0.90	0.10	0.32	0.53	0.65	0.74	0.90			
.01	-.212	-.758	-.980	-.824	-.806	-.792	.313	.501							
.03	-.179	-.783					.386	.494	.337	.300	.271				
.05	-.299	-.891					.405	.482		.287	.268				
.08	-.282	-.945	-1.001				.416	.502	.282						
.10	-.277	-.941					.417								
.15	-.202	-.766	-1.050				.427	.392	.231		.234				
.20	-.255						.416								
.25	-.190	-.136	-.733	-.940	-.854	-.761	.348	.173	.218	.191	.172	.228			
.30	-.254	-.271	-.475	-.948	-.870	-.763	.248	.112	.190	.165	.129	.213	9	-.209	
.35	-.256	-.342	-.385	-.954	-.911	-.766	.220	.070	.156	.128	.121	.163	10	-.221	
.40	-.203	-.344	-.373	-.963	-.923	-.762	.189	.042	.129	.091	.126	.121	11	-.017	
.45	-.269	-.348	-.367	-.967	-.923	-.758	.080	.077	.103	.055	.134	.111	12	.014	
.50	-.304	-.343	-.378	-.945	-.936	-.756	.098	.076	.054	.036	.131	.088	13	.025	
.55	-.304	-.343	-.383	-.875	-.943		.104	.042	-.004	.148	.141				
.60	-.314	-.344	-.394	-.777	-.940	-.755	.084	.038	-.079	.154	.119	.064			
.65	-.315	-.338	-.374	-.624	-.944		.066	-.022	.167	.142	.087				
.68															
.70	-.325	-.338	-.365	-.526	-.864	-.733	.038	-.147	.197	.093		.027			
.73													.039		
.75	-.320	-.328	-.221				.046	.082	.162				.062		
.77													.279		
.79													.279		
.80	-.301	-.230													
.82															
.83															
.85	-.159	.021													
.87		.050	-.037	-.067											
.88		.058													
.89															
.90	-.016														
.91		.016													
.92		.022													
.93		-.019	-.059												
.94		.037													
.95		-.044													
.96		.052													
.97		-.089													
.98		.068													

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TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

		$\delta = -7.5 \quad M = 1.09$						$\alpha = 4.9$	$P_{t,j}/P_{\infty} = 1.0$				
x	c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan					
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.01	-0.059	-0.562						.203	.332				
.03	-0.103	-0.500	-0.588	-0.531	-0.520	-0.521		.203	.318	.240	.272	.256	
.05	-0.137	-0.492						.152	.327		.226	.215	.172
.08	-0.116	-0.122	-0.610	-0.558	-0.521			.167	.584	.216			
.10	-0.119	-0.098						.157			.186		
.15	-0.106	-0.146	-0.426	-0.544				.232	.249	.133		.132	
.20	-0.067							.231	.070	.152	.119	.105	.109
.25	-0.033	-0.155	-0.093	-0.455	-0.548	-0.519		.460	.009	.120	.098	.047	.069
.30	-0.096	-0.164	-0.104	-0.340	-0.587	-0.525		.149	.021	.099	.066	.031	.012
.35	-0.193	-0.174	-0.131	-0.284	-0.545	-0.534		.106	.007	.075	.028	.014	.033
.40	-0.144	-0.182	-0.133	-0.224	-0.512	-0.544		.030	.058	.049	.000	.028	.037
.45	-0.151	-0.132	-0.126	-0.182	-0.480	-0.546		.077	.052	.003	.015	.024	.024
.50	-0.165	-0.086	-0.138	-0.180	-0.408	-0.550		.089	.042	.043	.010	.026	.076
.55	-0.081	-0.059	-0.151	-0.177	-0.393			.071	.008	.107	.125	.095	.091
.60	-0.023	-0.084	-0.148	-0.191	-0.346	-0.549		.042	.046	.217	.139	.124	
.65	-0.070	-0.116	-0.175	-0.188	-0.295								
.68													
.70	-0.093	-0.134	-0.185	-0.206	-0.216	-0.546		.014	-0.143	-0.152	-0.164		.108
.73													
.75	-0.134	-0.141	-0.099			-0.145		.062	-0.295	-0.088			
.77													
.79													
.80	-0.151	-0.148				.114		.165	-0.165				
.82						.076		.165	-0.165				
.83													
.85	-0.001	.183	.213	.201	.036	.005	.038	.249	.056	.390	.286		
.87								.0420	.392	.247			
.88								.0413					
.89													
.90	.191	.179	.111	.006	.080	.0454	.364	.355	.391	.248	.150	.155	
.91								.315	.335	.302			
.92	.148	.138	.070					.313					
.93								.310	.304				
.94	.118												
.95	.094												
.96	.093												
.97													
.98	.033	.046											
		$\delta = -7.5 \quad M = 1.09$						$\alpha = 4.9$	$P_{t,j}/P_{\infty} = 5.0$				
x	c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan					
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90
.01	-0.048	-0.558						.212	.363				
.03	-0.088	-0.493	-0.573	-0.519	-0.525	-0.507	-0.516	.209	.331	.242	.256	.258	
.05	-0.079	-0.270						.163	.340	.225	.214	.186	
.08	-0.105	-0.098	-0.601					.176	.372	.203			
.10	-0.137	-0.081						.172			.185		
.15	-0.066	-0.145	-0.420	-0.534				.246	.251	.127		.138	
.20	-0.051							.270					
.25	-0.018	-0.135	-0.081	-0.432	-0.524	-0.515		.241	.078	.153	.124	.113	.124
.30	-0.088	-0.153	-0.093	-0.322	-0.544	-0.526		.177	.026	.102	.083	.086	
.35	-0.181	-0.148	-0.125	-0.250	-0.529	-0.534		.156	.020	.099	.071	.041	.026
.40	-0.119	-0.170	-0.136	-0.199	-0.498	-0.539		.217	.021	.076	.038	.009	.019
.45	-0.134	-0.135	-0.133	-0.171	-0.461	-0.539		.028	.052	.054	.007	.017	.007
.50	-0.129	-0.103	-0.146	-0.174	-0.427	-0.541		.052	.043	.008	.008	.045	.013
.55	-0.093	-0.083	-0.158	-0.174	-0.369			.088	.032	.034	.020	.060	
.60	-0.051	-0.100	-0.147	-0.186	-0.326	-0.538		.066	.005	.100	.095	.063	.013
.65	-0.073	-0.124	-0.171	-0.182	-0.271			.041	.044	.202	.016	.026	
.68													
.70	-0.096	-0.137	-0.177	-0.177	-0.185	-0.526		.014	-0.150	.059	.005		.307
.73													
.75	-0.134	-0.142	-0.067			.118		.057	-0.078	.067			
.77													
.79													
.80	-0.147	-0.127				.095		.156	.008				
.82													
.83													
.85	.019	.193	.216	.050		.064		.039	.003				
.87													
.88													
.89													
.90	.201	.120	.004	-0.059	-0.427			.285					
.91								.311					
.92	.162	.179						.290					
.93								.310					
.94	.130	.137	.079					.434	.358				
.95	.101							.338					
.96	.101							.331					
.97								.240					
.98	.034	.043	.029					.317					

TABLE IV.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

TABLE IV. - Continued

TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$\delta = 7.5 \quad M = 0.80 \quad \alpha = -0.1 \quad P_{t,j}/P_\infty = 5.0$							
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan					Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74		
.01	-0.079	-0.034					
.03	-0.044	-0.044	-0.139	-0.107			
.05	-0.019	-0.046					
.08	-0.009	-0.059	-0.098				
.10	-0.008	-0.073					
.13	-0.002	-0.088	-0.107				
.20	-0.059						
.25	-0.039	-0.070	-0.084	-0.103	-0.100	-0.152	
.30	-0.059	-0.094	-0.093	-0.103	-0.130	-0.108	
.35	-0.073	-0.123	-0.110	-0.129	-0.150	-0.180	
.40	-0.136	-0.126	-0.127	-0.158	-0.156	-0.185	
.45	-0.098	-0.116	-0.138	-0.151	-0.157	-0.184	
.50	-0.103	-0.133	-0.151	-0.175	-0.185	-0.183	
.55	-0.095	-0.131	-0.162	-0.170	-0.191		
.60	-0.116	-0.134	-0.146	-0.191	-0.194	-0.146	
.65	-0.106	-0.141	-0.165	-0.185	-0.224		
.68							
.70	-0.139	-0.150	-0.192	-0.266	-0.217	-0.108	
.73							
.75	-0.139	-0.162	-0.241		-0.175		
.77							
.79							
.80	-0.172	-0.217					
.82							
.83							
.85	-0.252	-0.443	-0.318	-0.181			
.87	-0.635						
.88	-0.312						
.89							
.90	-0.317	-0.102					
.91	-0.187						
.92	-0.219						
.93	-0.134	-0.057					
.94	-0.153						
.95	-0.094						
.96	-0.110						
.97	-0.012						
.98	-0.078	-0.050					

$\delta = 7.5 \quad M = 0.80 \quad \alpha = 4.9 \quad P_{t,j}/P_\infty = 1.0$							
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan					Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74		
.01	-0.104	-0.747					
.03	-0.112	-0.774	-0.774	-0.693			
.05	-0.180	-0.770		-0.702	-0.664	-0.663	
.08	-0.175	-0.317		-0.819			
.10	-0.183	-0.242		-0.739	-0.675		
.15	-0.142	-0.222		-0.566	-0.698		
.20	-0.244						
.25	-0.177	-0.189	-0.169	-0.633	-0.713	-0.646	
.30	-0.192	-0.207	-0.192	-0.527	-0.727	-0.654	
.35	-0.200	-0.232	-0.207	-0.419	-0.724	-0.663	
.40	-0.257	-0.232	-0.224	-0.349	-0.685	-0.597	
.45	-0.198	-0.215	-0.231	-0.269	-0.641	-0.647	
.50	-0.198	-0.220	-0.240	-0.241	-0.590	-0.636	
.55	-0.188	-0.220	-0.242	-0.217	-0.514		
.60	-0.203	-0.209	-0.218	-0.228	-0.428	-0.502	
.65	-0.174	-0.208	-0.238	-0.215	-0.397		
.70	-0.197	-0.208	-0.257	-0.288	-0.285	-0.557	
.73	-0.191	-0.208	-0.303		-0.195		
.75	-0.191	-0.208					
.77							
.79							
.80	-0.215	-0.252					
.82							
.83							
.85	-0.262	-0.478	-0.304	-0.175	-0.056		
.87	-0.612						
.88	-0.314						
.89							
.90	-0.357						
.91	-0.204						
.92	-0.242						
.93	-0.141	-0.061					
.94	-0.163						
.95	-0.110						
.96	-0.085						
.97	-0.071	-0.053					
.98							

TABLE IV - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X c	$\delta = 7.5 \quad M = 0.80$						$\alpha = 4.9 \quad p_{t,j}/p_{\infty} = 3.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90			
*C1	-0.096	-0.743					+0.264	.324	
*03	-0.106	-0.770	-0.760	-0.685	-0.693	-0.669	+0.224	.288	+0.213
*35	-0.172	-0.757					+0.196	.259	+0.218
*09	-0.163	-0.308	-0.806				+0.203	.272	+0.174
*10	-0.114	-0.245					+0.186		+0.197
*15	-0.135	-0.217	-0.564	-0.737	-0.580	-0.601	+0.193	.169	+0.134
*20	-0.242						+0.175		+0.130
*25	-0.174	-0.184	-0.168	-0.627	-0.714	-0.639	+0.134	.028	.079
*30	-0.182	-0.206	-0.184	-0.519	-0.729	-0.648	+0.196	-0.10	.062
*35	-0.191	-0.232	-0.199	-0.416	-0.725	-0.654	+0.186	.028	.029
*40	-0.245	-0.226	-0.226	-0.347	-0.685	-0.647	+0.071	.039	.016
*45	-0.185	-0.210	-0.230	-0.273	-0.641	-0.633	+0.012	.050	.011
*50	-0.191	-0.217	-0.236	-0.251	-0.586	-0.625	+0.011	.039	.012
*55	-0.186	-0.212	-0.238	-0.222	-0.514	-0.625	+0.031	.039	.006
*60	-0.196	-0.203	-0.219	-0.235	-0.433	-0.594	+0.023	.040	.056
*65	-0.167	-0.203	-0.238	-0.219	-0.352	-0.611	+0.055	.117	.132
*68							+0.098		
*70	-0.190	-0.205	-0.252	-0.295	-0.277	-0.548	-0.001	.000	.166
*73							+0.038	.114	.219
*75	-0.188	-0.211	-0.303		-0.199		+0.156		
*77							+0.158		
*79							+0.048		
*80	-0.215	-0.252			-0.119	-0.491	+0.104	.192	
*82							+0.144		
*83							+0.245		
*85	-0.255	-0.473	-0.303	-0.179		-0.060	+0.213	.299	
*87	-0.500	-0.175	-0.110				+0.282	.183	.057
*88	-0.313						+0.243		
*89							+0.267		
*90	-0.354	-0.107					+0.226		
*91	-0.206						+0.168	.182	.041
*92	-0.237						+0.125	.105	
*93	-0.142	-0.061					+0.124		
*94	-0.156						+0.083		
*95	-0.089						+0.080		
*96	-0.106						+0.076		
*97	-0.014						+0.049		
*98	-0.045						+0.049		
X c	$\delta = 7.5 \quad M = 0.80$						$\alpha = 4.9 \quad p_{t,j}/p_{\infty} = 4.7$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90			
*C1	-0.090	-0.736	-0.752	-0.665	-0.675	-0.651	-0.646		
*03	-0.100	-0.760							
*05	-0.165	-0.747							
*38	-0.160	-0.285	-0.788						
*10	-0.158	-0.237		-0.716	-0.659				
*15	-0.131	-0.207	-0.552		-0.681				
*20	-0.241								
*25	-0.165	-0.178	-0.154	-0.603	-0.495	-0.629			
*30	-0.177	-0.177	-0.171	-0.503	-0.708	-0.639			
*35	-0.181	-0.225	-0.192	-0.406	-0.708	-0.645			
*40	-0.247	-0.221	-0.213	-0.331	-0.564	-0.639			
*45	-0.102	-0.204	-0.218	-0.256	-0.618	-0.626			
*50	-0.186	-0.213	-0.224	-0.235	-0.565	-0.617			
*55	-0.180	-0.206	-0.227	-0.209	-0.497				
*60	-0.189	-0.199	-0.205	-0.220	-0.416	-0.583			
*65	-0.165	-0.203	-0.221	-0.205	-0.340				
*68									
*70	-0.184	-0.200	-0.242	-0.283	-0.268	-0.540			
*75	-0.181	-0.203	-0.288		-0.184				
*77									
*79									
*80	-0.211	-0.245			-0.104	-0.483			
*82									
*83									
*85	-0.254	-0.468	-0.295	-0.170		-0.049			
*87	-0.493	-0.163	-0.101						
*88	-0.311								
*89									
*90	-0.349	-0.098		-0.364	-0.009	-0.429			
*91	-0.197								
*92	-0.231								
*93	-0.136	-0.049							
*94	-0.150								
*95	-0.098	-0.092							
*96									
*97									
*98	-0.059	-0.034							

TABLE IV. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

X C	Wing upper surface stations, fraction of semispan						$\alpha = 9.9$	$P_{t,j}/P_{\infty} = 1.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-405	-1.101	-1.151	-0.995			.380	.401		
.03	-354	-1.162	-1.151	-0.995			.403	.427		
.05	-428	-1.237	-1.197	-1.009	-0.912	-0.207	.379	.409		
.08	-444	-1.477	-1.262				.369	.430		
.10	-428	-1.544		-1.027	-0.933		.358			
.15	-346	-0.929	-1.271		-0.951		.351	.341		
.20	-492						.331			
.25	-358	-0.230	-0.783	-1.170	-0.970	-0.154	.276	.162		
.30	-289	-0.380	-0.450	-1.197	-1.008	-1.43	.210	.103		
.35	-305	-0.440	-0.547	-1.200	-1.054	-1.35	.202	.075		
.40	-417	-0.416	-0.505	-1.205	-1.085	-1.18	.192	.055		
.45	-358	-0.367	-0.467	-1.124	-1.120	-0.096	.075	.044		
.50	-349	-0.351	-0.445	-1.023	-1.151	-0.082	.082	.056		
.55	-319	-0.325	-0.422	-0.866	-1.172		.073	.041		
.60	-309	-0.303	-0.382	-0.726	-1.157	-0.044	.065	.024		
.65	-267	-0.288	-0.378	-0.545	-1.132		.057	-0.009		
.70	-268	-0.269	-0.375	-0.510	-1.088	-0.010	.051	.028		
.73	-258	-0.259	-0.398		-1.030		.074	.112		
.77				-0.344						
.79				-0.218						
.80	-264	-0.276			-0.930	.015	.124	.208		
.82			-0.403							
.83			-0.327		-1.08					
.85	-288	-0.462			-0.805		.237	.329		
.87	-345	-0.241	-0.661				.311	.135		
.88	-312						.249	.101		
.89							.285			
.90	.140		-0.178		-0.672	.036	.621			
.91	-206			-0.633			.586			
.92	.199						.218	.195		
.93	-147		-0.125				.538	.072		
.94	.251						.161	.149		
.95	-0.95				-0.506		.502			
.96	.290			-0.017			.113	.049		
.97	.321	-0.054		-0.075			.099			
.98							.070			
X C	Wing upper surface stations, fraction of semispan						$\alpha = 9.9$	$P_{t,j}/P_{\infty} = 3.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.80				
.01	-417	-1.116					.364	.404		
.03	-360	-1.181	-1.162	-1.004			.390	.425		
.05	-501	-1.250		-1.020	-0.919	-0.762	.367	.412		
.08	-455	-1.501	-1.273				.361	.429		
.10	-439	-1.567		-1.036	-0.939		.352			
.15	-355	-0.930	-1.282		-0.959		.344	.340		
.20	-494						.223			
.25	-360	-0.235	-0.788	-1.181	-0.978	-0.694	.272	.167		
.30	-306	-0.388	-0.654	-1.209	-1.017	-0.692	.201	.108		
.35	-318	-0.449	-0.592	-1.213	-1.062	-0.671	.190	.078		
.40	-437	-0.429	-0.512	-1.217	-1.094	-0.647	.176	.058		
.45	-377	-0.472	-0.472	-1.135	-1.181	-0.621	.079	.061		
.50	-362	-0.360	-0.450	-1.029	-1.162	-0.602	.078	.059		
.55	-338	-0.333	-0.420	-0.872	-1.184		.071	.046		
.60	-325	-0.313	-0.389	-0.730	-1.171	-0.553	.063	.035		
.65	-286	-0.297	-0.386	-0.551	-1.143		.053	.009		
.70	-286	-0.284	-0.383	-0.516	-1.101	-0.507	.063	.059		
.73	-276	-0.268	-0.406		-1.043		.084	.160		
.75	-276	-0.268	-0.406							
.77				-0.351						
.79				-0.225						
.80	-281	-0.288			-0.941	-0.460	.142	.233		
.82				-0.412						
.83				-0.336	-0.113					
.85	-307	-0.476			-0.816		.239	.340		
.87	-467	-0.251	-0.667				.310	.228		
.88	-323						.275			
.89										
.90	-517		-0.189		-0.683	-0.449	.500			
.91	.219			-0.040			.255			
.92	-242		-0.159				.192	.200		
.93				-0.136						
.94	-175						.143	.150		
.95	-125		-0.103				.106	.130		
.96					-0.515					
.97	-0.86	-0.062		-0.024			.092	.044		
.98	-0.086	-0.062		-0.087			.059			

TABLE IV.- Continued

TABLE IV.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						$P_{t,i}/P_{\infty} = 3.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90			
.01	.097	-.009					.111	.050	
.03	.062	-.023	-.142	-.108			.078	.027	-.072
.05	.043	-.027		-.097	-.105	-.136	.043	.042	-.102
.08	.028	-.039	-.095				.034	.055	-.095
.10	.008	-.064		-.102	-.105		.015		
.15	.019	-.087	-.102		-.103		.021	-.066	-.123
.20		-.048					.031		-.114
.25	-.035	-.066	-.079	-.106	-.094	-.157	.009	-.114	-.119
.30	-.054	-.106	-.087	-.098	-.123	-.175	-.026	-.178	-.133
.35	-.068	-.124	-.107	-.125	-.145	-.189	-.112	-.178	-.128
.40	-.138	-.128	-.124	-.152	-.145	-.208	-.095	-.178	-.133
.45	-.094	-.118	-.132	-.147	-.147	-.229	-.136	-.177	-.138
.50	-.104	-.134	-.145	-.163	-.177	-.252	-.119	-.168	-.160
.55	-.105	-.134	-.152	-.151	-.172		.145	-.159	-.138
.60	-.121	-.133	-.131	-.168	-.173	-.262	-.131	-.158	-.145
.65	-.107	-.136	-.145	-.145	-.206		-.117	-.170	-.040
.68									-.024
.70	-.138	-.136	-.154	-.203	-.271	-.234	-.103	-.086	-.109
.73									-.012
.75	-.125	-.135	-.169				-.035	.035	.180
.77									-.012
.79									-.129
.80	-.148	-.162					.049	.139	
.82									-.129
.83									.002
.85	-.205	-.314					.163	.266	
.87									-.035
.88									.022
.89									.062
.90	-.607	-.295					.206	.247	
.91	-.673							.171	.112
.92	-.518								.022
.93	-.319	-.109						.110	.082
.94									-.001
.95	-.316							.112	
.96	-.145							.068	
.97	-.172								-.010
.98	-.095	-.058							
									.031
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						$P_{t,i}/P_{\infty} = 5.0$	Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90			
.01	.109	-.001					.112	.049	
.03	.048	-.019	-.124	-.095	-.097	-.130	.075	.039	-.071
.05	.049	-.008					.040	.049	-.105
.08	.034	-.025	-.092				.032	.061	-.095
.10	.012	-.051					.016		-.114
.15	.025	-.085	-.099				.021	-.055	-.119
.20		-.038					.029		-.110
.25		-.028	-.058	-.075	-.098	-.091	.151	.014	-.106
.30	-.049	-.085	-.085	-.093	-.118	-.176	.024	-.161	-.131
.35	-.063	-.114	-.103	-.113	-.142	-.185	.104	-.170	-.126
.40	-.133	-.120	-.119	-.142	-.142	-.204	.093	-.165	-.128
.45	-.092	-.108	-.128	-.139	-.146	-.225	.132	-.161	-.130
.50	-.102	-.122	-.141	-.156	-.174	-.247	.113	-.159	-.151
.55	-.098	-.122	-.148	-.146	-.173		.140	-.145	-.124
.60	-.117	-.122	-.128	-.165	-.173	-.257	.127	-.144	-.031
.65	-.105	-.128	-.142	-.145	-.205		.112	-.150	-.043
.68									-.044
.70	-.134	-.127	-.150	-.202	-.267	-.229	-.089	-.061	-.105
.73									-.018
.75	-.120	-.127	-.166				-.021	.056	.174
.77									-.015
.79									-.129
.80	-.146	-.153					-.037		-.129
.82									.015
.83									-.005
.85	-.199	-.308					.162	.272	
.87	-.733	-.727	-.553					.267	.163
.88	-.731							.233	
.89									-.005
.90	-.604	-.307					.247	.208	
.91	-.660							.180	.120
.92	-.519								-.022
.93	-.326	-.109							-.014
.94	-.322								-.005
.95	-.140								
.96	-.174								
.97									
.98	-.094	-.048							

TABLE IV. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

5 = 7.5 M = 0.90							$\alpha = -0.1$ $P_{t,j}/P_{\infty} = 5/6$								
X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p	
	0.10	0.32	0.53	0.65	0.74	0.90	0.10	0.32	0.53	0.65	0.74	0.90			
.01	.111	.006					.130	.064							
.03	.075	-.011	-.119	-.096			.076	.049	-.062	-.093	-.092				
.05	.051	-.004		-.081	-.095	-.132	.051	.062		-.093	-.090	-.030			
.08	.039	-.016	-.085				.046	.073	-.086						
.10	.018	-.040			-.068	-.095	.027			-.101					
.15	.031	-.066	-.094			-.095	.027	-.043	-.112		-.101				
.20	-.032						.026								
.25	-.022	-.042	-.065	-.033	-.085	-.150	.021	-.094	-.108	-.094	-.055	-.021			
.30	-.043	-.072	-.077	-.066	-.117	-.17C	.016	-.134	-.123	-.098	-.064	-.047	9	-.106	
.35	-.056	-.095	-.097	-.102	-.138	-.186	.027	-.161	-.117	-.099	-.064	-.048	10	-.091	
.40	-.028	-.126	-.148	-.146	-.128	-.204	.084	-.148	-.119	-.106	-.066	-.069	11	-.043	
.45	-.084	-.097	-.125	-.122	-.139	-.225	.123	-.152	-.120	-.088	-.049	-.067	12	-.135	
.50	-.089	-.111	-.136	-.148	-.167	-.245	.103	-.144	-.138	-.044	-.031	-.066	13	-.090	
.55	-.093	-.114	-.141	-.14C	-.167		.129	-.134	-.112	-.017	-.007				
.60	-.103	-.111	-.123	-.157	-.167	-.251	.112	-.130	-.016	-.025	-.029	-.055			
.65	-.094	-.117	-.136	-.138	-.200		.100	-.131	.044	-.066	-.047				
.68												-.033			
.70	-.122	-.116	-.147	-.196	-.267	-.211	.076	-.045	.104	.127		-.046			
.73	-.109	-.118	-.164		-.593	-.323	.014	.064	.179			-.021			
.75					-.568							-.019			
.77															
.79															
.80	-.138	-.145		-.733		-.367	-.122	.058	.143						
.82				-.720	-.545										
.83															
.85	-.193	-.297		-.728	-.694	-.301	-.274	.162	.275						
.87															
.88															
.89															
.90	-.597	-.257			-.090	-.012	.250								
.91					-.105		.211								
.92	-.492	-.636					.162	.186	.030	.019	.007				
.93															
.94	-.285	-.268	-.090				.12G	.129	.091						
.95	-.153	-.107			-.004		.075	.086		.019					
.96															
.97															
.98	-.079	-.032													
5 = 7.5 M = 0.90							$\alpha = +0.9$ $P_{t,j}/P_{\infty} = 1.0$								
X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p	
	0.10	0.32	0.53	0.65	0.74	0.90	0.10	0.32	0.53	0.65	0.74	0.90			
.01	-.058	-.677					.275	.331							
.03	-.069	-.700	-.733	-.673			.253	.290	.198	.196	.210				
.05	-.137	-.649					.207	.265							
.08	-.137	-.265	-.760				.209	.285	.119						
.10							.195								
.15							.203	.171	.074	.116	.103				
.20							.186								
.25	-.096	-.171	-.192	-.629	-.703	-.733	.149	.020	.059	.055	.100	.157			
.30	-.192	-.224	-.200	-.529	-.716	-.755	.094	.027	.047	.035	.076	.143	9	-.094	
.35	-.182	-.245	-.226	-.419	-.715	-.780	.084	.048	.012	.025	.070	.104	10	-.092	
.40	-.272	-.241	-.245	-.354	-.675	-.786	.058	.038	-.009	.011	.057	.066	11	-.062	
.45	-.213	-.224	-.225	-.287	-.642	-.774	.030	.077	-.022	.020	.061	.060	12	-.075	
.50	-.212	-.242	-.261	-.271	-.603	-.760	.023	.062	-.055	.048	.068	.043	13	-.063	
.55	-.209	-.239	-.262	-.247	-.538		.050	.057	-.048	.096	.094				
.60	-.224	-.234	-.235	-.257	-.480	-.704	.049	.074	.011	.096	.100	.024			
.65	-.201	-.235	-.249	-.235	-.427		.038	-.108	.080	.126	.100				
.68															
.70	-.222	-.226	-.249	-.280	-.408	-.630	.034	-.052	.143	.177		-.007			
.73							.005	.060	.222						
.75	-.211	-.219	-.255		-.686	-.392									
.77					-.597										
.79															
.80	-.224	-.234		-.795		-.341	-.554	.092	.166						
.82															
.83															
.85	-.256	-.384		-.762	-.534	-.205	.202	.310							
.87															
.88															
.89															
.90	-.623	-.251		-.115	-.079	-.466	.277								
.91															
.92	-.464	-.546					.240	.211	.159	.040	.019	-.1C3			
.93															
.94	-.292	-.263	-.102				.185								
.95															
.96	-.169	-.138			-.008		.136								
.97							.092								
.98	-.095	-.048			-.014										

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TABLE IV. - Continued

$$\delta = 7.5 \quad M = 0.90 \quad a = 4.9 \quad p_{t,i}/p_{cm} = 5.0$$

TABLE IV.- Continued

TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$\delta = 7.5$						$M = 1.00$	$\alpha = -0.1$	$P_{t,j}/P_{\infty} = 1.0$						
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	.095	.074					.098	.123						
.03	.074	.062	-.073	-.055			.077	.111	-.011	-.055	-.084			
.05	.075	.068		-.049	-.053	-.081	.062	.128		-.081	-.087	-.171		
.08	.076	.055	-.038				.071	.148	-.038					
.10	.072	.056			-.050	-.065	.080				-.066			
.15	.089	-.060	-.042				.105	-.001	-.072		-.097			
.20	.073						.118							
.25	.034	.020	-.035	-.060	-.073	-.135	.075	-.019	-.095	-.066	-.087	-.177		
.30	.021	-.032	-.032	-.035	-.094	-.148	.036	-.095	-.088	-.075	-.112	-.202	9	-.349
.35	.023	-.058	-.061	-.089	-.113	-.159	.012	-.114	-.086	-.093	-.147	-.244	10	-.390
.40	.073	-.067	-.082	-.107	-.121	-.170	.041	-.132	-.092	-.128	-.185	-.244	11	-.078
.45	.032	-.054	-.097	-.107	-.122	-.186	.119	-.137	-.097	-.154	-.202	-.225	12	-.067
.50	.041	-.075	-.109	-.127	-.145	-.196	.083	-.142	-.145	-.168	-.225	-.126	13	-.078
.55	.045	-.086	-.117	-.125	-.159		.129	-.107	-.191	-.185	-.239			
.60	.072	-.069	-.101	-.130	-.151	-.211	.111	-.131	-.253	-.253	-.187	-.025		
.65	.066	-.098	-.119	-.119	-.165		.095	-.178	-.256	-.099	-.039			
.68														
.70	-.095	-.096	-.121	-.141	-.182	-.233	.121	-.283	-.028	.117		-.047		
.73														
.75	-.086	-.097	-.116		-.219		.173	-.086	-.127					
.77														
.79														
.80	-.108	-.108					.055	-.056						
.82														
.83														
.85	-.128	-.188												
.87														
.88														
.89														
.90	-.471		-.506		-.312	-.297	-.239							
.91														
.92	-.450													
.93														
.94	-.433													
.95														
.96	-.410													
.97														
.98	-.400		-.369											
$\delta = 7.5$						$M = 1.00$	$\alpha = -0.1$	$P_{t,j}/P_{\infty} = 3.0$						
$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	.101	.073					.108	.119						
.03	.085	.056	-.083	-.060			.072	.1C9	-.016	-.092	-.1C8			
.05	.086	.065		-.055	-.060	-.084	.076	.123		-.087	-.101	-.183		
.08	.083	.054	-.043				.084	.143	-.045					
.10	.076	.030		-.057	-.069		.098	.122	-.007	-.078				
.15	.088	-.085	-.050				.120							
.20	.075						.072	-.029	-.092	-.078	-.097	-.186		
.25	.031	.013	-.041	-.064	-.083	-.140	.035	.102	-.086	-.095	-.120	-.205	9	-.350
.30	.015	-.038	-.039	-.064	-.105	-.153	.0C4	.131	-.1C7	-.104	-.154	-.183	10	-.371
.35	.019	-.068	-.066	-.089	-.123	-.164	.043	.146	-.105	-.135	-.193	-.073	11	-.104
.40	-.073	-.076	-.086	-.114	-.126	-.175	.127	-.134	-.111	-.160	-.206	-.001	12	-.111
.45	-.039	-.064	-.104	-.114	-.131	-.188	.086	.151	-.152	-.173	-.230	-.030	13	-.108
.50	.047	-.085	-.116	-.130	-.152	-.202	.125	-.132	-.194	-.184	-.222			
.55	.052	-.056	-.122	-.129	-.165		.117	-.144	-.246	-.232	-.020	-.050		
.60	-.081	-.098	-.105	-.134	-.160	-.210	.167	-.171	-.144	-.246	-.232	-.020		
.65	-.075	-.108	-.123	-.122	-.171		.1C6	-.184	-.139	-.054	-.086			
.68														
.70	-.1C6	-.108	-.125	-.147	-.190	-.220	.134	-.274	.080	.156		-.074		
.73							.167	-.023	.181					
.75	-.092	-.104	-.119		-.222									
.77														
.79														
.80	-.114	-.113			-.266	-.225	.023	.123						
.82														
.83														
.85	-.133	-.203	-.539	-.419		-.291	.153	-.257	-.246	-.098		-.024		
.87														
.88														
.89														
.90	-.474		-.507		-.315	-.300	-.231	.217	.188	.128	-.013	-.053	-.135	
.91														
.92	-.450													
.93														
.94	-.434		-.512	-.452										
.95														
.96	-.412		-.476		-.338	-.292	.069	.075		-.049	-.053			
.97														
.98	-.405		-.426		-.376	-.338		.015						

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TABLE IV.- Continued

		$\delta = 7.5$			$M = 1.00$		$\alpha = 0.0$		$P_t/P_{\infty} = 5.0$					
x c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp
.01	+.09	+.082					+.112	+.135						
.03	-.090	+.065	-.066	-.060			+.081	+.120	-.013	-.085	-.101			
.05	-.090	+.074			-.047	-.056	-.078	+.081	+.137	-.083	-.096	-.172		
.08	-.091	+.063	-.041				+.088	+.154	-.044					
.10	-.077	+.039			-.051	-.063		+.094		-.057				
.15	-.094	-.058	-.049		-.068			+.117	+.006	-.074		-.104		
.20	-.077							+.124						
.25	.033	.023	-.036	-.060	-.078	-.134		+.077	-.018	-.086	-.074	-.092	-.176	
.30	.018	-.022	-.033	-.060	-.098	-.145		+.035	-.091	-.079	-.079	-.115	-.181	9 -.341
.35	.023	-.057	-.060	-.079	-.118	-.157		+.012	-.105	-.105	-.097	-.148	-.134	10 -.360
.40	-.073	-.056	-.081	-.107	-.121	-.170		-.038	-.135	-.102	-.128	-.185	-.004	11 -.053
.45	-.033	-.046	-.099	-.109	-.126	-.182		+.012	-.122	-.107	-.156	-.200	-.003	12 -.042
.50	-.040	-.073	-.111	-.126	-.146	-.194		-.084	-.139	-.147	-.165	-.223	-.031	13 -.047
.55	-.046	-.083	-.117	-.125	-.161	-.197		-.115	-.119	-.185	-.179	-.169		
.60	-.077	-.088	-.102	-.129	-.152	-.197		-.109	-.134	-.238	-.192	-.043	-.044	
.65	-.072	-.098	-.120	-.119	-.164			-.100	-.179	-.085	-.069	-.098		
.68												+.085		
.70	-.101	-.097	-.122	-.141	-.184	-.213		-.124	-.262	-.085	-.164		-.067	
.73												+.064		
.75	-.085	-.096	-.116		-.213			-.155	-.013	-.191		-.055		
.77					-.451						-.188			
.79					-.419						-.188			
.80	-.108	-.102				-.259	-.217		-.001	-.134			-.022	-.090
.82			-.544							+.420				
.83			-.532	-.411						+.260	-.107			
.85	-.125	-.185				-.282			+.164	-.278				
.87			-.546	-.539	-.345				+.286	-.199	-.035			
.88			-.544						+.250					
.89										+.254				
.90	-.466		-.501		-.292	-.222			+.229		-.146		-.043	-.125
.91		-.519			-.309					+.206		-.003		
.92	-.444										-.146			
.93			-.499	-.446						+.184				
.94	-.428										+.160	-.093		
.95			-.480							+.169				
.96	-.404					-.331	-.284			+.117			-.043	
.97						-.377					+.084	-.035		
.98	-.399	-.422									+.032			
											+.034			

X C	Wing upper surface stations, fraction of semispan						$\alpha = C_{\text{L}} = 0$	$P_t, J / P_{\infty} = 6:1$	Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p	
	0.19	0.32	0.53	0.65	0.74	0.90			0.19	0.32	0.53	0.65	0.74	0.90			
.01	.130	.094					.142	.134									
.03	.115	.075	-.077	-.051			.083	.125	-.067	-.082	-.101						
.05	.110	.077		-.045	-.056	-.074	.065	.138		-.080	-.095	-.166					
.08	.103	.062	-.036				.104	.156	-.034								
.10	.091	.039		-.048	-.065		.109			-.059							
.15	.093	-.052		-.049	-.070		.132	.003	-.069		-.101						
.20	.088						.130										
.25	.042	.021	-.036	-.057	-.079	-.130	.084	-.020	-.084	-.075	-.089	-.165					
.30	.028	.024	-.032	-.057	-.096	-.139	.047	-.070	-.079	-.112	-.156	9	-.337				
.35	.027	-.062	-.053	-.055	-.075	-.118	.015	-.119	-.104	-.145	-.092	10	-.342				
.40	-.071	-.056	-.056	-.060	-.078	-.123	.054	-.054	-.060	-.121	-.020	11	-.026				
.45	-.067	-.057	-.057	-.094	-.104	-.123	.013	-.124	-.103	-.151	-.057	12	-.030				
.50	-.042	-.081	-.113	-.124	-.147	-.185	.079	-.124	-.144	-.166	-.218	-.035	13	-.011			
.55	-.044	-.091	-.115	-.124	-.161		.115	-.125	-.183	-.172	-.111						
.60	-.071	-.090	-.097	-.123	-.152	-.188	.108	-.138	-.228	-.126	.053	-.044					
.65	-.065	-.100	-.115	-.115	-.162		.098	-.180	-.030	.074	.102						
.68																	
.70	-.098	-.103	-.116	-.140	-.181	-.205		-.120	-.245	.071	.167						
.73																	
.75	-.086	-.101	-.110		-.211			-.134	.028	.188							
.77																	
.79																	
.80	-.108	-.110			-.258	-.210		.010	.112								
.82																	
.83																	
.85	-.126	-.193	-.527	-.448	-.283			.164	.274	.393							
.87									.283	.266	.107						
.88									.257	.205	.029						
.89																	
.90	-.466	-.495		-.309	-.290	-.216		.225	.211	-.136	-.016	-.048	-.124				
.91									.179	.141	.070						
.92	-.442									.127	.069						
.93																	
.94	-.426																
.95	-.481	-.468			-.331	-.280											
.97																	
.98	-.391	-.414								.910	.021						

TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$\delta = 7.5 \quad M = 1.00 \quad \alpha = 4.9 \quad P_{t,j}/P_{\infty} = 1.0$						Nacelle Orifice number	Cp					
X c	Wing upper surface stations, fraction of semispan											
	0.19	0.32	0.53	0.65	0.74	0.90						
.01	-.010	-.536					.272	.391				
.03	-.017	-.545	-.629	-.558			.267	.359	.230	.229	.218	
.05	-.071	-.460		-.566	-.563	-.622	.253	.333		.188	.174	.235
.08	-.061	-.092		-.654			.261	.356	.180			
.10	-.062						.260					
.15	-.021	-.253		-.512	-.584	-.569	.273	.232	.105	.144		
.20	-.085						.264					
.25	-.080	-.102	-.114	-.500	-.599	-.652	.258	.072	.099	.065	.063	.169
.30	-.203	-.122	-.132	-.427	-.413	-.473	.150	.015	.074	.042	.033	.143
.35	-.152	-.164	-.170	-.371	-.411	-.700	.157	-.010	.041	.032	.032	.131
.40	-.158	-.171	-.193	-.331	-.379	-.718	.131	-.023	.016	.024	.011	.10
.45	-.128	-.154	-.212	-.285	-.545	-.730	.003	-.059	-.022	-.056	.073	.063
.50	-.135	-.175	-.231	-.270	-.517	-.742	.012	-.026	.002	.020	.058	.051
.55	-.136	-.188	-.240	-.253	-.468	-.718	.018	-.026	.006	.116	.106	.128
.60	-.166	-.196	-.218	-.248	-.424	-.747	.018	-.056	.046	.114	.121	.047
.65	-.160	-.202	-.238	-.226	-.383	-.708	.018	-.114	.099	.152	.136	
.68												
.70	-.197	-.207	-.243	-.261	-.366	-.739	.059	-.169	.156	.208		.010
.73												
.75	-.192	-.206	-.226				.050	.080	.237			
.77												
.79												
.80	-.217	-.222										
.82												
.83												
.85	-.217	-.286										
.87	-.591	-.608	-.410									
.88												
.89												
.90	-.552	-.588										
.91	-.569											
.92	-.512											
.93	-.566	-.533										
.94	-.505											
.95	-.555											
.96	-.483											
.97	-.472	-.429										
.98												
$\delta = 7.5 \quad M = 1.00 \quad \alpha = 4.9 \quad P_{t,j}/P_{\infty} = 3.0$						Nacelle Orifice number	Cp					
X c	Wing upper surface stations, fraction of semispan											
	0.19	0.32	0.53	0.65	0.74	0.90						
.01	.001	-.520	-.615	-.545			.268	.397				
.03	-.007	-.532					.281	.361	.243	.240	.229	
.05	-.053	-.440					.256	.342	.198	.184	.253	
.08							.265	.363	.172			
.10	-.047	-.084					.288					
.15	-.011	-.128					.285	.238	.118	.105		
.20	-.075						.273					
.25	-.067	-.091	-.107	-.483	-.576	-.638	.228	.079	.107	.081	.108	.186
.30	-.189	-.110	-.119	-.408	-.589	-.644	.158	.025	.077	.060	.095	.158
.35	-.140	-.151	-.154	-.368	-.585	-.688	.142	.006	.048	.032	.101	.147
.40	-.145	-.160	-.182	-.313	-.550	-.706	.137	-.009	.027	.005	.078	.107
.45	-.121	-.142	-.200	-.264	-.517	-.720	.019	-.016	.003	.031	.107	.098
.50	-.122	-.163	-.217	-.251	-.489	-.731	.023	-.024	.042	.089	.117	.081
.55	-.125	-.176	-.225	-.225	-.439	-.737	.063	-.018	.079	.146	.145	.061
.60	-.153	-.183	-.202	-.226	-.396	-.737	.003	-.045	.075	.150	.153	
.65	-.149	-.190	-.223	-.207	-.360	-.737	.003	-.083	.155	.184	.155	
.68												
.70	-.184	-.194	-.221	-.240	-.347	-.733	.026	-.020	.212	.233		.024
.73												
.75	-.179	-.193	-.208				.027	.142	.277			
.77												
.79												
.80	-.203	-.204										
.82												
.83												
.85	-.210	-.273	-.589	-.458			.269	.348	.313	.134		
.87	-.591	-.589	-.389					.343	.239	.057		
.88	-.594							.307				
.90	-.528											
.91	-.548	-.567	-.355	-.336	-.560	-.293						
.92	-.488											
.93	-.547	-.512										
.94	-.481											
.95	-.535											
.96	-.461											
.97												
.98	-.452	-.452										

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TABLE IV - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

5 = 7.5 M = 1.00						$\alpha = 4.9$	$p_{t,j}/p_{\infty} = 5:1$	Wing lower surface stations, fraction of semispan			Nacelle				
X	C	Wing upper surface stations, fraction of semispan						0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01		-0.011	-0.519												
.03		-0.011	-0.534	-0.619	-0.533	-0.567	-0.634	+0.269	+0.192						
.05		-0.012	-0.435					+0.287	+0.360	+0.241	+0.241	+0.232			
.08		-0.014	-0.080	-0.632	-0.542	-0.535	-0.604	+0.254	+0.337	+0.198	+0.198	+0.189	+0.260		
.10		-0.015	-0.088					+0.268	+0.367	+0.172					
.15		-0.016	-0.239	-0.486	-0.560	-0.542		+0.269		+0.155					
.20		-0.017						+0.297	+0.237	+0.116	+0.116	+0.111			
.25		-0.019	-0.095	-0.095	-0.479	-0.567	-0.634	+0.276	+0.076	+0.111	+0.087	+0.121	+0.191		
.30		-0.018	-0.110	-0.113	-0.405	-0.578	-0.660	+0.158	+0.324	+0.281	+0.064	+0.127	+0.161	9 -0.061	
.35		-0.017	-0.149	-0.152	-0.349	-0.575	-0.683	+0.144	+0.005	+0.053	+0.040	+0.109	+0.152	10 -0.011	
.40		-0.015	-0.157	-0.177	-0.308	-0.543	-0.702	+0.144	+0.009	+0.030	+0.019	+0.104	+0.113	11 +0.137	
.45		-0.017	-0.138	-0.192	-0.259	-0.515	-0.716	+0.020	+0.018	+0.008	+0.052	+0.112	+0.104	12 +0.160	
.50		-0.019	-0.157	-0.211	-0.245	-0.484	-0.729	+0.029	+0.023	+0.038	+0.101	+0.123	+0.086	13 +0.160	
.55		-0.019	-0.170	-0.217	-0.222	-0.432		+0.002	+0.017	+0.066	+0.156	+0.150			
.60		-0.017	-0.179	-0.198	-0.219	-0.388	-0.735	+0.002	+0.39	+0.087	+0.156	+0.158	+0.066		
.65		-0.012	-0.186	-0.214	-0.201	-0.352		+0.004	+0.75	+0.167	+0.193	+0.159			
.68		-0.180	-0.191	-0.215	-0.235	-0.340	-0.731	-0.014	+0.12	+0.214	+0.237	+0.237	+0.029		
.73		-0.173	-0.188	-0.203		-0.342		+0.051	+0.154	+0.279		+0.113			
.77					-0.469	-0.543					+0.226	+0.226	+0.101		
.79															
.80		-0.198	-0.198				-0.604	-0.338	-0.653	+0.158	+0.247		+0.67	-0.014	
.82											+0.416				
.85		-0.200	-0.271	-0.286	-0.453	-0.387	-0.382	+0.277	+0.361	+0.325	+0.141	+0.69	+0.032		
.87											+0.323				
.88															
.89															
.90		-0.525	-0.563	-0.350	-0.331	-0.557		+0.339	+0.300	+0.191	+0.022	+0.004	+0.078		
.91		-0.544						+0.251							
.92		-0.486						+0.239							
.93		-0.546	-0.503					+0.185	+0.194	+0.147					
.94		-0.479						+0.126	+0.151	+0.055					
.95		-0.532				-0.352		+0.082		+0.014					
.96		-0.457			-0.416			+0.080							
.97		-0.449	-0.454												
.98															
5 = 7.5 M = 1.00						$\alpha = 9.9$	$p_{t,j}/p_{\infty} = 1:0$	Wing lower surface stations, fraction of semispan			Nacelle				
X	C	Wing upper surface stations, fraction of semispan						0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01		-0.129	-0.733					+0.425	+0.527						
.03		-0.122	-0.767	-0.942	-0.797	-0.812	-0.773	-0.776	+0.475	+0.530	+0.371	+0.341	+0.298		
.05		-0.246	-0.867					+0.478	+0.509	+0.316	+0.301	+0.280			
.08		-0.220	-0.956	-1.003				+0.477	+0.525	+0.320					
.10		-0.212	-0.975		-0.842	-0.793		+0.467		+0.283					
.15		-0.149	-0.816	-1.095		-0.819		+0.457	+0.425	+0.269	+0.255				
.20		-0.221						+0.435							
.25		-0.177	-0.096	-0.641	-0.949	-0.833	-0.786	+0.375	+0.223	+0.249	+0.219	+0.247	+0.267		
.30		-0.277	-0.212	-0.443	-0.952	-0.863	-0.807	+0.295	+0.167	+0.221	+0.194	+0.224	+0.260		
.35		-0.320	-0.286	-0.374	-0.960	-0.896	-0.828	+0.267	+0.134	+0.181	+0.175	+0.217	+0.214	10 +0.020	
.40		-0.231	-0.312	-0.383	-0.968	-0.947	-0.841	+0.229	+0.117	+0.151	+0.159	+0.202	+0.181	11 +0.175	
.45		-0.228	-0.291	-0.398	-0.946	-0.922	-0.893	+0.151	+0.108	+0.129	+0.162	+0.202	+0.173	12 +0.166	
.50		-0.227	-0.293	-0.365	-0.907	-0.933	-0.860	+0.157	+0.117	+0.085	+0.087	+0.212	+0.156	13 +0.174	
.55		-0.257	-0.228	-0.367	-0.819	-0.940		+0.138	+0.092	+0.075	+0.075	+0.223	+0.214	+0.130	
.60		-0.273	-0.296	-0.343	-0.716	-0.943	-0.880	+0.110	+0.080	+0.044	+0.044	+0.223	+0.204	+0.185	
.65		-0.265	-0.291	-0.356	-0.657	-0.949		+0.096	+0.066	+0.026	+0.026	+0.248	+0.204	+0.185	
.68															
.70		-0.292	-0.301	-0.356	-0.513	-0.980	-0.900	+0.074	+0.044	+0.264	+0.283	+0.291			
.73		-0.287	-0.292	-0.352		-0.994		+0.091	+0.166	+0.331		+0.155			
.77					-0.699						+0.256	+0.259			
.79					-0.610										
.80		-0.311	-0.309			-0.998	-0.789	+0.178	+0.278			+0.107	+0.041		
.82				-0.698							+0.513				
.83				-0.674	-0.476		-0.933	+0.305	+0.401	+0.381	+0.174				
.85		-0.322	-0.350			-0.629	-0.360					+0.371			
.87		-0.644													
.88		-0.645													
.90		-0.576	-0.596	-0.335	-0.962	-0.685		+0.380	+0.337						
.91		-0.614						+0.294	+0.239						
.92		-0.540						+0.273		+0.222	+0.178				
.93		-0.609	-0.604					+0.214		+0.161					
.94		-0.526													
.95		-0.606													
.96		-0.510	-0.413	-0.348	-0.934			+0.150		+0.111		+0.036			
.97		-0.501	-0.335					+0.078							

CONTINUATION

TABLE IV - Continued

X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle		
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p	
.01	-0.142	-0.743					.438	.523							
.03	-0.123	-0.778	-0.959	-0.809			.484	.524	.372	.340	.295				
.05	-0.265	-0.874		-0.824	-0.784	-0.787	.477	.505		.316	.305	.274			
.08	-0.226	-0.962	-1.011				.476	.522	.321						
.10	-0.214	-0.986		-0.852	-0.805		.466			.286					
.15	-0.151	-0.831	-1.105		-0.830		.456	.421	.271		.263				
.20	-0.233						.435								
.25	-0.184	-0.101	-0.650	-0.958	-0.840	-0.797	.374	.217	.249	.227	.257	.264			
.30	-0.295	-0.228	-0.453	-0.561	-0.871	-0.814	.286	.160	.221	.206	.233	.257	9	aC40	
.35	-0.333	-0.299	-0.385	-0.971	-0.906	-0.838	.266	.132	.184	.158	.226	.211	10	aC50	
.40	-0.246	-0.324	-0.373	-0.980	-0.924	-0.852	.230	.107	.159	.176	.211	.178	11	a168	
.45	-0.236	-0.303	-0.366	-0.959	-0.933	-0.863	.137	.106	.141	.182	.209	.170	12	a167	
.50	-0.267	-0.307	-0.373	-0.917	-0.943	-0.870	.138	.102	.106	.205	.207	.151	13	a172	
.55	-0.271	-0.309	-0.373	-0.833	-0.952		.133	.088	.109	.247	.224				
.60	-0.284	-0.310	-0.349	-0.727	-0.954	-0.891	.117	.054	.178	.235	.215	.125			
.65	-0.274	-0.304	-0.363	-0.577	-0.960		.100	.027	.245	.254	.201				
.68											.181				
.70	-0.307	-0.313	-0.361	-0.523	-0.990	-0.911	.096	.080	.296	.282					
.73															
.75	-0.296	-0.303	-0.357			-1.006	.123	.207	.334						
.77						-0.709									
.79						-0.618									
.80	-0.321	-0.319		-0.704		-1.011	-0.801	.205	.301						
.82															
.83				-0.680	-0.484										
.85	-0.330	-0.361				-0.997	.328	.395							
.87	-0.667		-0.635	-0.368			.379	.329							
.88	-0.658						.378								
.89							.334								
.90	-0.581		-0.602		-0.977	-0.698	.265								
.91	-0.623		-0.344				.262								
.92	-0.544						.203								
.93	-0.617	-0.609					.139								
.94	-0.530														
.95	-0.615														
.96	-0.513														
.97	-0.506	-0.577		-0.426			.094								
.98							.049								

$$\delta = 7.5 \quad M = 1.05 \quad \alpha = -0.1 \quad g_{\text{eff}}/\sqrt{E_\infty} = 1.0$$

TABLE IV.- Continued

TABLE IV. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

 $\delta = 7.5 \quad M = 1.05 \quad \alpha = -0.1 \quad P_{t,i}/P_{\infty} = 6.3$

x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.088	.039					.085	.079						
.03	.058	.036	-.113	-.090			.062	.077	-.042	-.100	-.120			
.05	.042	.046		-.081	-.087	-.094	.024	.112		-.109	-.118	-.172		
.08	.025	.055		-.068			.016	.151	-.082					
.10	.069	.048			-.100	-.091	.002			-.114				
.13	.024		-.055	-.081		-.100	.043	.028	-.097		-.148			
.20	.054	.026		-.097	-.088	-.108	.141	.084	-.048	-.122	-.131	-.165		
.30	.025	.047	-.059	-.082		-.125	.151	.016	.097	-.129	-.120	-.183	9	-.332
.35	.040	.097	-.080	-.107		-.146	.141	.055	.148	-.167	-.122	-.216	10	-.363
.40	.099	.117	-.108	-.130		-.146	.173	.042	.185	-.137	-.146	-.228	11	-.075
.45	.080	.099	-.122	-.130		-.151	.189	.149	.179	-.136	-.170	-.242	12	-.059
.50	.085	.107	-.138	-.151		-.171	.203	.137	.167	-.171	-.170	-.207	13	-.048
.55	.080	.115	-.145	-.150		-.181	.166	.155	.163	-.229	-.228	-.234		
.60	.101	.121	-.124	-.146		-.177	.228	.140	.189	-.229	-.210	-.110		
.65	.093	.119	-.139	-.140		-.184						-.021		
.68														
.70	-.115	-.127	-.143	-.164		-.198	-.235		-.141	-.257	-.029	.031		-.067
.73												.024		
.75	-.107	-.116	-.129			-.206			-.174	-.060	.042		.033	
.77												.165		
.79												.165		
.80	-.128	-.127				-.236	-.232		-.128	-.013				
.82												.335		
.83												.240	.097	
.85	-.124	-.175				-.261				.088	.222			
.87											.240	.195	.036	
.88											.220			
.89														
.90	-.441	-.471				-.276	-.229			.216	.201	.158		-.027
.91												.004		-.068
.92	-.410											.172		
.93														
.94														
.95	-.398	-.445				-.274				.138				
.96														
.97	-.378					-.307				.093	.120			-.027
.98												.044		
												.034		

x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-.042	-.592					.220	.347						
.03	-.046	-.559	-.652	-.605			.224	.315	-.187	.174	.192			
.05	-.124	-.361		-.612			.174	.307		.138	.149	.132		
.08	-.120	-.104		-.667			.150	.342	-.117					
.10	-.129	-.110			-.630	-.603	.146	.341						
.15	-.072	-.206			-.518	-.624	.233	.214	-.041		.071			
.20	-.101						.253							
.25	-.660	-.192	-.180	-.502	-.623	-.605	.211	.037	-.053	.045	.053	.097		
.30	-.152	-.183	-.191	-.412	-.630	-.618	.126	.022	-.026	.030	.013	.152	9	-.273
.35	-.238	-.224	-.222	-.345	-.613	-.631	.100	.065	-.008	.004	.018	.123	10	-.298
.40	-.242	-.235	-.236	-.307	-.572	-.638	.112	.104	-.009	-.029	-.062	.085	11	-.030
.45	-.187	-.212	-.212	-.278	-.528	-.645	.073	.088	-.321	-.056	-.081	.080	12	-.024
.50	-.181	-.221	-.237	-.278	-.490	-.652	.050	.100	-.063	-.074	-.106	.072	13	-.027
.55	-.162	-.228	-.221	-.272	-.432		.076	.055	-.107	-.089	-.094			
.60	-.203	-.236	-.240	-.274	-.391	-.660	.054	.068	-.171	-.158	.109			
.65	-.193	-.231	-.234	-.264	-.351		.024	.124	-.183	.084	.149			
.68														
.70	-.229	-.238	-.254	-.288	-.336	-.666	.056	.213	-.057	.204		.037		
.73												.118		
.75	-.213	-.229	-.243	-.329				.101	-.092	.203		.109		
.77												.230		
.79												.230		
.80	-.234	-.238			-.348	-.598		.127	.111					
.82												.432		
.83												.315	.153	
.85	-.227	-.281			-.346			.185	.287					
.87												.309	.258	.078
.88												.290		
.89														
.90	-.911	-.555			-.343	-.533		.272	.248	-.205		.016		-.055
.91														
.92	-.479	-.538			-.360			.230	.199	.154				
.93														
.94	-.469	-.533	-.525											
.95	-.451	-.534			-.365	-.341								
.96														
.97	-.451	-.452												
.98														

CONFIDENTIAL

TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$\delta = 7.5$						$M = 1.05$						$\alpha = 4.9$						$P_{t,j}/P_{\infty} = 3.0$					
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle										
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp	Orifice number	Cp	Orifice number	Cp					
.01	-0.043	-0.592	-0.644	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.213	-0.351	-0.209	-0.315	-0.191	-0.187	-0.197	-0.149			
.03	-0.049	-0.550	-0.644	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.161	-0.209	-0.161	-0.215	-0.191	-0.151	-0.153	-0.149			
.05	-0.124	-0.339	-0.644	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.173	-0.346	-0.173	-0.322	-0.192	-0.151	-0.153	-0.149			
.08	-0.116	-0.094	-0.65b	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.181	-0.181	-0.181	-0.181	-0.181	-0.181	-0.181	-0.149			
.10	-0.129	-0.167	-0.644	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.230	-0.216	-0.230	-0.216	-0.204	-0.187	-0.197	-0.149			
.15	-0.070	-0.207	-0.500	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.255	-0.255	-0.255	-0.255	-0.246	-0.204	-0.204	-0.149			
.20	-0.090	-0.240	-0.190	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.269	-0.269	-0.269	-0.269	-0.261	-0.204	-0.204	-0.149			
.25	-0.149	-0.180	-0.183	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.120	-0.181	-0.120	-0.181	-0.130	-0.099	-0.116	-0.170	9	-0.265	
.35	-0.228	-0.231	-0.213	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.098	-0.061	-0.098	-0.061	-0.095	-0.012	-0.015	-0.136	10	-0.281	
.40	-0.237	-0.233	-0.227	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.115	-0.099	-0.115	-0.099	-0.092	-0.019	-0.058	-0.100	11	-0.331	
.45	-0.182	-0.213	-0.234	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.069	-0.073	-0.069	-0.073	-0.014	-0.047	-0.077	-0.095	12	-0.036	
.50	-0.184	-0.216	-0.248	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.045	-0.080	-0.045	-0.080	-0.054	-0.065	-0.054	-0.082	13	-0.036	
.55	-0.178	-0.224	-0.253	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.062	-0.058	-0.062	-0.058	-0.099	-0.078	-0.117	-0.073			
.60	-0.199	-0.230	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.048	-0.065	-0.048	-0.065	-0.158	-0.027	-0.152	-0.073			
.65	-0.189	-0.225	-0.244	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.027	-0.106	-0.027	-0.106	-0.004	-0.169	-0.163	-0.146			
.68	-0.219	-0.233	-0.245	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.055	-0.195	-0.055	-0.195	-0.165	-0.237	-0.237	-0.044			
.73	-0.208	-0.225	-0.234	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.095	-0.024	-0.095	-0.024	-0.255	-0.239	-0.239	-0.112			
.75	-0.228	-0.231	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.021	-0.191	-0.021	-0.191	-0.420	-0.326	-0.159	-0.079	-0.06		
.77	-0.181	-0.218	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.174	-0.346	-0.174	-0.346	-0.125	-0.103	-0.077	-0.043			
.79	-0.194	-0.217	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.183	-0.215	-0.183	-0.215	-0.046	-0.075	-0.020	-0.051			
.80	-0.228	-0.231	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.292	-0.292	-0.292	-0.292	-0.201	-0.037	-0.020	-0.051			
.82	-0.181	-0.218	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.242	-0.256	-0.242	-0.256	-0.080	-0.198	-0.148	-0.020			
.83	-0.184	-0.218	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.195	-0.140	-0.195	-0.140	-0.086	-0.020					
.85	-0.184	-0.216	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.072	-0.084	-0.072	-0.084	-0.010	-0.125	-0.115	-0.010			
.86	-0.184	-0.216	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.072	-0.084	-0.072	-0.084	-0.010	-0.125	-0.115	-0.010			
.88	-0.184	-0.216	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.195	-0.140	-0.195	-0.140	-0.086	-0.020					
.89	-0.184	-0.216	-0.231	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.218	-0.218	-0.218	-0.218	-0.160	-0.042	-0.176	-0.167			
.90	-0.502	-0.547	-0.547	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.292	-0.398	-0.292	-0.398	-0.173	-0.239	-0.239	-0.010			
.91	-0.528	-0.547	-0.547	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.242	-0.398	-0.242	-0.398	-0.125	-0.239	-0.239	-0.010			
.92	-0.472	-0.527	-0.513	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.195	-0.140	-0.195	-0.140	-0.086	-0.020					
.93	-0.462	-0.527	-0.513	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.140	-0.086	-0.140	-0.086	-0.084	-0.020					
.94	-0.446	-0.527	-0.513	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.140	-0.149	-0.140	-0.149	-0.157	-0.024					
.95	-0.446	-0.529	-0.513	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.140	-0.149	-0.140	-0.149	-0.084	-0.024					
.96	-0.446	-0.446	-0.446	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.067	-0.084	-0.067	-0.084	-0.084	-0.024					
.97	-0.329	-0.483	-0.483	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.067	-0.084	-0.067	-0.084	-0.084	-0.024					
.98	-0.329	-0.483	-0.483	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.067	-0.084	-0.067	-0.084	-0.084	-0.024					

$\delta = 7.5$						$M = 1.05$						$\alpha = 4.9$						$P_{t,j}/P_{\infty} = 5.0$					
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle										
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp	Orifice number	Cp	Orifice number	Cp					
.01	-0.047	-0.590	-0.644	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.210	-0.356	-0.210	-0.356	-0.194	-0.186	-0.198	-0.170	9	-0.279	
.03	-0.055	-0.556	-0.644	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.166	-0.310	-0.166	-0.310	-0.150	-0.155	-0.155	-0.170	10	-0.013	
.05	-0.138	-0.353	-0.644	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.174	-0.346	-0.174	-0.346	-0.127	-0.014	-0.014	-0.136	11	-0.018	
.08	-0.124	-0.092	-0.65b	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.166	-0.332	-0.166	-0.332	-0.104	-0.047	-0.077	-0.095	12	-0.014	
.10	-0.134	-0.107	-0.644	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.183	-0.215	-0.183	-0.215	-0.103	-0.077	-0.082	-0.098	13	-0.003	
.15	-0.070	-0.204	-0.517	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.252	-0.252	-0.252	-0.252	-0.039	-0.071	-0.052	-0.059			
.20	-0.098	-0.219	-0.169	-0.594	-0.601	-0.587	-0.600	-0.591	-0.595	-0.597	-0.598	-0.599	-0.211	-0.356	-0.211	-0.356	-0.020	-0.037	-0.017	-0.040			
.25	-0																						

TABLE IV.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

S = 7.5 M = 1.05						$\alpha = 10^\circ$ $P_{t,j}/P_{\infty} = 1.0$								
X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.173	-0.740					.346	.504						
.03	-0.159	-0.758	-0.963	-0.800			.410	.504	.337	.314	.278			
.05	-0.265	-0.862					.422	.482						
.08	-0.250	-0.916	-0.983		-0.777	-0.769	.432	.506	.281	.291	.273	.302		
.10	-0.254	-0.910					.426							
.15	-0.183	-0.723	-1.033				.433	.395	.229					
.20	-0.221						.422							
.25	-0.178	-0.142	-0.682	-0.915	-0.820	-0.769	.356	.174	.221	.194	.228	.296		
.30	-0.234	-0.260	-0.442	-0.920	-0.850	-0.784	.261	.113	.198	.168	.218	.289	9	-0.134
.35	-0.372	-0.350	-0.366	-0.928	-0.877	-0.800	.230	.086	.161	.144	.220	.250	10	-0.022
.40	-0.310	-0.342	-0.355	-0.938	-0.888	-0.810	.196	.057	.132	.124	.214	.219	11	-0.198
.45	-0.256	-0.338	-0.352	-0.936	-0.891	-0.820	.084	.056	.139	.150	.220	.215	12	-0.208
.50	-0.298	-0.334	-0.364	-0.909	-0.899	-0.827	.101	.081	.082	.196	.227	.199	13	-0.200
.55	-0.289	-0.329	-0.368	-0.829	-0.905		.110	.069	.025	.246	.251			
.60	-0.297	-0.328	-0.342	-0.720	-0.908	-0.843	.092	.044	.163	.247	.248	.178		
.65	-0.284	-0.318	-0.357	-0.560	-0.906		.079	-0.306	.228	.278	.240			
.68												.224		
.70	-0.310	-0.324	-0.352	-0.481	-0.923	-0.865	.056	-0.022	.285	.321		.145		
.73												.196		
.75	-0.299	-0.317	-0.342		-0.934		.073	.187	.356			.185		
.77												.305		
.79												.305		
.80	-0.316	-0.326					.098	-0.774	.205	.281				
.82												.153		
.83												.1CC		
.85	-0.325	-0.352					.046							
.87														
.88														
.89														
.90	-0.547	-0.560					.019	-0.675	.375	.290	.290	.030		
.91	-0.592						.320							
.92	-0.519													
.93	-0.583	-0.559												
.94	-0.506													
.95	-0.579													
.96	-0.488													
.97	-0.475													
.98	-0.464	-0.545												
S = 7.5 M = 1.05						$\alpha = 9^\circ$ $P_{t,j}/P_{\infty} = 3.0$								
X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle Orifice number	C _p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	-0.197	-0.737					.318	.509						
.03	-0.172	-0.758	-0.956	-0.826			.389	.507	.341	.321	.293			
.05	-0.279	-0.861					.406	.487						
.08	-0.263	-0.917	-0.988				.422	.510	.286					
.10	-0.258	-0.909					.425							
.15	-0.184	-0.707	-1.043				.433	.401	.237					
.20	-0.213						.421							
.25	-0.174	-0.130	-0.677	-0.922	-0.823	-0.772	.356	.182	.227	.213	.261	.299		
.30	-0.234	-0.243	-0.438	-0.928	-0.855	-0.790	.265	.126	.201	.193	.244	.292	9	-0.070
.35	-0.364	-0.333	-0.364	-0.937	-0.885	-0.806	.231	.101	.171	.180	.244	.250	10	-0.079
.40	-0.300	-0.329	-0.352	-0.943	-0.896	-0.816	.201	.076	.148	.176	.234	.221	11	-0.196
.45	-0.253	-0.328	-0.349	-0.940	-0.899	-0.825	.093	.033	.131	.192	.238	.215	12	-0.194
.50	-0.291	-0.319	-0.358	-0.911	-0.907	-0.832	.110	.093	.096	.227	.240	.197	13	-0.198
.55	-0.285	-0.313	-0.362	-0.829	-0.913		.122	.086	.107	.271	.259			
.60	-0.293	-0.316	-0.335	-0.717	-0.917	-0.850	.106	.070	.197	.247	.253	.175		
.65	-0.276	-0.304	-0.349	-0.555	-0.915		.098	.035	.271	.289	.242			
.68												.223		
.70	-0.305	-0.308	-0.343	-0.478	-0.932	-0.871	.100	.014	.323	.324		.139		
.73												.194		
.75	-0.289	-0.299	-0.331		-0.946		.146	.240	.368			.181		
.77												.297		
.79												.297		
.80	-0.311	-0.309					.549	-0.779	.237	.342				
.82												.148		
.83														
.85	-0.318	-0.334					.549	-0.942	.361	.433				
.87												.213		
.88	-0.397											.385		
.89														
.90	-0.543	-0.558					.547	-0.927	.375	.415				
.91	-0.575						.521	-0.721	.522	.266				
.92	-0.512	-0.567	-0.558				.311		.261	.212				
.93														
.94	-0.497													
.95	-0.554													
.96	-0.477													
.97	-0.474													
.98	-0.457	-0.541												

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TABLE IV.- Concluded
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 4

$\frac{x}{c}$	$\delta = 7.5$ $M = 1.05$						$\alpha = 10.0$ $P_{t,f}/P_{\infty} = 4.7$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.50		
.01	-0.194	-0.746					+0.321	+0.505						
.03	-0.172	-0.766	-0.556	-0.803			+0.389	+0.503	+0.342	+0.317	+0.292	+0.304		
.05	-0.282	-0.869					+0.408	+0.486		+0.295				
.08	-0.265	-0.917	-0.988				+0.423	+0.508	+0.285					
.10	-0.263	-0.917		-0.839	-0.789		+0.421			+0.272				
.15	-0.188	-0.724	-1.038		-0.810		+0.429	+0.400	+0.237		+0.253			
.20	-0.220						+0.418							
.25	-0.178	-0.136	-0.683	-0.922	-0.925	-0.770	+0.352	+0.175	+0.227	+0.210	+0.260	+0.300		
.30	-0.239	-0.252	-0.441	-0.927	-0.054	-0.787	+0.253	+0.119	+0.198	+0.191	+0.245	+0.294	9	+0.77
.35	-0.368	-0.344	-0.367	-0.935	-0.883	-0.803	+0.228	+0.096	+0.166	+0.177	+0.245	+0.249	10	+0.86
.40	-0.299	-0.338	-0.355	-0.944	-0.893	-0.814	+0.206	+0.071	+0.144	+0.175	+0.232	+0.220	11	+0.214
.45	-0.254	-0.335	-0.351	-0.941	-0.898	-0.82	+0.096	+0.085	+0.128	+0.192	+0.237	+0.214	12	+0.214
.50	-0.293	-0.329	-0.361	-0.913	-0.906	-0.829	+0.108	+0.080	+0.095	+0.226	+0.239	+0.197	13	+0.217
.55	-0.288	-0.324	-0.363	-0.833	-0.912		+0.119	+0.079	+0.109	+0.269	+0.259			
.60	-0.296	-0.324	-0.356	-0.724	-0.914	-0.846	+0.106	+0.062	+0.201	+0.266	+0.251	+0.175		
.65	-0.282	-0.316	-0.350	-0.563	-0.913		+0.090	+0.030	+0.271	+0.288	+0.243			
.68												+0.223		
.70	-0.307	-0.320	-0.347	-0.484	-0.932	-0.269	+0.098	+0.111	+0.319	+0.321		+0.140		
.73												+0.193		
.75	-0.294	-0.309	-0.333		-0.944		+0.144	+0.236	+0.365			+0.181		
.77												+0.295		
.79												+0.295		
.80	-0.315	-0.318					+0.239	+0.329					+0.145	+0.097
.82												+0.506		
.83												+0.404	+0.210	
.85	-0.319	-0.344	-0.610	-0.415	-0.941		+0.353	+0.427					+0.110	
.87												+0.421	+0.329	+0.132
.88												+0.381		
.89														
.90	-0.541	-0.561		-0.325	-0.928	-0.575	+0.416							
.91	-0.584						+0.377		+0.267				+0.078	+0.024
.92	-0.512							+0.317		+0.083				
.93	-0.570	-0.560					+0.314							
.94	-0.498							+0.261		+0.199				
.95	-0.572						+0.256							
.96	-0.478						+0.176						+0.078	
.97							+0.191						+0.044	
.98	-0.459	-0.548		-0.460	-0.334					+0.127				
											+0.083			

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TABLE V

		$\delta = 0.0$					$M = 0.80$					$\alpha = 0.0$		$P_{t,j}/P_{\infty} = 1.0$					Nacelle	
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p						
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90								
.01	.005	-0.015					.078	-0.032					1	.761						
.03	.056	-0.030	-0.070	-0.119			.039	-0.041	-0.061	-0.130	-0.182		2	-0.72						
.05	.031	-0.024			-0.117	-0.061		.023	-0.038	-0.189	-0.153	-0.161	3	-0.183						
.06	.016	-0.033	-0.058				.001	-0.032	-0.070				4	-0.92						
.10	.000	-0.037			-0.125	-0.060		.008		-0.356			5	-0.124						
.15	.013	-0.047	-0.074		-0.057		.043	-0.056	-0.106		-0.174		6	-0.186						
.20	-0.026						.013						7	-0.157						
.25	-0.023	-0.035	-0.095	-0.082	-0.045	-0.052	.034	-0.051	-0.157	-0.168	-0.153	-0.085	8	-0.077						
.30	-0.036	-0.057	-0.094	-0.043	-0.072	-0.025	.038	-0.080	-0.176	-0.166	-0.085	-0.102	9	-0.149						
.35	-0.058	-0.088	-0.094	-0.079	-0.082	-0.073	.053	-0.114	-0.168	-0.135	-0.090	-0.098	10	-0.188						
.40	-0.051	-0.095	-0.096	-0.102	-0.092	-0.081	.078	-0.123	-0.153	-0.132	-0.124	-0.115	11	-0.064						
.45	-0.078	-0.058	-0.094	-0.091	-0.077	-0.051	.115	-0.142	-0.135	-0.073	-0.127	-0.095	12	-0.064						
.50	-0.075	-0.094	-0.090	-0.097	-0.085	-0.086	.108	-0.129	-0.129	-0.058	-0.120	-0.094	13	-0.065						
.55	-0.071	-0.094	-0.093	-0.081	-0.077		.150	-0.132	-0.112	-0.055	-0.092									
.60	-0.081	-0.087	-0.062	-0.051	-0.069	-0.086	.123	-0.118	-0.094	-0.086	-0.062	-0.043								
.65	-0.074	-0.082	-0.065	-0.051	-0.074		.098	-0.125	-0.086	-0.076	-0.053									
.68																				
.70	-0.070	-0.070	-0.056	-0.069	-0.059	-0.047	.082	-0.090	-0.071	-0.064										
.75	-0.063	-0.049	-0.040		-0.042		.113	-0.083	-0.051											
.77																				
.79																				
.80	-0.057	-0.042			-0.021	-0.010	.061	-0.053												
.82																				
.83																				
.85	-0.039	-0.044					.034	-0.047												
.87																				
.88																				
.90																				
.91	-0.006																			
.93	-0.006	-0.026																		
.95	-0.007	-0.004																		
.97																				
.98																				
	.015																			
		$\delta = 0.0$					$M = 0.80$					$\alpha = 0.0$		$P_{t,j}/P_{\infty} = 3.0$					Nacelle	
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Orifice number	C_p						
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90								
.01	.086	-0.022					.078	-0.029					1	.701						
.03	.052	-0.039	-0.079	-0.141			.038	-0.042	-0.063	-0.119	-0.178		2	-0.162						
.05	.023	-0.030			-0.136	-0.073		.022	-0.040	-0.179	-0.156	-0.137	3	-0.167						
.08	.008	-0.038	-0.049				.001	-0.032	-0.075				4	-0.089						
.10	-0.007	-0.042			-0.142	-0.084		.010			-0.339		5	-0.121						
.15	.006	-0.055	-0.084				.044	-0.055	-0.112		-0.174		6	-0.177						
.20	-0.030						.016						7	-0.150						
.25	-0.030	-0.041	-0.104	-0.094	-0.084	-0.064	.037	-0.054	-0.156	-0.168	-0.100	-0.078	8	-0.073						
.30	-0.045	-0.064	-0.105	-0.073	-0.077	-0.074	.040	-0.081	-0.175	-0.149	-0.081	-0.096	9	-0.150						
.33	-0.059	-0.095	-0.105	-0.091	-0.094	-0.080	.054	-0.119	-0.171	-0.135	-0.086	-0.097	10	-0.168						
.40	-0.060	-0.100	-0.107	-0.111	-0.096	-0.092	.080	-0.125	-0.156	-0.128	-0.117	-0.107	11	-0.058						
.45	-0.084	-0.093	-0.104	-0.101	-0.084	-0.070	.120	-0.142	-0.140	-0.069	-0.113	-0.089	12	-0.049						
.50	-0.081	-0.102	-0.100	-0.106	-0.094	-0.093	.105	-0.129	-0.130	-0.051	-0.106	-0.091	13	-0.050						
.55	-0.077	-0.101	-0.103	-0.059	-0.086		.152	-0.134	-0.113	-0.045	-0.076									
.60	-0.089	-0.094	-0.072	-0.089	-0.080	-0.071	.120	-0.113	-0.090	-0.070	-0.047	-0.068								
.65	-0.078	-0.087	-0.072	-0.061	-0.087		.098	-0.124	-0.081	-0.059	-0.034	-0.044								
.68																				
.70	-0.076	-0.074	-0.066	-0.077	-0.070	-0.054	.081	-0.084	-0.064	-0.049		-0.046								
.75	-0.069	-0.056	-0.052			-0.051	.112	-0.080	-0.047			-0.030								
.77												-0.019								
.79																				
.80	-0.063	-0.047					.031	-0.011	-0.057	-0.053										
.82																				
.83																				
.85	-0.043	-0.052					.013		-0.039	-0.048										
.87									-0.042	-0.013	-0.001									
.88									-0.040											
.90										-0.034	-0.012	-0.016								
.91	-0.013						.011		-0.001	-0.012										
.93	-0.013						.013													
.95	-0.012	-0.003					.031		-0.019											
.97										-0.042	-0.028	-0.037								
.98										-0.007										

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TABLE V. - Continued

TABLE V. - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

 $\delta = 0.0$ $M = 0.80$ $\alpha = 5^{\circ}$ $P_{t,i}/P_{\infty} = 1.0$

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle		
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p	
.01	-156	-725	-668	-832	-985	-960	-664	-242	-255	-425	-182	-128	1	-0.36	
.03	-150	-742	-668	-832	-985	-960	-664	-226	-224	-426	-182	-128	2	-0.07	
.05	-155	-735	-635	-708	-800	-800	-664	-192	-169	-416	-143	-218	3	-0.121	
.08	-154	-315	-708	-800	-800	-800	-664	-169	-156	-185	-	-	4	-0.030	
.10	-156	-150	-	-951	-	-	-664	-148	-	-	-263	-	5	-0.040	
.15	-156	-196	-766	-	-268	-	-664	-152	-097	-138	-	-060	6	-0.137	
.20	-153	-	-	-	-	-	-664	-139	-	-	-	-	7	-0.143	
.25	-175	-198	-493	-510	-213	-695	-	-097	-082	-111	-048	-072	8	-0.088	
.30	-188	-214	-284	-428	-234	-712	-	-095	-052	-076	-063	-045	9	-0.168	
.35	-228	-237	-141	-400	-235	-759	-	-082	-021	-014	-027	-039	10	-0.117	
.40	-204	-235	-124	-379	-228	-751	-	-046	-011	-021	-059	-003	11	-0.062	
.45	-215	-218	-143	-342	-207	-724	-	-018	-008	-046	-006	-008	12	-0.063	
.50	-198	-213	-159	-318	-204	-703	-	-010	-007	-055	-025	-017	13	-0.066	
.55	-184	-199	-174	-272	-185	-	-	-013	-032	-045	-024	-001	-	-	
.60	-185	-180	-143	-248	-173	-577	-	-020	-034	-033	-015	-022	14	-0.116	
.65	-160	-166	-145	-186	-163	-	-	-019	-051	-033	-015	-023	-	-	
.68	-	-	-	-	-	-	-	-	-	-	-	-003	-	-	
.70	-149	-146	-151	-204	-137	-343	-	-023	-033	-026	-309	-	-001	-	
.73	-	-	-	-	-	-	-	-	-054	-038	-019	-	-014	-	-026
.75	-133	-118	-108	-	-	-	-	-	-	-	-	-	-005	-	-009
.77	-	-	-	-	-150	-	-	-	-	-	-	-	-010	-	-
.79	-	-	-	-	-140	-	-	-	-	-	-	-	-017	-	-
.80	-116	-101	-	-	-	-095	-	-017	-023	-	-	-	-029	-	-010
.82	-	-	-092	-	-	-	-	-	-013	-	-	-	-	-	-
.83	-	-	-074	-	-112	-	-	-	-006	-	-017	-	-	-	-
.85	-089	-092	-	-	-	-083	-	-027	-022	-	-	-	-031	-	-
.87	-	-	-078	-054	-	-	-	-	-021	-003	-022	-	-	-	-
.88	-	-	-070	-	-	-	-	-	-025	-	-	-025	-012	-	-
.90	-	-	-032	-	-	-054	-	-	-023	-004	-	-025	-	-	-
.91	-	-	-051	-	-058	-	-	-	-035	-015	-020	-	-	-	-
.93	-	-	-044	-012	-	-	-	-	-041	-026	-	-020	-	-	-
.95	-046	-028	-	-	-024	-	-	-	-011	-026	-	-019	-	-	-
.96	-	-	-	-	-031	-	-	-	-	-026	-	-	-	-	-
.97	-	-	-007	-	-	-	-	-	-	-011	-	-	-	-	-
.98	-	-	-016	-	-	-	-	-	-	-	-	-	-	-	-

$\frac{x}{c}$	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C_p
.01	-156	-711	-653	-816	-950	-953	-653	-251	-274	-433	-199	-232	1	-0.274
.03	-161	-723	-653	-816	-950	-953	-653	-218	-230	-426	-158	-232	2	-0.057
.05	-170	-699	-569	-869	-950	-953	-653	-224	-175	-426	-158	-232	3	-0.080
.08	-179	-274	-491	-	-	-	-	-181	-165	-195	-	-	4	-0.039
.10	-183	-145	-	-935	-	-250	-	-160	-	-274	-	-077	5	-0.038
.15	-142	-186	-747	-	-252	-	-	-163	-106	-148	-	-	6	-0.104
.20	-181	-	-	-	-	-	-	-148	-	-	-	-	7	-0.105
.25	-160	-184	-461	-490	-202	-689	-	-108	-092	-124	-032	-092	8	-0.038
.30	-174	-203	-255	-468	-220	-716	-	-104	-060	-089	-046	-086	9	-0.127
.35	-194	-224	-126	-1380	-224	-754	-	-093	-031	-028	-051	-069	10	-0.069
.40	-189	-221	-114	-361	-215	-749	-	-060	-018	-005	-040	-036	11	-0.040
.45	-203	-205	-132	-326	-197	-725	-	-027	-004	-027	-028	-022	12	-0.043
.50	-185	-199	-150	-305	-196	-701	-	-021	-004	-035	-046	-015	13	-0.039
.55	-169	-186	-163	-259	-175	-	-	-001	-019	-026	-050	-031	-	-
.60	-173	-167	-132	-237	-165	-572	-	-007	-020	-012	-017	-049	14	-0.030
.65	-147	-153	-134	-175	-156	-	-	-005	-039	-016	-018	-043	-	-
.68	-	-	-	-	-	-	-	-	-	-	-	-023	-	-
.70	-140	-135	-120	-195	-131	-332	-	-008	-019	-001	-018	-	15	-0.012
.73	-	-	-	-	-	-	-	-037	-022	-006	-	-	16	-
.75	-121	-109	-098	-	-109	-	-	-	-	-	-	-	17	-
.77	-	-	-	-	-130	-	-	-	-	-	-	-	18	-
.79	-	-	-	-	-	-	-	-	-	-	-	-	19	-
.80	-105	-091	-	-083	-	-087	-	-001	-005	-	-	-030	20	-0.001
.82	-	-	-066	-102	-	-075	-	-	-	-007	-008	-024	21	-0.031
.83	-	-	-067	-046	-	-075	-	-011	-	-014	-	-021	22	-
.85	-078	-081	-	-066	-	-087	-	-	-007	-008	-014	-021	23	-
.87	-	-	-012	-046	-	-075	-	-	-012	-	-	-	24	-
.88	-	-	-060	-	-	-	-	-	-	-015	-	-024	25	-0.026
.90	-	-	-025	-	-039	-072	-	-	-013	-	-022	-	26	-0.026
.91	-	-	-041	-	-047	-	-	-	-023	-	-021	-	27	-
.93	-	-	-032	-	-005	-	-	-	-029	-	-021	-	28	-
.95	-036	-019	-	-	-017	-	-	-	-	-015	-	-021	29	-
.96	-	-	-	-	-022	-	-	-	-	-015	-	-028	30	-
.97	-	-	-014	-	-	-	-	-	-	-033	-	-021	31	-
.98	-	-	-008	-	-	-	-	-	-	-006	-	-	32	-

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TABLE V.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

$\delta = 0.0$						$M = 0.80$						$\alpha = 5.0$						$P_{t,j}/P_{\infty} = 5.1$	
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle						
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp	1	-0.251			
.01	-0.148	-0.700	-0.647	-0.808	-0.858	-0.942	-0.650	-0.256	-0.278	-0.268	-0.238	-0.207	1	-0.251					
.05	-0.160	-0.715	-0.647	-0.808	-0.858	-0.942	-0.650	-0.219	-0.233	-0.268	-0.238	-0.207	2	-0.040					
.08	-0.165	-0.690	-0.647	-0.808	-0.858	-0.942	-0.650	-0.208	-0.280	-0.248	-0.211	-0.238	3	-0.047					
.10	-0.175	-0.646	-0.680	-0.928	-0.240	-0.243	-0.183	-0.169	-0.199	-0.276	-0.238	-0.084	4	-0.024					
.15	-0.137	-0.141	-0.734	-0.243	-0.167	-0.108	-0.153	-0.154	-0.084	-0.027	-0.027	-0.027	5	-0.027					
.20	-0.176	-0.180	-0.734	-0.243	-0.167	-0.108	-0.153	-0.154	-0.084	-0.027	-0.027	-0.027	6	-0.027					
.25	-0.156	-0.181	-0.444	-0.480	-0.195	-0.686	-0.115	-0.094	-0.130	-0.025	-0.103	-0.152	7	-0.027					
.30	-0.173	-0.197	-0.244	-0.400	-0.214	-0.712	-0.110	-0.045	-0.092	-0.039	-0.094	-0.125	8	-0.026					
.35	-0.190	-0.220	-0.120	-0.371	-0.216	-0.749	-0.099	-0.036	-0.033	-0.041	-0.076	-0.095	9	-0.027					
.40	-0.185	-0.238	-0.109	-0.354	-0.210	-0.747	-0.064	-0.026	-0.002	-0.030	-0.044	-0.061	10	-0.024					
.45	-0.196	-0.200	-0.124	-0.319	-0.190	-0.721	-0.035	-0.007	-0.021	-0.035	-0.028	-0.056	11	-0.023					
.50	-0.182	-0.195	-0.145	-0.297	-0.187	-0.699	-0.027	-0.008	-0.025	-0.057	-0.026	-0.040	12	-0.020					
.55	-0.164	-0.181	-0.157	-0.251	-0.170	-0.570	-0.005	-0.013	-0.016	-0.062	-0.039	-0.030	13	-0.020					
.60	-0.169	-0.163	-0.125	-0.230	-0.156	-0.570	-0.001	-0.015	-0.004	-0.027	-0.052	-0.028	-0.028	-0.028					
.65	-0.143	-0.150	-0.127	-0.168	-0.149	-0.001	-0.030	-0.002	-0.026	-0.045	-0.028	-0.028	-0.028	-0.028					
.68	-0.170	-0.136	-0.113	-0.187	-0.125	-0.327	-0.001	-0.011	-0.068	-0.028	-0.030	-0.012	-0.012	-0.012					
.73	-0.118	-0.101	-0.091	-0.135	-0.102	-0.027	-0.017	-0.012	-0.023	-0.011	-0.034	-0.011	-0.011	-0.011					
.77	-0.122	-0.122	-0.077	-0.002	-0.006	-0.001	-0.006	-0.001	-0.023	-0.032	-0.001	-0.001	-0.001	-0.001					
.80	-0.100	-0.087	-0.075	-0.005	-0.006	-0.001	-0.006	-0.001	-0.008	-0.014	-0.026	-0.030	-0.006	-0.006					
.82	-0.072	-0.072	-0.057	-0.005	-0.006	-0.001	-0.006	-0.001	-0.006	-0.017	-0.022	-0.030	-0.006	-0.006					
.85	-0.064	-0.039	-0.069	-0.055	-0.006	-0.006	-0.006	-0.001	-0.006	-0.017	-0.022	-0.030	-0.006	-0.006					
.87	-0.057	-0.018	-0.032	-0.074	-0.020	-0.020	-0.020	-0.027	-0.026	-0.027	-0.027	-0.001	-0.001	-0.001					
.91	-0.039	-0.017	-0.041	-0.017	-0.010	-0.017	-0.022	-0.017	-0.024	-0.024	-0.024	-0.029	-0.029	-0.029					
.95	-0.031	-0.017	-0.021	-0.017	-0.017	-0.017	-0.022	-0.011	-0.026	-0.026	-0.026	-0.029	-0.029	-0.029					
.98	-0.002	-0.002	-0.002	-0.002	-0.002	-0.002	-0.002	-0.002	-0.002	-0.002	-0.002	-0.002	-0.002	-0.002	-0.002				
$\delta = 0.0$						$M = 0.80$						$\alpha = 5.0$						$P_{t,j}/P_{\infty} = 6.7$	
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle						
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp	1	-0.243			
.01	-0.152	-0.700	-0.648	-0.818	-0.869	-0.959	-0.653	-0.256	-0.280	-0.265	-0.235	-0.208	1	-0.243					
.03	-0.162	-0.715	-0.648	-0.818	-0.869	-0.959	-0.653	-0.222	-0.235	-0.265	-0.249	-0.215	2	-0.032					
.05	-0.145	-0.497	-0.280	-0.687	-0.934	-0.252	-0.174	-0.209	-0.180	-0.165	-0.145	-0.240	3	-0.029					
.08	-0.174	-0.280	-0.137	-0.327	-0.327	-0.174	-0.184	-0.170	-0.194	-0.279	-0.166	-0.168	4	-0.028					
.10	-0.179	-0.137	-0.177	-0.743	-0.248	-0.154	-0.154	-0.111	-0.149	-0.089	-0.154	-0.154	5	-0.020					
.15	-0.136	-0.177	-0.177	-0.177	-0.177	-0.177	-0.154	-0.116	-0.127	-0.025	-0.106	-0.154	6	-0.078					
.20	-0.173	-0.178	-0.178	-0.178	-0.178	-0.178	-0.154	-0.111	-0.127	-0.042	-0.099	-0.127	7	-0.047					
.25	-0.154	-0.178	-0.156	-0.468	-0.196	-0.694	-0.116	-0.096	-0.127	-0.025	-0.106	-0.154	8	-0.083					
.30	-0.170	-0.196	-0.259	-0.411	-0.216	-0.714	-0.111	-0.068	-0.094	-0.042	-0.099	-0.127	9	-0.081					
.35	-0.188	-0.219	-0.128	-0.385	-0.221	-0.758	-0.102	-0.036	-0.031	-0.041	-0.082	-0.089	10	-0.001					
.40	-0.185	-0.216	-0.114	-0.364	-0.214	-0.761	-0.066	-0.027	-0.006	-0.030	-0.050	-0.057	11	-0.147					
.45	-0.195	-0.198	-0.131	-0.327	-0.194	-0.724	-0.032	-0.010	-0.020	-0.035	-0.037	-0.053	12	-0.155					
.50	-0.180	-0.194	-0.145	-0.304	-0.194	-0.710	-0.031	-0.010	-0.028	-0.060	-0.029	-0.036	13	-0.122					
.55	-0.164	-0.178	-0.160	-0.257	-0.173	-0.580	-0.010	-0.009	-0.018	-0.064	-0.039	-0.024	-0.024	-0.024					
.60	-0.168	-0.161	-0.131	-0.235	-0.160	-0.580	-0.004	-0.012	-0.032	-0.027	-0.049	-0.024	-0.024	-0.024					
.65	-0.141	-0.147	-0.139	-0.173	-0.153	-0.006	-0.029	-0.003	-0.022	-0.035	-0.013	-0.013	-0.013	-0.013					
.68	-0.134	-0.126	-0.119	-0.191	-0.129	-0.335	-0.002	-0.009	-0.005	-0.015	-0.019	-0.009	-0.009	-0.009					
.73	-0.115	-0.101	-0.094	-0.138	-0.128	-0.027	-0.013	-0.006	-0.009	-0.001	-0.023	-0.023	-0.023	-0.023					
.77	-0.102	-0.087	-0.078	-0.064	-0.098	-0.081	-0.005	-0.006	-0.003	-0.001	-0.012	-0.029	-0.029	-0.029					
.80	-0.074	-0.074	-0.061	-0.044	-0.074	-0.059	-0.005	-0.005	-0.004	-0.001	-0.015	-0.029	-0.029	-0.029					
.82	-0.055	-0.025	-0.025	-0.046	-0.037	-0.075	-0.005	-0.005	-0.004	-0.011	-0.019	-0.026	-0.005	-0.005					
.85	-0.031	-0.013	-0.028	-0.002	-0.016	-0.021	-0.010	-0.010	-0.010	-0.020	-0.030	-0.030	-0.030	-0.030					
.87	-0.017	-0.001	-0.023	-0.017	-0.023	-0.017	-0.017	-0.017	-0.017	-0.030	-0.030	-0.030	-0.030	-0.030					

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TABLE V.- Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

$\frac{x}{c}$	$\delta = 0.0$ $M = 0.80$ $\alpha = 10.0$ $P_{t,j}/P_{\infty} = 1.0$						Nacelle Orifice number: C_p	
	0.19	0.32	0.53	0.65	0.74	0.80		
.01	-0.495	-1.091						
.03	-0.515	-1.154	-0.978	-1.079				
.05	-0.464	-1.225		-1.091	-0.874	-0.735		
.08	-0.460	-1.436		-0.997				
.10	-0.457	-1.481		-0.921	-0.991			
.15	-0.392	-0.862	-1.055		-1.010			
.20	-0.371							
.25	-0.331	-0.256	-1.294	-0.585	-0.376	-0.797		
.30	-0.334	-0.305	-1.343	-0.507	-0.279	-0.917		
.35	-0.345	-0.333	-1.234	-0.431	-0.200	-0.858		
.40	-0.339	-0.322	-0.992	-0.355	-0.309	-0.844		
.45	-0.330	-0.297	-0.611	-0.368	-0.308	-0.811		
.50	-0.295	-0.267	-0.223	-0.458	-0.319	-0.855		
.55	-0.260	-0.225	-0.018	-0.473	-0.312			
.60	-0.234	-0.186	.008	-0.452	-0.323	-0.829		
.65	-0.192	-0.165	-0.030	-0.425	-0.326			
.68								
.70	-0.176	-0.155	-0.169	-0.466	-0.318	-0.752		
.73								
.75	-0.159	-0.150	-0.213		-0.303			
.77								
.79								
.80	-0.158	-0.155						
.82								
.83								
.85	-0.137	-0.159						
.87	-0.146	-0.208	-0.322		-0.272			
.88	-0.139							
.90								
.91	-0.115							
.93	-0.103	-0.147		-0.274		-0.517		
.95	-0.101	-0.083						
.96								
.97								
.98	-0.070							

$\frac{x}{c}$	$\delta = 0.0$ $M = 0.80$ $\alpha = 10.0$ $P_{t,j}/P_{\infty} = 3.1$						Nacelle Orifice number: C_p	
	0.19	0.32	0.52	0.65	0.74	0.80		
.01	-0.488	-1.083						
.03	-0.508	-1.124	-0.964	-1.084				
.05	-0.457	-1.213		-1.099	-0.870	-0.741		
.08	-0.453	-1.421	-0.998					
.10	-0.435	-1.446		-0.930	-0.994			
.15	-0.334	-0.850	-1.057		-1.014			
.20								
.25	-0.335	-0.252	-1.300	-0.589	-0.385	-0.802		
.30	-0.331	-0.298	-1.343	-0.511	-0.286	-0.840		
.35	-0.342	-0.328	-1.237	-0.437	-0.207	-0.840		
.40	-0.337	-0.328	-0.997	-0.364	-0.314	-0.851		
.45	-0.325	-0.293	-0.639	-0.371	-0.311	-0.829		
.50	-0.296	-0.265	-0.226	-0.443	-0.323	-0.853		
.55	-0.261	-0.216	-0.020	-0.476	-0.316			
.60	-0.230	-0.178	.012	-0.484	-0.326	-0.830		
.65	-0.186	-0.156	-0.088	-0.425	-0.329			
.68								
.70	-0.173	-0.146	-0.163	-0.468	-0.324	-0.757		
.73								
.75	-0.154	-0.144	-0.211		-0.310			
.77								
.79								
.80	-0.153	-0.148						
.82								
.83								
.85	-0.134	-0.153						
.87	-0.139	-0.209	-0.327		-0.280			
.88	-0.133							
.90								
.91	-0.108							
.93	-0.096	-0.146		-0.277				
.95	-0.099	-0.077						
.96								
.97								
.98	-0.054							

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TABLE V - Cont.nued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

TABLE V - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

X C	$\delta = 0.0$ $M = 0.90$						$\alpha = 0.0$ $P_{t,f}/P_{\infty} = 1.0$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.017	-.011					.097	-.014					1	.504
.03	.071	-.023	-.083	-.167	-.174	-.077	.026	.058	-.027	-.043	-.066	-.201	2	-.214
.05	.042	-.020					.037	-.026		-.129	-.179	-.208	3	-.292
.08	.026	-.027	-.069				.015	-.018	-.062				4	-.091
.10	.006	-.034					.006						5	-.136
.15	.019	-.049	-.091	-.263	-.073	-.075	-.026	-.042	-.102				6	-.235
.20	-.023	-.049					.000						7	-.237
.25	-.019	-.042	-.130	-.103	-.062	-.059	.023	-.043	-.157	-.138	-.108	-.097	8	-.078
.30	-.038	-.068	-.141	-.084	-.093	-.079	.029	-.078	-.216	-.15C	-.085	-.103	9	-.256
.35	-.062	-.101	-.126	-.102	-.108	-.093	.042	-.123	-.289	-.136	-.097	-.108	10	-.300
.40	-.061	-.114	-.120	-.122	-.1C8	-.097	.079	-.135	-.242	-.14C	-.148	-.125	11	-.054
.45	-.093	-.110	-.119	-.116	-.100	-.073	.128	-.165	-.102	-.080	-.189	-.105	12	-.057
.50	-.093	-.126	-.117	-.121	-.113	-.102	.117	-.163	-.136	-.075	-.196	-.099	13	-.052
.55	-.094	-.125	-.121	-.105	-.103	-.103	.189	-.190	-.126	-.102	-.138			
.60	-.109	-.115	-.088	-.106	-.092	-.082	.263	-.150	-.119	-.154	-.075	-.056		
.65	-.099	-.108	-.089	-.074	-.098	-.140	.144	-.136	-.110	-.047				
.68														
.70	-.096	-.094	-.080	-.091	-.078	-.055	.105	-.116	-.118	-.074				
.73														
.75	-.085	-.073	-.062				.142	-.112	-.069					
.77														
.79														
.80	-.075	-.061					.027	.000	.065	-.079				
.82														
.83														
.85	-.056	-.064												
.87	-.043	-.015	-.003											
.88	-.037													
.90														
.92														
.93														
.95	-.011	-.001												
.96														
.97														
.98														
X C	$\delta = 0.0$ $M = 0.90$						$\alpha = 0.0$ $P_{t,f}/P_{\infty} = 3.0$						Nacelle Orifice number	C_p
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90		
.01	.093	-.024					.081	-.036					1	.421
.03	.057	-.039	-.097	-.183	-.102	-.046	.038	-.051	-.068	-.096	-.235	-.212	2	-.230
.05	.028	-.036					.021	-.047		-.161	-.208	-.212	3	-.264
.08	.009	-.044	-.09C				.004	-.038	-.085				4	-.100
.10	-.009	-.051					.013			-.330			5	-.156
.15	.004	-.065	-.113		-.094		.047	-.063	-.127		-.227		6	-.250
.20	-.038						.019						7	-.215
.25	-.036	-.061	-.160	-.125	-.082	-.095	.043	-.061	-.185	-.172	-.131	-.101	8	-.084
.30	-.055	-.088	-.166	-.107	-.114	-.109	.049	-.098	-.241	-.145	-.103	-.109	9	-.234
.35	-.079	-.121	-.152	-.124	-.131	-.120	.061	-.144	-.314	-.154	-.112	-.113	10	-.280
.40	-.078	-.134	-.144	-.146	-.131	-.125	.099	-.154	-.284	-.157	-.160	-.136	11	-.077
.45	-.111	-.129	-.143	-.138	-.122	-.102	.148	-.183	-.125	-.095	-.195	-.122	12	-.077
.50	-.111	-.146	-.146	-.145	-.137	-.133	.139	-.189	-.149	-.086	-.181	-.122	13	-.066
.55	-.114	-.146	-.143	-.128	-.125	-.114	.213	-.211	-.143	-.106	-.118			
.60	-.129	-.137	-.109	-.131	-.118	-.114	.186	-.173	-.131	-.139	-.069	-.044		
.65	-.119	-.129	-.112	-.099	-.123	-.159	.159	-.159	-.139	-.096	-.044	-.057		
.68														
.70	-.116	-.116	-.102	-.117	-.104	-.085	.121	-.129	-.108	-.064				
.73														
.75	-.106	-.094	-.086		-.081		.157	-.119	-.066					
.77														
.79														
.80	-.097	-.083	-.073		-.056	-.014	.078	-.083						
.82														
.83														
.85	-.077	-.086	-.059	-.046	-.032		.072	-.059						
.87	-.086	-.040	-.040	-.025			.036	-.036	-.024					
.88	-.058													
.90														
.91	-.039													
.93	-.034													
.95	-.031	-.024												
.96														
.97														
.98	-.012													

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TABLE V - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

$\delta = 0.0$						$M = 0.90$						$\alpha = 0.0$						$p_{t,j}/p_{\infty} = 5.0$											
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle																
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp															
.01	.117	.002					.106	.008					1	.385															
.03	.080	-.014	-.077	-.170			.066	-.023	-.043	-.068	-.194		2	-.184															
.05	.050	-.012		-.179	-.077	-.025	.044	-.020		-.131	-.174	-.157	3	-.163															
.08	.032	-.020		-.066			.023	-.013	-.057				4	.007															
.10	.015	-.027			-.268	-.372	.013			-.300			5	-.121															
.15	.029	-.041	-.087		-.069		.020	-.034	-.099		-.292		6	-.197															
.20	-.012						.006						7	-.126															
.25	-.039	-.035	-.123	-.110	-.057	-.071	.014	-.035	-.156	-.134	-.086	-.072	8	.017															
.30	-.029	-.059	-.137	-.095	-.090	-.081	.022	-.072	-.210	-.125	-.071	-.087	9	-.147															
.35	-.052	-.031	-.123	-.110	-.105	-.024	.035	-.113	-.282	-.080	-.003	-.001	10	-.145															
.40	-.053	-.107	-.118	-.128	-.106	-.098	.070	-.225	-.234	-.120	-.110	-.045																	
.45	-.084	-.102	-.118	-.122	-.096	-.083	.119	-.156	-.006	-.066	-.142	-.095	12	.057															
.50	-.085	-.119	-.115	-.119	-.111	-.107	.104	-.160	-.125	-.056	-.119	-.096	13	.053															
.55	-.086	-.117	-.117	-.102	-.102	-.102	.180	-.178	-.113	-.066	-.075																		
.60	-.123	-.108	-.082	-.082	-.081	-.086	.150	-.139	-.098	-.090	-.042	-.067																	
.65	-.092	-.101	-.088	-.074	-.099	-.124	.124	-.132	-.100	-.062	-.027																		
.68																													
.70	-.087	-.099	-.076	-.091	-.080	-.058	.090	-.097	-.069	-.043	-.044	-.047																	
.73																													
.75	-.078	-.065	-.062		-.055		.123	-.086	-.043		-.034	-.028																	
.77																													
.79																													
.80	-.069	-.056					.034	.005	-.047	-.056		-.027																	
.82																													
.83																													
.85	-.050	-.059																											
.87																													
.88																													
.89																													
.90																													
.91																													
.93																													
.95	-.006	.002					.034		-.046	-.027		-.039	.050																
.96																													
.97																													
.98																													
$\delta = 0.0$						$M = 0.90$						$\alpha = 0.0$						$p_{t,j}/p_{\infty} = 7.0$											
x/c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle						Orifice number										
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.35	0.74	0.90	Orifice number	Cp					1	.312									
.01	.095	-.021					.084	.033					2	-.185															
.03	.056	-.038	-.099	-.180			.043	-.047	-.062	-.090	-.219		3	-.124															
.05	.028	-.054					.024	-.044	-.044	-.155	-.198	-.161	4	.076															
.08	.010	-.044					.000	-.038	-.082		-.324		5	-.139															
.10	-.008						.045	-.059	-.121		-.213		6	.195															
.15	-.062	-.111					.018						7	-.091															
.20	-.038						.041	-.058	-.177	-.156	-.118	-.102	8	.072															
.25	-.035	-.059	-.145	-.122	-.078	-.104	.046	-.096	-.234	-.146	-.089	-.124	9	-.122															
.30	-.053	-.083	-.160	-.101	-.111	-.116	.059	-.140	-.357	-.148	-.097	-.135	10	.161															
.35	-.076	-.115	-.145	-.122	-.128	-.130	.096	-.150	-.255	-.144	-.126	-.160	11	.104															
.40	-.078	-.131	-.142	-.142	-.130	-.131	.144	-.183	-.119	-.080	-.137	-.146	12	.153															
.45	-.105	-.127	-.138	-.135	-.119	-.121	.144	-.183	-.119	-.080	-.137	-.146	13	.143															
.50	-.108	-.143	-.135	-.143	-.135	-.139	.135	-.185	-.144	-.265	-.121	-.146	14	.143															
.55	-.110	-.142	-.137	-.127	-.126	-.128	.205	-.203	-.130	-.367	-.090		15																
.60	-.126	-.134	-.105	-.127	-.118	-.118	.175	-.160	-.111	-.289	-.072	-.110	16																
.65	-.116	-.127	-.111	-.100	-.124		.1-6	-.151	-.104	-.071	-.077		17																
.68													18	-.107															
.70	-.113	-.112	-.101	-.117	-.106	-.085	.113	-.118	-.078	-.078	-.078	-.078	19	.078															
.73													20	.101															
.75	-.104	-.090	-.085		-.075	-.081	.145	-.106	-.069		-.099	-.084	21																
.77													22																
.79													23																
.80	-.094	-.081					.073	-.056	-.018	-.068	-.079		24																
.82													25																
.83													26																
.85	-.076	-.084					.057	-.036	-.034	-.068	-.076	-.071	27	-.049	-.030	-.024													
.87													28																
.88													29																
.89													30																
.90													31																
.91													32																
.93													33																
.95	-.027	-.024					.019	-.019	-.011	-.074	-.055	-.014	34																
.96													35																
.97													36																
.98													37																

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TABLE V.- Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

		$\delta = 0.0$						$M = 0.90$						$\alpha = 5.0$		$P_{t,j}/P_{\infty} = 1.0$						
X	c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						α	δ	Nacelle						
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			Orifice number	C_p					
.01	-.092	-.651						.265	.279							1	-.022					
.03	-.119	-.685	-.632	-.804				.230	.244	.258	.451	.187				2	-.101					
.05	-.133	-.612		-.588	-.983	-.677		.216	.188		.441	.145	.220			3	-.139					
.08	-.146	-.208		-.658				.191	.177		.192					4	-.076					
.10	-.155	-.138			-.1c63	-.445		.172			.273					5	-.051					
.15	-.116	-.178		-.705		-.277		.176	.114	.152		.059				6	-.156					
.20	-.159							.161								7	-.161					
.25	-.145	-.179	-.490	-.483	-.223	-.695		.121	.098	.129	-.081	.071	.144			8	-.059					
.30	-.165	-.209	-.384	-.394	-.246	-.722		.112	.069	.086	-.081	.069	.120			9	-.188					
.35	-.192	-.240	-.336	-.375	-.251	-.743		.104	.036	.005	-.077	.043	.092			10	-.123					
.40	-.199	-.250	-.148	-.365	-.237	-.747		.065	.023	.035	-.058	-.001	.057			11	-.052					
.45	-.219	-.240	-.125	-.337	-.216	-.721		.029	-.002	-.060	.016	-.024	.051			12	-.048					
.50	-.216	-.270	-.163	-.321	-.216	-.702		.021	-.007	-.080	-.027	-.027	.041			13	-.051					
.55	-.224	-.279	-.193	-.280	-.196			.007	-.035	-.048	-.010	-.010										
.60	-.250	-.207	-.158	-.256	-.177	-.622		.020	-.043	-.046	-.029	-.023	.032									
.65	-.206	-.172	-.158	-.198	-.166			.023	-.062	-.065	-.021	-.032										
.68													.015									
.70	-.167	-.161	-.137	-.206	-.137	-.448		-.029	-.040	-.036	-.005		.016									
.73																						
.75	-.145	-.133	-.109					-.066	-.046	-.022												
.77																						
.79																						
.80	-.126	-.108																				
.82																						
.83																						
.85	-.087	-.090																				
.87	-.070	-.041	-.689																			
.88	-.062																					
.90																						
.91	-.041																					
.93	-.030	-.003																				
.95	-.036	-.016																				
.96																						
.97																						
.98																						
		$\delta = 0.0$						$M = 0.90$						$\alpha = 5.0$		$P_{t,j}/P_{\infty} = 3.0$						
X	c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						α	δ	Nacelle						
		0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90			Orifice number	C_p					
.01	-.100	-.657		-.644	-.817			.258	.272		.449	.177				1	-.037					
.03	-.129	-.673	-.644	-.817	-.681	-.1015	-.713	.223	.235	.248	.449	.177				2	-.093					
.05	-.141	-.626						.259	.180		.439	.134	.216			3	-.119					
.08	-.154	-.209		-.672				.186	.171	.185						4	-.056					
.10	-.165	-.144			-.1078	-.475		.162			.273					5	-.063					
.15	-.169	-.185		-.722		-.286		.169	.197	.143		.052				6	-.151					
.20								.155								7	-.148					
.25	-.154	-.188	-.505	-.499	-.233	-.740		.114	.092	.121	-.085	.072	.142			8	-.055					
.30	-.175	-.217	-.397	-.409	-.256	-.759		.108	.061	.080	-.085	.067	.113			9	-.175					
.35	-.197	-.250	-.347	-.391	-.263	-.793		.097	.029	.002	-.078	.046	.097			10	-.101					
.40	-.205	-.257	-.161	-.380	-.248	-.796		.059	.016	.041	-.056	.007	.047			11	-.043					
.45	-.227	-.248	-.133	-.352	-.228	-.763		.024	-.004	-.064	-.020	-.011	.040			12	-.053					
.50	-.225	-.280	-.174	-.335	-.230	-.755		.015	-.007	-.034	-.033	-.016	.023			13	-.043					
.55	-.234	-.289	-.202	-.293	-.211			.010	-.038	-.047	-.023	-.008										
.60	-.258	-.215	-.173	-.272	-.192	-.676		.022	-.045	-.035	-.010	-.036	.014									
.65	-.217	-.182	-.172	-.214	-.182			.025	-.062	-.037	-.003	-.039										
.68																						
.70	-.176	-.170	-.152	-.224	-.152	-.502		-.028	-.040	-.021	-.015		.010									
.73																						
.75	-.154	-.143	-.126					-.061	-.041	-.006												
.77																						
.79																						
.80	-.135	-.117																				
.82																						
.83																						
.85	-.099	-.102																				
.87	-.080	-.058																				
.88	-.073																					
.90																						
.91																						
.93																						
.95																						
.96																						
.97																						
.98																						

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TABLE V - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle		
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	Cp	
	-0.377	-0.360	-0.908	-1.048	-1.080	-0.765	-0.396	-0.379	-0.399	-0.375	-0.371	-0.621	-0.357	1	-0.06
-0.03	-0.400	-0.002	-0.908	-1.048	-1.080	-0.765	-0.403	-0.328	-0.374	-0.316	-0.349	-0.684	-0.319	2	-0.044
-0.05	-0.392	-1.144	-0.933	-1.080	-1.007	-0.765	-0.403	-0.328	-0.374	-0.316	-0.349	-0.684	-0.319	3	-0.125
-0.08	-0.385	-1.222	-0.933	-1.145	-1.050	-0.764	-0.345	-0.249	-0.344	-0.249	-0.328	-0.543	-0.215	4	-0.098
-0.10	-0.377	-1.237	-0.994	-1.145	-1.064	-0.764	-0.350	-0.225	-0.350	-0.225	-0.308	-0.485	-0.208	5	-0.008
-0.15	-0.362	-0.884	-0.994	-1.145	-1.064	-0.764	-0.261	-0.194	-0.252	-0.194	-0.242	-0.402	-0.184	6	-0.092
-0.20	-0.337	-0.205	-0.994	-1.145	-1.064	-0.764	-0.261	-0.194	-0.252	-0.194	-0.242	-0.402	-0.184	7	-0.137
-0.25	-0.297	-0.252	-1.193	-0.775	-0.841	-0.822	-0.240	-0.162	-0.208	-0.175	-0.223	-0.395	-0.208	8	-0.076
-0.30	-0.218	-0.305	-1.240	-0.634	-0.560	-0.821	-0.240	-0.162	-0.208	-0.175	-0.223	-0.395	-0.218	9	-0.203
-0.35	-0.134	-0.360	-1.257	-0.504	-0.374	-0.842	-0.194	-0.154	-0.194	-0.154	-0.113	-0.031	-0.102	10	-0.074
-0.40	-0.058	-0.586	-1.220	-0.355	-0.331	-0.842	-0.180	-0.122	-0.180	-0.122	-0.061	-0.067	-0.071	11	-0.052
-0.45	-0.038	-0.560	-0.865	-0.288	-0.334	-0.842	-0.154	-0.114	-0.154	-0.114	-0.039	-0.105	-0.059	12	-0.087
-0.50	-0.026	-0.407	-0.545	-0.294	-0.358	-0.837	-0.158	-0.076	-0.158	-0.076	-0.056	-0.087	-0.063	13	-0.091
-0.55	-0.017	-0.425	-0.181	-0.353	-0.343	-0.828	-0.172	-0.056	-0.172	-0.056	-0.037	-0.036	-0.080	14	-0.086
-0.60	-0.010	-0.366	-0.079	-0.420	-0.341	-0.828	-0.078	-0.017	-0.078	-0.017	-0.018	-0.029	-0.074	15	-0.049
-0.65	-0.008	-0.197	-0.172	-0.405	-0.346	-0.828	-0.078	-0.017	-0.078	-0.017	-0.018	-0.029	-0.074	16	-0.040
-0.68	-0.009	-0.150	-0.217	-0.454	-0.389	-0.791	-0.051	-0.024	-0.013	-0.026	-0.026	-0.055	-0.055	17	-0.040
-0.73	-0.014	-0.174	-0.262	-0.422	-0.321	-0.791	-0.041	-0.008	-0.010	-0.026	-0.033	-0.058	-0.058	18	-0.058
-0.77	-0.016	-0.262	-0.411	-0.300	-0.739	-0.739	-0.024	-0.009	-0.012	-0.015	-0.031	-0.048	-0.048	19	-0.098
-0.79	-0.018	-0.230	-0.349	-0.276	-0.633	-0.633	-0.007	-0.014	-0.015	-0.017	-0.017	-0.028	-0.028	20	-0.092
-0.80	-0.018	-0.197	-0.290	-0.274	-0.379	-0.276	-0.024	-0.009	-0.012	-0.015	-0.017	-0.005	-0.009	21	-0.092
-0.82	-0.017	-0.217	-0.274	-0.379	-0.276	-0.276	-0.007	-0.014	-0.015	-0.017	-0.017	-0.025	-0.025	22	-0.092
-0.83	-0.017	-0.197	-0.260	-0.349	-0.276	-0.276	-0.007	-0.014	-0.015	-0.017	-0.017	-0.025	-0.025	23	-0.092
-0.85	-0.017	-0.217	-0.274	-0.379	-0.276	-0.276	-0.007	-0.014	-0.015	-0.017	-0.017	-0.025	-0.025	24	-0.092
-0.87	-0.017	-0.197	-0.260	-0.349	-0.276	-0.276	-0.007	-0.014	-0.015	-0.017	-0.017	-0.025	-0.025	25	-0.092
-0.88	-0.018	-0.230	-0.349	-0.276	-0.633	-0.633	-0.007	-0.014	-0.015	-0.017	-0.017	-0.025	-0.025	26	-0.092
-0.90	-0.018	-0.195	-0.228	-0.228	-0.228	-0.228	-0.037	-0.047	-0.042	-0.009	-0.024	-0.058	-0.059	27	-0.059
-0.91	-0.018	-0.195	-0.228	-0.228	-0.228	-0.228	-0.037	-0.047	-0.042	-0.009	-0.024	-0.058	-0.059	28	-0.059
-0.93	-0.018	-0.195	-0.228	-0.228	-0.228	-0.228	-0.037	-0.047	-0.042	-0.009	-0.024	-0.058	-0.059	29	-0.059
-0.95	-0.018	-0.195	-0.228	-0.228	-0.228	-0.228	-0.037	-0.047	-0.042	-0.009	-0.024	-0.058	-0.059	30	-0.059
-0.96	-0.018	-0.268	-0.268	-0.268	-0.268	-0.268	-0.044	-0.024	-0.024	-0.024	-0.024	-0.024	-0.024	31	-0.024
-0.97	-0.018	-0.157	-0.157	-0.157	-0.157	-0.157	-0.044	-0.024	-0.024	-0.024	-0.024	-0.024	-0.024	32	-0.024
-0.98	-0.018	-0.072	-0.072	-0.072	-0.072	-0.072	-0.044	-0.024	-0.024	-0.024	-0.024	-0.024	-0.024	33	-0.024

TABLE V. - Continued WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5																
x c	$\delta = 0.0$ $M = 0.90$						$\alpha = 10.0$ $P_{t,j}/P_{\infty} = 5.0$			Nacelle Orifice number	C_p					
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.367	-0.950									1 -0.032 2 -0.024 3 -0.040 4 -0.026 5 -0.031 6 -0.055 7 -0.071 8 -0.039 9 -0.125 10 -0.006					
.03	-0.391	-0.985	-0.883	-1.030	-1.068	-1.023	-0.786	-0.400	-0.382	-0.377	-0.626	-0.363				
.05	-0.383	-1.080									-0.405	-0.337				
.08	-0.375	-1.193	-0.921					-0.377	-0.325	-0.357	-0.688	-0.323				
.10	-0.370	-1.209									-0.348	-0.547				
.15	-0.284	-0.865	-0.984					-0.346	-0.260	-0.335						
.20	-0.331							-0.350								
.25	-0.295	-0.239	-1.180	-0.782	-0.873	-0.847	-0.265	-0.235	-0.314	-0.094	-0.221	-0.246				
.30	-0.311	-0.290	-1.227	-0.644	-0.580	-0.860	-0.255	-0.207	-0.269	-0.054	-0.199	-0.227				
.35	-0.327	-0.342	-1.247	-0.512	-0.369	-0.849	-0.242	-0.176	-0.184	-0.038	-0.171	-0.179				
.40	-0.331	-0.370	-1.225	-0.357	-0.319	-0.873	-0.197	-0.168	-0.124	-0.047	-0.124	-0.144				
.45	-0.363	-0.365	-0.867	-0.261	-0.321	-0.875	-0.187	-0.139	-0.074	-0.118	-0.098	-0.133				
.50	-0.357	-0.394	-0.551	-0.277	-0.346	-0.877	-0.156	-0.129	-0.055	-0.129	-0.086	-0.110				
.55	-0.371	-0.420	-0.189	-0.335	-0.336	-0.166	-0.093	-0.055	-0.119	-0.092						
.60	-0.396	-0.384	-0.077	-0.403	-0.335	-0.863	-0.118	-0.073	-0.058	-0.071						
.65	-0.349	-0.128	-0.168	-0.401	-0.341	-0.087	-0.040	-0.048	-0.065	-0.085						
.68																
.70	-0.214	-0.136	-0.212	-0.446	-0.395	-0.811	-0.064	-0.047	-0.046	-0.054						
.73																
.75	-0.160	-0.159	-0.256					-0.038	-0.032	-0.040						
.77																
.79																
.80	-0.175	-0.184														
.82																
.85	-0.271	-0.287	-0.271	-0.378												
.85	-0.172	-0.208														
.87	-0.188	-0.257	-0.345					-0.018	-0.016	-0.001						
.88	-0.177							-0.002								
.90																
.91	-0.149							-0.004								
.93	-0.132	-0.195							-0.028	-0.013	-0.030	-0.090				
.95	-0.129	-0.105							-0.028	-0.034	-0.066	-0.071				
.96																
.97																
.98																
x c	$\delta = 0.0$ $M = 0.90$						$\alpha = 10.0$ $P_{t,j}/P_{\infty} = 7.0$			Nacelle Orifice number	C_p					
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.378	-0.961														
.03	-0.400	-0.991	-0.898	-1.042												
.05	-0.319	-1.085	-1.075	-1.029	-0.790											
.08	-0.384	-1.202	-0.931													
.10	-0.377	-1.222														
.15	-0.292	-0.884	-0.991													
.20	-0.339															
.25	-0.305	-0.242	-1.190	-0.785	-0.889	-0.851	-0.266	-0.236	-0.316	-0.094	-0.229	-0.240				
.30	-0.319	-0.292	-1.237	-0.647	-0.600	-0.860	-0.255	-0.209	-0.270	-0.056	-0.210	-0.222				
.35	-0.331	-0.346	-1.256	-0.504	-0.382	-0.870	-0.244	-0.180	-0.183	-0.041	-0.177	-0.172				
.40	-0.336	-0.373	-1.232	-0.366	-0.328	-0.874	-0.199	-0.164	-0.123	-0.051	-0.134	-0.134				
.45	-0.365	-0.368	-0.878	-0.268	-0.332	-0.873	-0.189	-0.139	-0.076	-0.123	-0.108	-0.126				
.50	-0.362	-0.395	-0.575	-0.281	-0.358	-0.871	-0.162	-0.128	-0.056	-0.136	-0.096	-0.102				
.55	-0.377	-0.424	-0.208	-0.339	-0.344	-0.857	-0.121	-0.075	-0.064	-0.081	-0.093					
.60	-0.400	-0.375	-0.088	-0.411	-0.344	-0.857	-0.092	-0.059	-0.056	-0.126	-0.093					
.65	-0.347	-0.128	-0.176	-0.408	-0.347			-0.092	-0.039	-0.031	-0.072	-0.069				
.68																
.70	-0.213	-0.140	-0.218	-0.454	-0.343	-0.808	-0.063	-0.050	-0.051	-0.049						
.75	-0.163	-0.162	-0.263					-0.068	-0.031	-0.040						
.77																
.79																
.80	-0.180	-0.188														
.82																
.83																
.85	-0.177	-0.213														
.87	-0.190	-0.266	-0.354					-0.022	-0.002	-0.013	-0.010	-0.007				
.88	-0.181							-0.001	-0.003	-0.006	-0.001					
.90																
.91	-0.153							-0.007	-0.003							
.93	-0.136	-0.203							-0.038	-0.021	-0.026	-0.034				
.95	-0.133	-0.109							-0.029	-0.039	-0.084	-0.097				
.96																
.97																
.98																
x c	$\delta = 0.0$ $M = 0.90$						$\alpha = 10.0$ $P_{t,j}/P_{\infty} = 7.0$			Nacelle Orifice number	C_p					
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90				
.01	-0.399	-0.389														

TABLE V.- Continued

X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	.099	.064					.089	.055					1	.318
.03	.078	.050	-.002	-.072			.064	.043	-.049	.048	-.209		2	-.147
.05	.069	.056		-.077	-.103		.062	.050		-.012	-.239	-.177	3	-.322
.08	.067	.052	.001		-.181	-.105	.059	.062	.025				4	-.318
.10	.061	.047			-.181	-.105	.058			-.124			5	-.149
.15	.085	.036	-.029			-.089	.071	.046	-.033		-.231		6	-.187
.20	.055						.081						7	-.314
.25	.055	.023	-.047	-.107	-.074	-.137	.067	.050	-.058	-.352	-.263	-.190	8	-.277
.30	.042	.003	-.079	-.101	-.114	-.154	.063	.015	-.093	-.340	-.065	-.215	9	-.313
.35	.030	-.021	-.143	-.120	-.142	-.173	.055	-.036	-.168	-.150	-.054	-.272	10	-.303
.40	.030	-.046	-.157	-.137	-.148	-.179	.010	-.058	-.266	-.168	-.078	-.300	11	-.283
.45	-.017	-.034	-.140	-.134	-.139	-.157	-.033	-.082	-.310	-.075	-.126	-.316	12	-.279
.50	-.024	-.066	-.147	-.154	-.155	-.199	-.045	-.086	-.257	-.034	-.177	-.351	13	-.270
.55	-.031	-.100	-.153	-.150	-.159		-.089	-.118	-.114	-.057	-.224			
.60	-.071	-.145	-.145	-.153	-.147	-.213	-.087	-.162	-.098	-.137	-.273	-.377		
.65	-.107	-.139	-.150	-.136	-.175		-.127	-.240	-.092	-.215	-.343			
.68											-.339			
.70	-.142	-.127	-.139	-.151	-.182	-.236	-.162	-.226	-.131	-.298		-.306		
.73												-.282		
.75	-.124	-.111	-.116		-.181		-.272	-.138	-.188			-.088		
.77												-.298		
.79												-.245		
.80	-.125	-.115		-.155		-.168	-.194	-.109	-.088			-.052	-.063	
.82												-.309		
.83												-.298	-.065	
.85	-.119	-.147		-.142	-.143	-.159	-.095	-.076					-.049	
.87				-.135	-.133	-.141		-.215						
.88				-.129				-.173						
.90				-.105	-.114	-.121		-.207						
.91				-.107	-.105			-.033						
.93				-.080	-.083	-.076		-.227	-.051					
.95								-.115					-.018	
.97								-.023						
.98								-.029						
								-.057						

TABLE V.- Continued

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TABLE V. - Continued

GOVERNMENT OF
CANADA

TABLE V - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

TABLE V. - Continued

TABLE V.- Continued

TABLE V - Continued

WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	.073	-.028					.062	-.047					1	.217
.03	.044	-.039	-.037	-.116			.029	-.057	.032	.049	-.227		2	-.154
.05	.021	-.037					.013	-.059		-.001	-.222	-.211	3	-.362
.08	.003	-.044	-.028				-.011	-.054	.009				4	-.308
.10	-.012	-.047					-.021						5	-.214
.15	.005	-.052	-.051				.050	-.047	-.035	-.093			6	-.219
.20	-.034						.023						7	-.317
.25	-.024	-.027	-.091	-.131	-.105	-.151	.040	-.012	-.059	-.322	-.287	-.162	8	-.281
.30	-.036	-.039	-.090	-.129	-.139	-.173	.012	-.016	-.106	-.419	-.156	-.175	9	-.359
.35	-.034	-.067	-.142	-.147	-.164	-.192	.007	-.054	-.137	-.285	-.102	-.236	10	-.307
.40	-.023	-.088	-.264	-.165	-.170	-.197	.022	-.086	-.247	-.193	-.123	-.255	11	-.269
.45	-.068	-.079	-.178	-.167	-.164	-.204	.061	-.113	-.313	-.132	-.130	-.276	12	-.273
.50	-.061	-.100	-.174	-.181	-.186	-.218	.065	-.129	-.350	-.080	-.164	-.308	13	-.259
.55	-.069	-.116	-.181	-.179	-.193		.120	-.144	-.288	-.080	-.189			
.60	-.097	-.146	-.149	-.180	-.189	-.231	.127	-.158	-.125	-.127	-.235	-.350		
.65	-.110	-.182	-.166	-.168	-.214		.130	-.225	-.142	-.195	-.308			
.68														
.70	-.151	-.180	-.169	-.190	-.219	-.257	.156	-.264	-.130	-.270		-.370		
.73	-.190	-.166	-.160				.275	-.277	-.161					
.75														
.77														
.79														
.80	-.171	-.162					.211	-.252	-.247	-.177				
.82														
.83														
.85	-.158	-.181	-.186	-.188			.180	-.137						
.87														
.88														
.90														
.91														
.93														
.95	-.122	-.143					.166							
.96														
.97														
.98														

X C	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
.01	.076	-.017					.046	-.040					1	.172
.03	.047	-.033	-.030	-.116	-.094	-.122	.034	-.050	.037	.053	-.213		2	-.162
.05	.025	-.030					.017	-.051		.001	-.222	-.208	3	-.303
.08	.008	-.037	-.026				-.004	-.046	.014				4	-.320
.10	-.008	-.039					.016	-.016					5	-.205
.15	.010	-.046	-.049				.040	-.038	-.034				6	-.222
.20	-.027						.014						7	-.312
.25	-.019	-.023	-.089	-.131	-.098	-.150	.032	-.022	-.087	-.327	-.287	-.162	8	-.292
.30	-.029	-.032	-.087	-.126	-.134	-.167	.062	-.005	-.102	-.413	-.158	-.174	9	-.351
.35	-.026	-.059	-.141	-.145	-.164	-.184	.016	-.048	-.134	-.272	-.096	-.218	10	-.305
.40	-.014	-.079	-.201	-.165	-.168	-.195	.012	-.079	-.243	-.197	-.12C	-.259	11	-.307
.45	-.060	-.072	-.176	-.161	-.161	-.200	.052	-.103	-.307	-.133	-.133	-.278	12	-.304
.50	-.055	-.094	-.170	-.178	-.183	-.216	.057	-.119	-.340	-.076	-.164	-.311	13	-.267
.55	-.063	-.108	-.178	-.176	-.192		.111	-.135	-.275	-.076	-.189			
.60	-.090	-.139	-.152	-.178	-.188	-.229	.117	-.149	-.128	-.138	-.236	-.351		
.65	-.103	-.176	-.161	-.167	-.208		.120	-.219	-.140	-.194	-.308			
.68														
.70	-.145	-.172	-.163	-.188	-.214	-.255	.148	-.255	-.129	-.268		-.374		
.73														
.75	-.181	-.157	-.156				.264	-.258	-.161					
.77														
.79														
.80	-.161	-.152					.205	-.244	-.226	-.169				
.82														
.83														
.85	-.149	-.172					.165	-.127						
.87														
.88														
.90														
.91														
.93														
.95	-.116	-.132					.182							
.96														
.97														
.98														

~~CONFIDENTIAL~~

TABLE V.- Continued

TABLE V - Continued

$$\delta = 0.0 \quad M = 1.05 \quad \alpha = 5.0 \quad p_{t,i}/p_m = 3.1$$

CONFIDENTIAL

TABLE V.- Continued

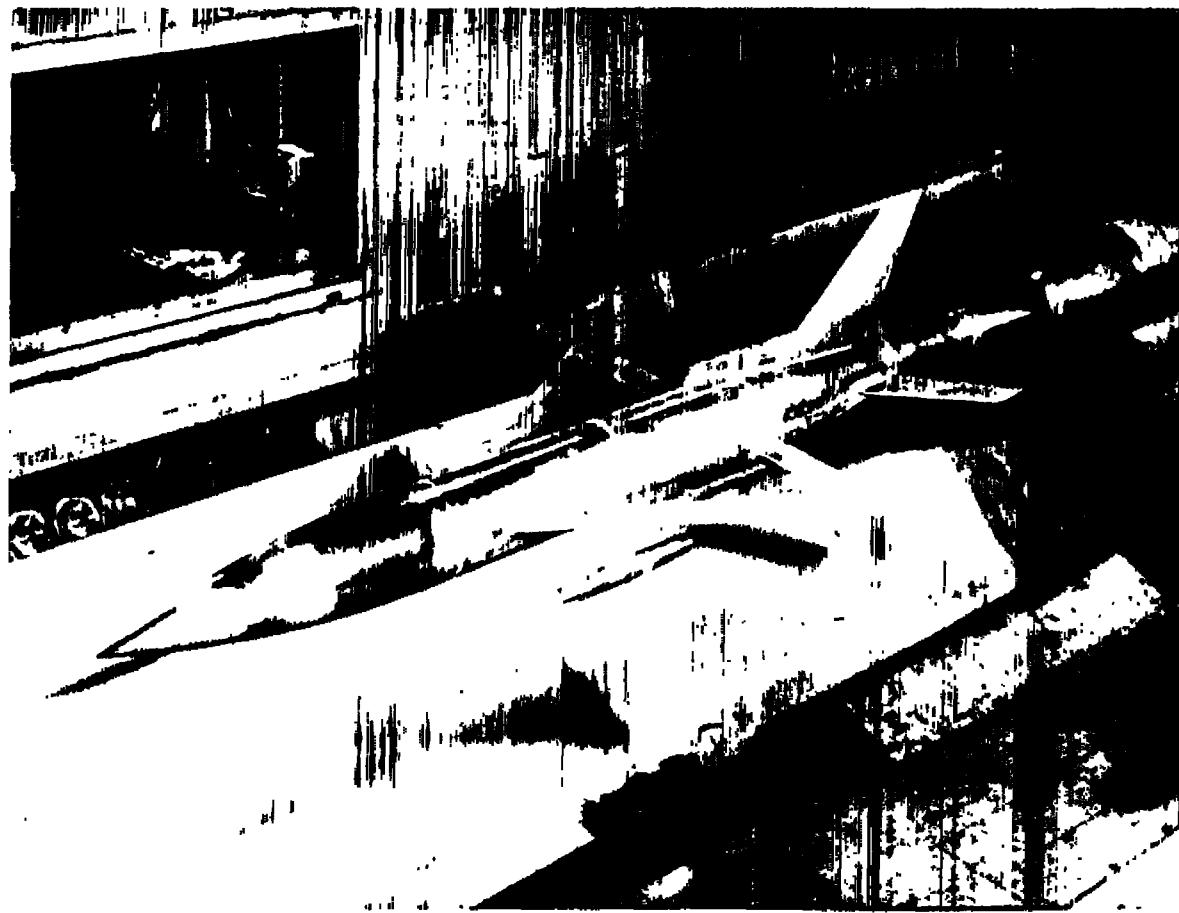
X c	Wing upper surface stations, fraction of semispan						Wing lower surface stations, fraction of semispan						Nacelle	
	0.19	0.32	0.53	0.65	0.74	0.90	0.19	0.32	0.53	0.65	0.74	0.90	Orifice number	C _p
	0.01	-0.071	-0.612	-0.547	-0.532	-0.631	-0.667	-0.780	-0.770	-0.203	-0.246	-0.289	-0.536	-0.133
-0.03	-0.111	-0.547	-0.532	-0.631	-0.667	-0.780	-0.770	-0.201	-0.227	-0.289	-0.536	-0.133	-0.178	-0.060
-0.05	-0.124	-0.360	-0.532	-0.631	-0.667	-0.780	-0.770	-0.182	-0.172	-0.229	-0.498	-0.153	-0.178	-0.231
-0.06	-0.133	-0.176	-0.532	-0.553	-0.631	-0.667	-0.780	-0.160	-0.174	-0.225	-0.359	-0.153	-0.178	-0.142
-0.08	-0.140	-0.182	-0.532	-0.577	-0.631	-0.659	-0.780	-0.141	-0.198	-0.202	-0.16	-0.16	-0.178	-0.097
-0.10	-0.107	-0.144	-0.532	-0.577	-0.631	-0.659	-0.780	-0.164	-0.198	-0.202	-0.16	-0.16	-0.178	-0.104
-0.15	-0.141	-0.141	-0.532	-0.577	-0.631	-0.659	-0.780	-0.191	-0.140	-0.211	-0.098	-0.202	-0.013	-0.254
-0.20	-0.114	-0.126	-0.345	-0.570	-0.329	-0.785	-0.191	-0.141	-0.169	-0.172	-0.073	-0.202	-0.013	-0.145
-0.30	-0.121	-0.148	-0.192	-0.470	-0.328	-0.781	-0.148	-0.111	-0.169	-0.172	-0.073	-0.202	-0.021	-0.291
-0.35	-0.130	-0.168	-0.199	-0.425	-0.329	-0.804	-0.151	-0.077	-0.072	-0.249	-0.068	-0.049	-0.10	-0.203
-0.40	-0.128	-0.196	-0.317	-0.390	-0.316	-0.811	-0.103	-0.077	-0.011	-0.103	-0.024	-0.106	-0.1	-0.052
-0.45	-0.166	-0.179	-0.391	-0.367	-0.298	-0.806	-0.092	-0.065	-0.090	-0.018	-0.025	-0.043	-0.12	-0.055
-0.50	-0.157	-0.190	-0.243	-0.362	-0.311	-0.799	-0.086	-0.064	-0.162	-0.066	-0.074	-0.041	-0.13	-0.003
-0.55	-0.159	-0.205	-0.247	-0.340	-0.309	-0.774	-0.099	-0.024	-0.158	-0.029	-0.124	-0.017	-0.200	-0.066
-0.60	-0.184	-0.224	-0.242	-0.336	-0.36	-0.774	-0.059	-0.020	-0.047	-0.055	-0.166	-0.200	-0.066	-0.066
-0.65	-0.194	-0.272	-0.265	-0.306	-0.323	-0.780	-0.013	-0.077	-0.027	-0.130	-0.200	-0.032	-0.066	-0.066
-0.68	-0.225	-0.321	-0.263	-0.340	-0.326	-0.693	-0.036	-0.110	-0.059	-0.207	-0.003	-0.023	-0.023	-0.023
-0.73	-0.289	-0.266	-0.250	-0.317	-0.323	-0.780	-0.117	-0.137	-0.116	-0.066	-0.051	-0.062	-0.005	-0.064
-0.77	-0.289	-0.266	-0.250	-0.317	-0.323	-0.780	-0.117	-0.137	-0.116	-0.066	-0.051	-0.062	-0.005	-0.064
-0.79	-0.289	-0.266	-0.250	-0.291	-0.317	-0.780	-0.117	-0.137	-0.116	-0.066	-0.051	-0.062	-0.005	-0.064
-0.80	-0.342	-0.259	-0.269	-0.264	-0.315	-0.780	-0.121	-0.056	-0.232	-0.136	-0.058	-0.062	-0.005	-0.064
-0.82	-0.245	-0.269	-0.264	-0.315	-0.308	-0.780	-0.062	-0.059	-0.088	-0.023	-0.008	-0.058	-0.062	-0.064
-0.83	-0.245	-0.264	-0.264	-0.315	-0.308	-0.780	-0.062	-0.059	-0.088	-0.023	-0.008	-0.058	-0.062	-0.064
-0.85	-0.245	-0.269	-0.272	-0.319	-0.308	-0.780	-0.062	-0.059	-0.088	-0.023	-0.008	-0.058	-0.062	-0.064
-0.87	-0.252	-0.272	-0.319	-0.308	-0.308	-0.780	-0.062	-0.059	-0.088	-0.023	-0.008	-0.058	-0.062	-0.064
-0.88	-0.245	-0.272	-0.319	-0.308	-0.308	-0.780	-0.062	-0.059	-0.088	-0.023	-0.008	-0.058	-0.062	-0.064
-0.90	-0.224	-0.278	-0.282	-0.303	-0.296	-0.780	-0.062	-0.024	-0.131	-0.012	-0.027	-0.022	-0.064	-0.064
-0.91	-0.221	-0.274	-0.278	-0.282	-0.296	-0.780	-0.062	-0.024	-0.160	-0.014	-0.027	-0.022	-0.064	-0.064
-0.95	-0.222	-0.203	-0.227	-0.280	-0.269	-0.780	-0.151	-0.138	-0.028	-0.013	-0.028	-0.022	-0.064	-0.064
-0.97	-0.172	-0.227	-0.280	-0.269	-0.269	-0.780	-0.085	-0.085	-0.028	-0.013	-0.028	-0.022	-0.064	-0.064

TABLE V. - Continued
WING AND NACELLE PRESSURE COEFFICIENTS - CONFIGURATION 5

S = 0.0 M = 1.05 $\alpha = 10.0$ $P_{t,j}/P_{\infty} = 1.0$							
X/C	Wing upper surface stations, fraction of semispan					Nacelle Orifice number	Cp
	0.19	0.32	0.53	0.65	0.74	0.90	
.01	-0.221	-0.759	-0.711	-0.790	-0.816	-0.942	-0.975
.03	-0.272	-0.779	-0.723	-0.803	-0.828	-0.954	-0.979
.05	-0.279	-0.773	-0.722	-0.724	-0.874	-0.942	-0.975
.10	-0.276	-0.719	-0.786	-0.893	-0.926	-0.975	-0.975
.15	-0.222	-0.716	-0.786	-0.893	-0.926	-0.975	-0.975
.20	-0.242						
.25	-0.211	-0.180	-0.895	-0.736	-0.626	-0.980	-0.980
.30	-0.218	-0.209	-0.929	-0.754	-0.540	-0.992	-0.992
.35	-0.226	-0.247	-0.923	-0.757	-0.540	-1.005	-0.995
.40	-0.227	-0.279	-0.956	-0.763	-0.560	-1.011	-0.996
.45	-0.257	-0.268	-0.964	-0.745	-0.576	-1.011	-0.996
.50	-0.253	-0.281	-0.874	-0.554	-0.597	-1.011	-0.996
.55	-0.253	-0.296	-0.610	-0.461	-0.568	-1.017	-0.996
.60	-0.275	-0.319	-0.231	-0.469	-0.523	-1.017	-0.996
.65	-0.287	-0.353	-0.193	-0.420	-0.468	-1.017	-0.996
.68							
.70	-0.321	-0.412	-0.245	-0.388	-0.411	-1.007	-0.996
.73	-0.365	-0.450	-0.294				
.75				-0.361			
.77				-0.341			
.79				-0.355			
.80	-0.442	-0.427		-0.323	-0.950	-0.946	-0.935
.82				-0.330			
.83				-0.317	-0.368		
.85	-0.451	-0.304		-0.290			
.87				-0.282	-0.346		
.88	-0.293			-0.280			
.90				-0.228	-0.259	-0.913	-0.913
.91	-0.277			-0.328			
.93	-0.270	-0.202		-0.258			
.95	-0.221	-0.230		-0.215	-0.069	-0.052	-0.052
.96				-0.316			
.97				-0.194			
.98				-0.149			
					-0.030	-0.069	
S = 0.0 M = 1.05 $\alpha = 10.0$ $P_{t,j}/P_{\infty} = 3.1$							
X/C	Wing upper surface stations, fraction of semispan					Nacelle Orifice number	Cp
	0.19	0.32	0.53	0.65	0.74	0.90	
.01	-0.229	-0.765					
.03	-0.278	-0.786	-0.723	-0.803	-0.828	-0.954	-0.979
.05	-0.284	-0.883					
.08	-0.288	-0.933	-0.733				
.10	-0.285	-0.928		-0.888	-0.939		
.15	-0.225	-0.717	-0.798	-0.956			
.20	-0.247						
.25	-0.214	-0.184	-0.908	-0.747	-0.646	-0.987	-0.987
.30	-0.222	-0.216	-0.940	-0.765	-0.547	-0.999	-0.999
.35	-0.232	-0.253	-0.940	-0.770	-0.549	-1.016	-0.999
.40	-0.228	-0.286	-0.867	-0.774	-0.570	-1.024	-0.999
.45	-0.263	-0.276	-0.876	-0.742	-0.585	-1.024	-0.999
.50	-0.258	-0.290	-0.881	-0.760	-0.604	-1.024	-0.999
.55	-0.258	-0.302	-0.605	-0.471	-0.577	-1.030	-0.999
.60	-0.279	-0.325	-0.230	-0.478	-0.529	-1.030	-0.999
.65	-0.293	-0.361	-0.201	-0.429	-0.473		
.68							
.70	-0.326	-0.421	-0.256	-0.397	-0.414	-1.019	-0.998
.73							
.75	-0.372	-0.459	-0.303		-0.383	-0.987	-0.985
.77							
.79				-0.352			
.80	-0.450	-0.427		-0.323	-0.975	-0.939	-0.926
.82				-0.342			
.83				-0.327	-0.382		
.85	-0.456	-0.313		-0.294			
.87				-0.303	-0.295		
.88				-0.302			
.90				-0.240	-0.337	-0.949	-0.949
.91	-0.287	-0.279	-0.214				
.93				-0.228	-0.242		
.95				-0.206	-0.326	-0.226	-0.070
.96							
.97							
.98				-0.167			-0.016

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TABLE V - Concluded



(a) Configuration 1: $\delta = 0^\circ$; $\frac{y_n}{b/2} = 0.41$; $\frac{x_e}{c} = 0.38$; $\frac{z}{d_j} = 4.0$. L-57-31

Figure 1.- Photographs of the model in the 16-foot transonic tunnel showing the various pylon-nacelle configurations tested.



(b) Configuration 2: $\delta = 0^\circ$; $\frac{y_n}{b/2} = 0.41$; $\frac{x_e}{c} = 0.69$; $\frac{z}{d_j} = 2.0$. L-57-69

Figure 1.- Continued.

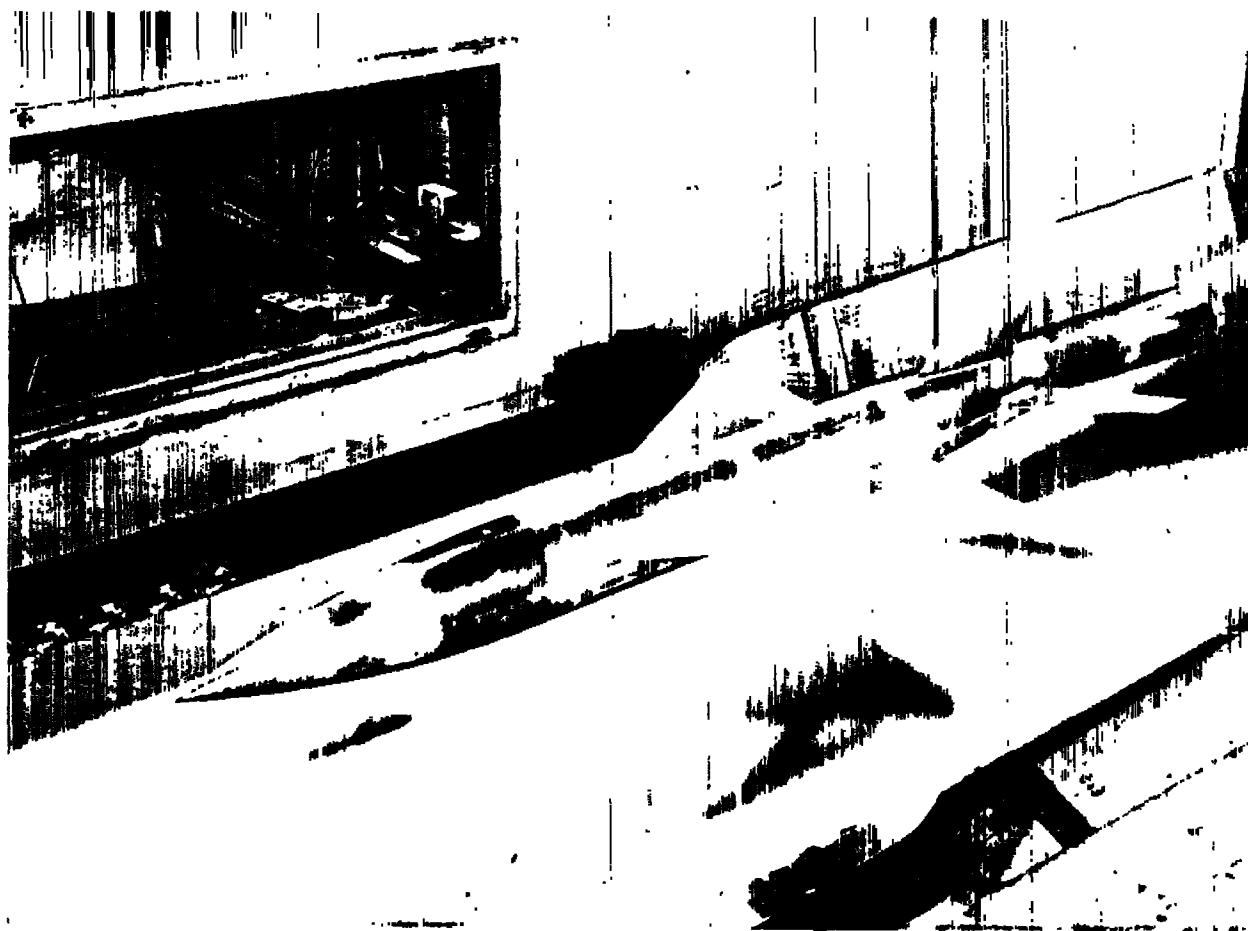


(c) Configuration 3: $\delta = 7.5^\circ$; $\frac{y_n}{b/2} = 0.41$; $\frac{x_e}{c} = 0.69$; $\frac{z}{d_j} = 2.0$. L-57-288



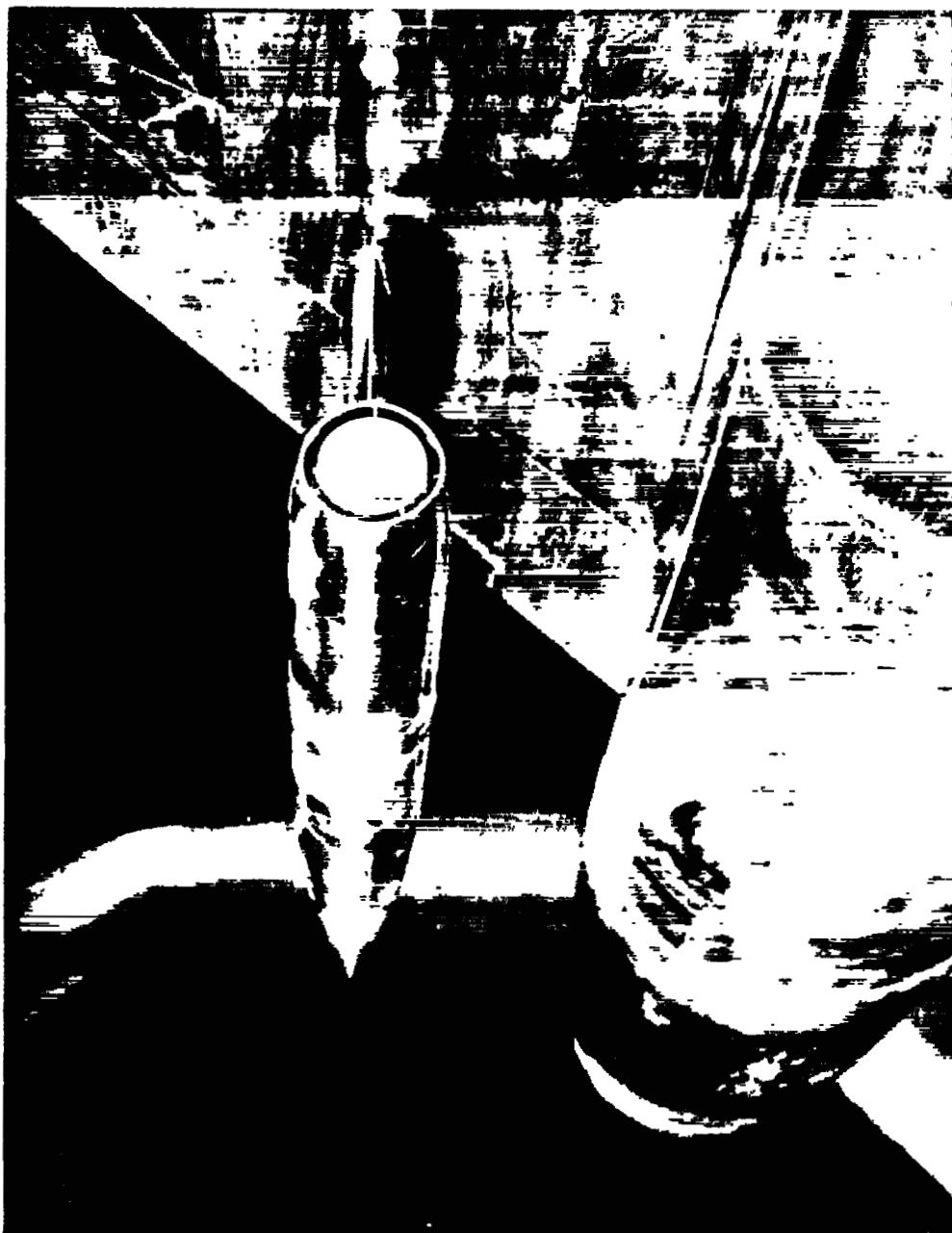
(d) Configuration 4: $\delta = 7.5^\circ$; $\frac{y_n}{b/2} = 0.41$; $\frac{x_e}{c} = 0.69$; $\frac{z}{d_j} = 1.23$. L-57-395

Figure 1.- Continued.



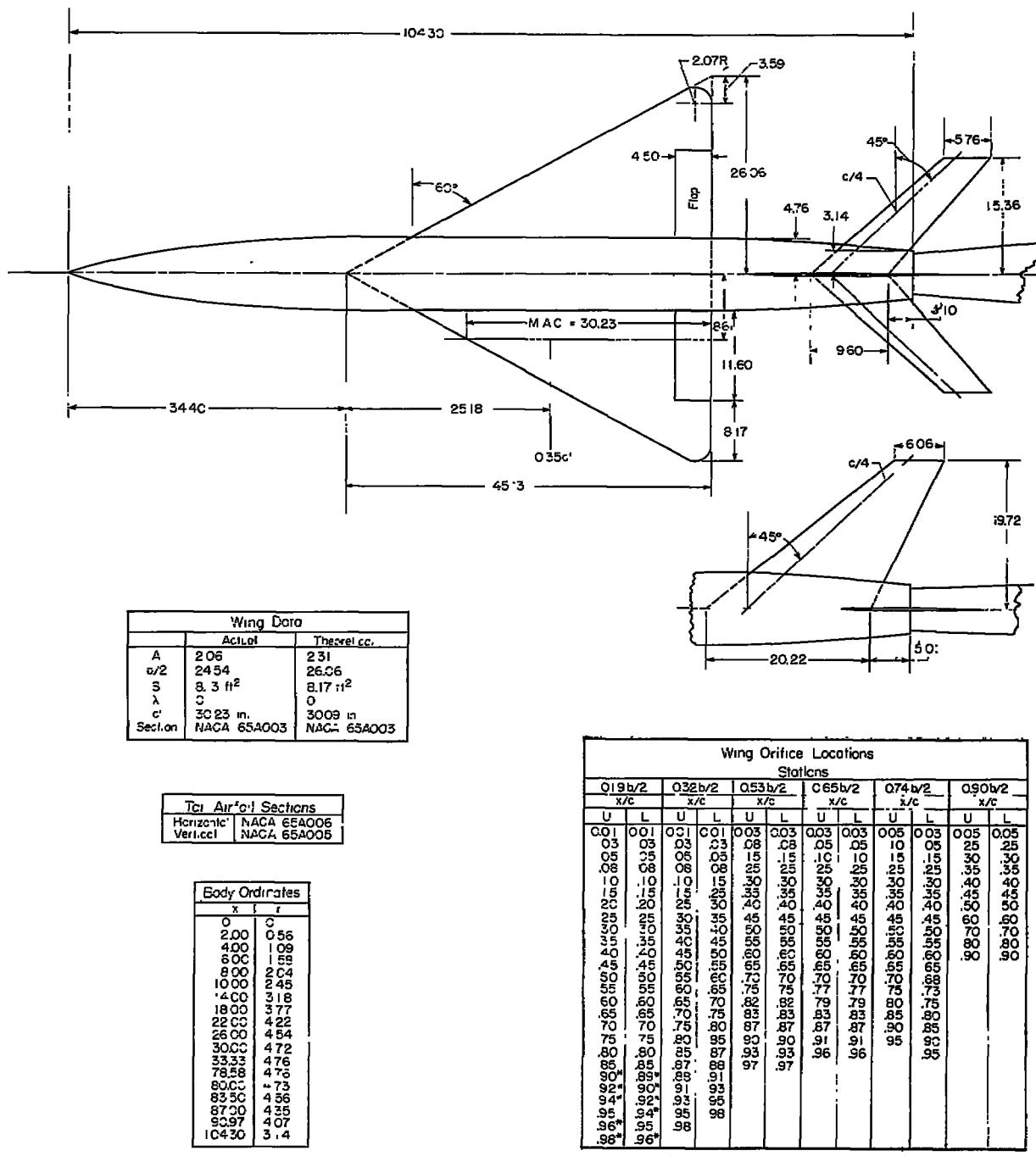
(e) Configuration 5: $\delta = 0^\circ$; $\frac{y_n}{b/2} = 0.69$; $\frac{x_e}{c} = 0.62$; $\frac{z}{d_j} = 4.0$. L-57-100

Figure 1.- Continued.



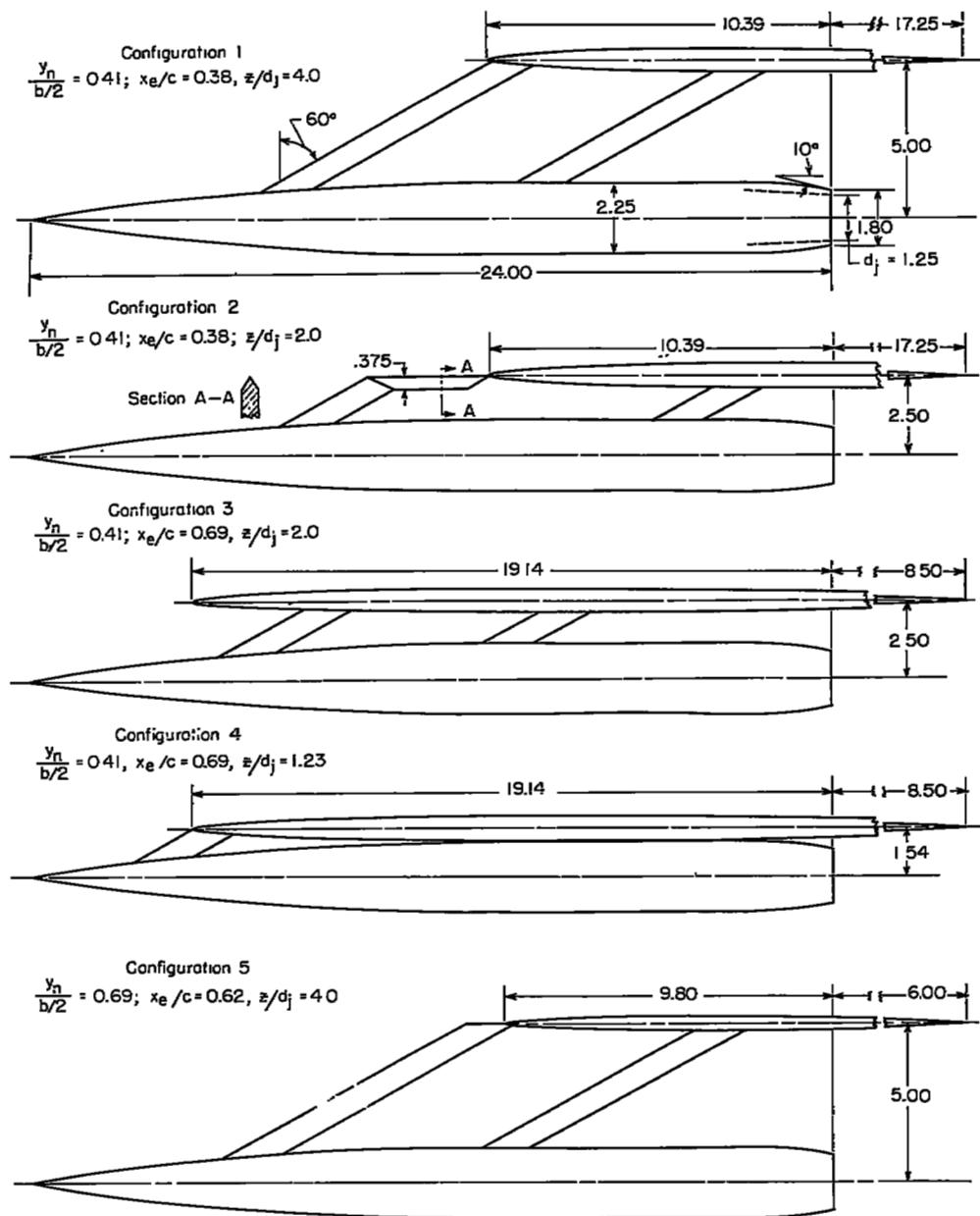
(f) Close-up of the nacelle base. L-57-32

Figure 1.- Concluded.



* These tubes used only when the flap is deflected.

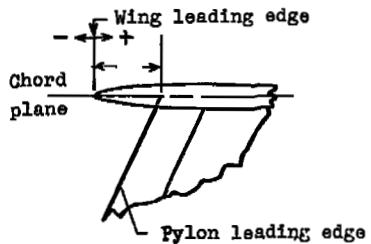
Figure 2.- Model details and wing pressure orifice locations. All model dimensions are in inches unless otherwise noted.



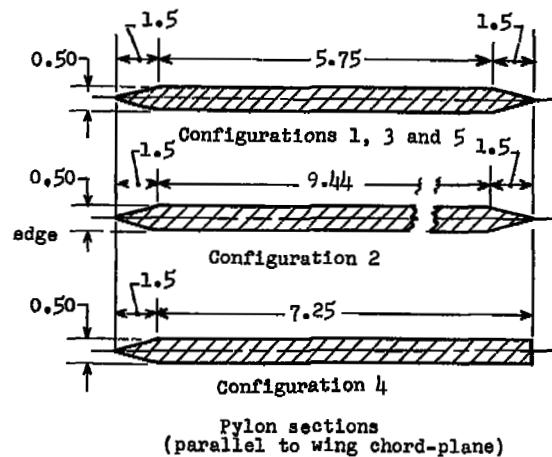
(a) Pylon-nacelle configurations.

Figure 3.- Pylon-nacelle details and nacelle pressure orifice locations.
 All dimensions are in inches unless otherwise noted.

Chordwise Pylon Position	
Configuration	x
1	0.28
2	-3.60
3	3.78
4	-0.03
5	-1.15

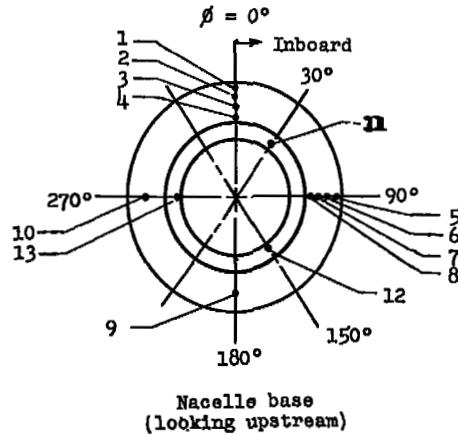


Location of wing chord-plane--pylon leading-edge juncture.



Nacelle Orifice Locations		
Orifice No.	ϕ , deg	x
1	0	21.00
2		22.00
3		23.00
4		23.88
5	90	21.00
6		22.00
7		23.00
8		23.88
9	180	23.00
10	270	23.00
11	30	Base
12	150	
13	270	

Nacelle Ordinates	
x	r
0.00	0.00
0.50	0.08
1.50	0.24
2.50	0.38
3.50	0.51
4.50	0.63
7.00	0.87
10.50	1.07
13.50	1.13
20.50	1.13
21.50	1.12
22.50	1.07
23.50	0.98
24.00	0.90



(b) Dimensional and instrumentation details.

Figure 3.- Concluded.

Flap Orifice Locations									
Station									
A				B		C		D	
0.07 b_f				0.26 b_f		0.71 b_f		0.96 b_f	
$\frac{x}{c}$	$\frac{x}{c_f}$	$\frac{x}{c}$	$\frac{x}{c_f}$	$\frac{x}{c}$	$\frac{x}{c_f}$	$\frac{x}{c}$	$\frac{x}{c_f}$	$\frac{x}{c}$	$\frac{x}{c_f}$
Upper	Lower	Upper and Lower	Upper and Lower	Upper and Lower	Upper and Lower	Upper and Lower	Upper and Lower	Upper and Lower	Upper and Lower
0.90 .92 .94 .96 .98	0.17 .33 .50 .67 .83	0.89 .90 .92 .94 .96	0.08 .17 .33 .50 .67	0.87 0.88 .91 .93 .98	0.08 .17 .33 .50 .67	0.82 .83 .87 .90 .93	0.08 .17 .33 .50 .67	0.77 .79 .83 .87 .96	0.08 .17 .33 .50 .67

Note: The orifices at station A were used for deflected flap only.

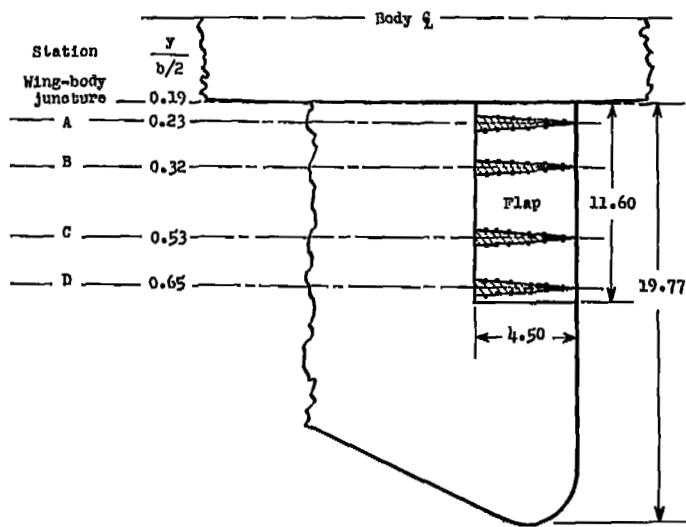


Figure 4.- Sketch of the flap showing pressure orifice locations based on wing and flap chords. All dimensions are in inches.

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